

**Draft Recommendation for  
Space Data System Standards**

**ORBIT DATA  
MESSAGES**

**DRAFT RECOMMENDED STANDARD**

**CCSDS 502.0-P-2.1**

**PINK BOOK**  
October 2021

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**(WHEN THIS RECOMMENDED STANDARD IS FINALIZED, IT WILL CONTAIN THE FOLLOWING STATEMENT OF AUTHORITY:)**

This document has been approved for publication by the Management Council of the Consultative Committee for Space Data Systems (CCSDS) and represents the consensus technical agreement of the participating CCSDS Member Agencies. The procedure for review and authorization of CCSDS documents is detailed in *Organization and Processes for the Consultative Committee for Space Data Systems* (CCSDS A02.1-Y-4), and the record of Agency participation in the authorization of this document can be obtained from the CCSDS Secretariat at the email address below.

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## STATEMENT OF INTENT

### (WHEN THIS RECOMMENDED STANDARD IS FINALIZED, IT WILL CONTAIN THE FOLLOWING STATEMENT OF INTENT:)

The Consultative Committee for Space Data Systems (CCSDS) is an organization officially established by the management of its members. The Committee meets periodically to address data systems problems that are common to all participants, and to formulate sound technical solutions to these problems. Inasmuch as participation in the CCSDS is completely voluntary, the results of Committee actions are termed **Recommended Standards** and are not considered binding on any Agency.

This **Recommended Standard** is issued by, and represents the consensus of, the CCSDS members. Endorsement of this **Recommendation** is entirely voluntary. Endorsement, however, indicates the following understandings:

- o Whenever a member establishes a CCSDS-related **standard**, this **standard** will be in accord with the relevant **Recommended Standard**. Establishing such a **standard** does not preclude other provisions which a member may develop.
- o Whenever a member establishes a CCSDS-related **standard**, that member will provide other CCSDS members with the following information:
  - The **standard** itself.
  - The anticipated date of initial operational capability.
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No later than five years from its date of issuance, this **Recommended Standard** will be reviewed by the CCSDS to determine whether it should: (1) remain in effect without change; (2) be changed to reflect the impact of new technologies, new requirements, or new directions; or (3) be retired or canceled.

In those instances when a new version of a **Recommended Standard** is issued, existing CCSDS-related member standards and implementations are not negated or deemed to be non-CCSDS compatible. It is the responsibility of each member to determine when such standards or implementations are to be modified. Each member is, however, strongly encouraged to direct planning for its new standards and implementations towards the later version of the Recommended Standard.

## FOREWORD

This document is a Recommended Standard for Orbit Data Messages and has been prepared by the CCSDS. The set of orbit data messages described in this Recommended Standard is the baseline concept for trajectory representation in data interchange applications that are cross-supported between Agencies of the CCSDS.

This Recommended Standard establishes a common framework and provides a common basis for the interchange of orbit and orbit-relevant data. It allows implementing organizations within each Agency to proceed coherently with the development of compatible derived standards for the flight and ground systems that are within their cognizance. Derived Agency standards may implement only a subset of the optional features allowed by the Recommended Standard and may incorporate features not addressed by this Recommended Standard.

Through the process of normal evolution, it is expected that expansion, deletion, or modification of this document may occur. This Recommended Standard is therefore subject to CCSDS document management and change control procedures, which are defined in the *Organization and Processes for the Consultative Committee for Space Data Systems* (CCSDS A02.1-Y-4). Current versions of CCSDS documents are maintained at the CCSDS Web site:

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Questions relating to the contents or status of this document should be sent to the CCSDS Secretariat at the email address indicated on page i.

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- National Aeronautics and Space Administration (NASA)/USA.
- UK Space Agency/United Kingdom.

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- South African National Space Agency (SANSA)/Republic of South Africa.
- Space and Upper Atmosphere Research Commission (SUPARCO)/Pakistan.
- Swedish Space Corporation (SSC)/Sweden.
- Swiss Space Office (SSO)/Switzerland.
- United States Geological Survey (USGS)/USA.

## PREFACE

This document is a draft CCSDS Recommended Standard. Its ‘Pink Book’ status indicates that the CCSDS believes the document to be technically mature and has released it for formal review by appropriate technical organizations. As such, its technical contents are not stable, and several iterations of it may occur in response to comments received during the review process.

Implementers are cautioned **not** to fabricate any final equipment in accordance with this document’s technical content.

Recipients of this draft are invited to submit, with their comments, notification of any relevant patent rights of which they are aware and to provide supporting documentation.

**DOCUMENT CONTROL**

<b>Document</b>	<b>Title</b>	<b>Date</b>	<b>Status</b>
CCSDS 502.0-B-1	Orbit Data Messages, Issue 1	September 2004	Original issue, superseded
CCSDS 502.0-B-2	Orbit Data Messages, Recommended Standard, Issue 2	November 2009	Current issue
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NOTE – Changes from the current issue are too numerous to permit meaningful markup.



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# 1 INTRODUCTION

## 1.1 PURPOSE AND SCOPE

This Orbit Data Messages (ODM) Recommended Standard specifies four standard message formats for use in transferring spacecraft orbit information between space agencies and commercial or governmental spacecraft operators: The Orbit Parameter Message (OPM), the Orbit Mean-Elements Message (OMM), the Orbit Ephemeris Message (OEM), and the Orbit Comprehensive Message (OCM). Such exchanges are used for:

- a) pre-flight planning for tracking or navigation support;
- b) scheduling tracking support;
- c) carrying out tracking operations (sometimes called metric predicts);
- d) performing orbit comparisons;
- e) carrying out navigation operations such as orbit propagation and orbit reconstruction;
- f) assessing mutual physical and electromagnetic interference among satellites orbiting the same celestial body (primarily Earth, Moon, and Mars at present);
- g) performing orbit conjunction (collision avoidance) studies; and
- h) developing and executing collaborative maneuvers to mitigate interference or enhance mutual operations.

This Recommended Standard includes sets of requirements and criteria that the message formats have been designed to meet. For exchanges in which these requirements do not capture the needs of the participating agencies and satellite operators, another mechanism may be selected.

The ODM Recommended Standard is an international standard published under the auspices of CCSDS and International Standards Organization (ISO) Technical Committee 20, Subcommittee 13, developed jointly and in concert with the ISO TC20/SC14. As such, this CCSDS standard is also properly labeled as ISO 26900.

The recommended Orbit Data Message format is ASCII (reference [4]).

This ODM document describes both ‘Keyword = Value Notation’ (KVN) as well as Extensible Markup Language (XML) (reference [5]) formatted messages. Selection of KVN or XML format should be mutually agreed between message exchange partners.

NOTE – As currently specified, an OPM, OMM, or OEM file is to represent orbit data for a single spacecraft, and the OCM is to represent orbit data for either a single spacecraft or single parent spacecraft of a parent/child spacecraft deployment scenario. It is possible that the architecture may support multiple spacecraft per file; this could be considered in the future.

## 1.2 APPLICABILITY

The Orbit Data Message family of standardized orbit messages is applicable to all phases of the spacecraft and launch vehicle life cycle. The rationale behind the design of each orbit data message is described in annex E and may help the application engineer to select a suitable message. Definition of the orbit accuracy underlying a particular orbit message is outside of the scope of this Recommended Standard and should be mutually agreed upon between message exchange partners (or specified via COMMENT sections in the message itself). Applicability information specific to each orbit data message format appears in sections 3, 4, 5, and 6, as well as in annex subsections E2.4 and E2.5.

This Recommended Standard is applicable only to the message format and content, but not to its transmission. The transmission of the message between agencies and operators is outside the scope of this document and should be mutually agreed between message exchange partners.

Description of the message formats based on the use of XML is detailed in section 8.

## 1.3 RATIONALE

This update to version 2 of the Orbit Data Messages adds a fourth message type, the OCM, based on collaboration of the CCSDS Navigation Working Group and the ISO Technical Committee 20, Subcommittee 14, Working Group 3 (ISO TC20/SC14/WG3). A full list of the changes in this document is in annex J.

## 1.4 DOCUMENT STRUCTURE

Section 1 provides a brief overview of the CCSDS-recommended Orbit Data Message types, the OPM, OMM, OEM, and OCM.

Section 3 provides details about the structure and content of the OPM.

Section 4 provides details about the structure and content of the OMM.

Section 5 provides details about the structure and content of the OEM.

Section 6 provides details about the structure and content of the OCM.

Section 7 discusses the syntax considerations of the set of Orbit Data Messages (OPM, OMM, OEM, and OCM).

Section 8 provides details on the XML instantiations of the OPM, OMM, OEM, and OCM.

Following the principal content of the document, there are several annexes, both normative and informative, to guide the ODM user.

## 1.5 CONVENTIONS AND DEFINITIONS

### 1.5.1 NOTATION

#### 1.5.1.1 Unit Notations

The following conventions for unit notations apply throughout this Recommended Standard, with message-specific guidance provided in 7.7. Units are drawn from the International System of Units (SI); units are either SI base units, SI derived units, or units outside the SI that are accepted for use with the SI (reference [1]). Except as noted, the units used within this document are as follows:

- d: days, 86400 SI seconds;
- kg: kilograms;
- km: kilometers;
- m: meters;
- n/a: (units are not applicable);
- %: percent;
- s: SI seconds;
- SFU: Solar Flux Units, equivalent to  $10^{-22}$  W/(m<sup>2</sup>\*Hz);
- W: watts.

#### 1.5.1.2 General

The following notational conventions are used in this document:

- a) multiplication of units is denoted with a single asterisk '\*' (e.g., 'kg\*s');
- b) exponents of units are denoted with a double asterisk '\*\*' (e.g., m<sup>2</sup> = m\*\*2);
- c) square roots of units are denoted by the same exponent notation of a double asterisk '\*\*' (e.g.,  $\sqrt{km}$  = km\*\*0.5);
- d) division of units is denoted with a single forward slash '/' (e.g., m/s);
- e) the usual order of operations ordering applies (e.g., exponents before multiplication).

## 1.5.2 NOMENCLATURE

### 1.5.2.1 Normative Text

The following conventions apply for the normative specifications in this Recommended Standard:

- a) the words ‘shall’ and ‘must’ imply a binding and verifiable specification;
- b) the word ‘should’ implies an optional, but desirable, specification;
- c) the word ‘may’ implies an optional specification;
- d) the words ‘is’, ‘are’, and ‘will’ imply statements of fact.

NOTE – These conventions do not imply constraints on diction in text that is clearly informative in nature.

### 1.5.2.2 Informative Text

In the normative sections of this document, informative text is set off from the normative specifications either in notes or under one of the following subsection headings:

- Overview;
- Background;
- Rationale;
- Discussion.

## 1.5.3 DEFINITIONS

For the purposes of this document, the following definitions apply:

- a) the word ‘agencies’ may also be construed as meaning ‘satellite operators’ or ‘satellite service providers’;
- b) the word ‘participant’ denotes an entity that can acquire or broadcast navigation messages and/or radio frequencies, for example, a spacecraft, a tracking station, a tracking instrument, or an agency/operator;
- c) the notation ‘n/a’ signifies ‘not applicable’;
- d) depending on context, the term ‘ODM’ may be used to refer to this document or may be used to refer collectively to the OPM, OMM, OEM, and OCM messages;
- e) an ‘observation’ is a unique measurement set of a satellite’s state from a single sensor configuration at a single time (e.g., azimuth from a single sensor at a single time);



- f) a ‘sensor track’ is a set of observations within a specified number of minutes for the same object, observed by the same sensor configuration, where each observation is within a specified number of minutes (which is dependent on the orbit regime of the object) of the other observations in the track (e.g., a set of 10 two-way transponder range measurements from the same sensor using the same transponder on the satellite), and where the number of minutes could alternately be defined as the time between start and stop of the measurement ‘session’ or signal modulation that enables metric tracking.

## 1.6 REFERENCES

The following publications contain provisions which, through reference in this text, constitute provisions of this document. At the time of publication, the editions indicated were valid. All publications are subject to revision, and users of this document are encouraged to investigate the possibility of applying the most recent editions of the publications indicated below. The CCSDS Secretariat maintains a register of currently valid CCSDS publications.

- [1] *The International System of Units (SI)*. 8th ed., 2006; updated in 2014. Sèvres, France: BIPM, 2006.
- [2] *Time Code Formats*. Issue 4. Recommendation for Space Data System Standards (Blue Book), CCSDS 301.0-B-4. Washington, D.C.: CCSDS, November 2010.
- [3] “Online Index of Objects Launched into Outer Space.” United Nations Office for Outer Space Affairs (UNOOSA). <http://www.unoosa.org/osa/osoindex>.
- [4] *Information Technology—8-Bit Single-Byte Coded Graphic Character Sets—Part 1: Latin Alphabet No. 1*. International Standard, ISO/IEC 8859-1:1998. Geneva: ISO, 1998.
- [5] *XML Specification for Navigation Data Messages*. Recommendation for Space Data System Standards, CCSDS 505.0-B-1. Blue Book. Issue 1. Washington, D.C.: CCSDS, December 2010.
- [6] Paul V. Biron and Ashok Malhotra, eds. *XML Schema Part 2: Datatypes*. 2nd ed. W3C Recommendation.
- [7] *IEEE Standard for Binary Floating-Point Arithmetic*. IEEE Std 754-1985. New York: IEEE, 1985.
- [8] Henry S. Thompson, et al., eds. *XML Schema Part 1: Structures*. 2nd ed. W3C Recommendation.
- [9] *Tracking Data Message*. Issue 2. Recommendation for Space Data System Standards (Blue Book), CCSDS 503.0-B-2. Washington, D.C.: CCSDS, June 2020.

- [10] *Attitude Data Messages*. Issue 1. Recommendation for Space Data System Standards (Blue Book), CCSDS 504.0-B-1. Washington, D.C.: CCSDS, May 2008.
- [11] “CCSDS Navigation Standards Normative Annexes.” Space Assigned Numbers Authority. [https://sanaregistry.org/r/navigation\\_standard\\_normative\\_annexes](https://sanaregistry.org/r/navigation_standard_normative_annexes).
- [12] *Re-entry Data Message*. Issue 1. Recommendation for (Blue Book), CCSDS 508.1-B-1. Washington, D.C.: CCSDS, November 2019.
- [13] *Pointing Request Message*. Issue 1. Recommendation for (Blue Book), CCSDS 509.0-B-1. Washington, D.C.: CCSDS, February 2018.
- [14] *Conjunction Data Message*. Issue 1. Recommendation for Space Data System Standards (Blue Book), CCSDS 508.0-B-1. Washington, D.C.: CCSDS, June 2013.

## 2 OVERVIEW

### 2.1 ORBIT PARAMETER MESSAGE

An OPM specifies the position and velocity of a single object at a specified epoch. Optionally, osculating Keplerian elements may be provided. It should be noted that a sequence of OPMs for either a single object or for multiple objects can be aggregated into a single Navigation Data Message (NDM) XML file as described in 8.12 and shown in annex G. This message is suited to exchanges that (1) involve automated interaction and/or human interaction, and (2) do not require high-fidelity dynamic modeling.

The OPM requires the use of a propagation technique to determine the position and velocity at times different from the specified epoch, leading to a higher level of effort for software implementation than for the OEM.

The OPM also contains an optional 6x6 position/velocity covariance matrix that reflects the uncertainty of the orbit state and may be used in the propagation process to estimate future uncertainties.

The OPM allows for modeling of any number of maneuvers (as both finite and instantaneous events) and simple modeling of solar radiation pressure and atmospheric drag.

Though primarily intended for use by computers, the attributes of the OPM also make it suitable for applications such as exchanges by email, FAX, or voice, or applications in which the message is to be frequently interpreted by humans.

### 2.2 ORBIT MEAN-ELEMENTS MESSAGE

An OMM specifies the orbital characteristics of a single object at a specified epoch, expressed in mean Keplerian elements. It may be noted that a sequence of OMMs for either a single object or for multiple objects can be aggregated into a single NDM XML file as described in 8.12 and shown in annex G. The OMM is suited to exchanges that (1) involve automated interaction and/or human interaction, and (2) do not require high-fidelity dynamic modeling. Such exchanges may be inter-agency exchanges, or ad hoc exchanges among satellite operators when interface control documents have not been negotiated. Ad hoc interactions usually involve more than one satellite, each satellite controlled and operated by a different operating authority.

The OMM includes keywords and values that may be used to generate canonical NORAD Two Line Element (TLE) sets to accommodate the needs of heritage users (see annex H, reference [H3]).

The OMM also contains an optional covariance matrix that reflects the uncertainty of the mean Keplerian elements. This information may be used to determine contact parameters that encompass uncertainties in predicted future states of orbiting objects of interest.

This message is suited for directing antennas and planning contacts with satellites. It is not recommended for assessing mutual physical or electromagnetic interference among Earth-orbiting spacecraft, developing collaborative maneuvers, or propagating precisely the orbits of active satellites, inactive man-made objects, and near-Earth debris fragments. It is not suitable for numerical integration of the governing equations.

Though primarily intended for use by computers, the attributes of the OMM also make it suitable for applications such as exchanges by email, FAX, or voice, or applications in which the message is to be frequently interpreted by humans.

### **2.3 ORBIT EPHEMERIS MESSAGE**

An OEM specifies the position and velocity of a single object at multiple epochs contained within a specified time range. It should be noted that a sequence of OEMs for either a single object or for multiple objects can be aggregated into a single NDM XML file as described in 8.12 and shown in annex G. The OEM is suited to exchanges that (1) involve automated interaction (e.g., computer-to-computer communication when frequent, fast automated time interpretation and processing is required), and (2) require higher fidelity or higher precision dynamic modeling than is possible with the OPM.

The OEM allows for dynamic modeling of any number of gravitational and non-gravitational accelerations. The OEM requires the use of an interpolation technique to interpret the position and velocity at times different from the tabular epochs.

The OEM also contains an optional covariance matrix that reflects the uncertainty of the orbit solution used to generate states in the ephemeris.

### **2.4 ORBIT COMPREHENSIVE MESSAGE**

An OCM specifies position and velocity of either a single object or an en masse parent/child deployment scenario stemming from a single object. It should be noted that a sequence of OCMs for either a single object or for multiple objects can be aggregated into a single NDM XML file as described in 8.12 and shown in annex G.

The OCM aggregates and extends OPM, OEM, and OMM content in a single comprehensive hybrid message (file) and includes the following additional capabilities:

- Optional Earth Orientation (UT1 and UTC) at a nearby (relevant) reference epoch;
- Optional Leap second specification;
- Optional area cross-sections for drag, Solar Radiation Pressure (SRP) perturbations modeling;
- Optional spacecraft dimensions and orientation information for collision probability estimation;

- Optional orbit states (specified using one or more of Cartesian and orbit elements and reference frames) for a single or parent object at either a single epoch or as a time history (ephemeris);
- Optional covariance matrix of selectable/arbitrary order for a single or parent object at either a single epoch or as a time history (ephemeris) that reflects the uncertainty of the orbit solution or simulation used to obtain the nominal states in the orbit state(s);
- Optional covariance content options (e.g., Cartesian 3x3, 6x6, 7x7, or any combination of order, reference frame, and orbit elements);
- Optional maneuver specification (impulsive or finite burn);
- Optional perturbations model specification;
- Optional orbit determination data and metrics.

The OCM simultaneously emphasizes flexibility and message conciseness by offering extensive optional content while minimizing mandatory content. The OCM is well-suited for exchanges that (1) involve automated interaction (e.g., computer-to-computer communication when frequent, fast automated time interpretation and processing is required), and (2) involve regular orbit data transfer for numerous objects (e.g., 200,000) using minimal network bandwidth, disk storage, and quantity of files. The OCM allows the user, in a single message/file, to either embed high-fidelity orbit propagation into an ephemeris time history (akin to the OEM ephemeris) or specify orbital states that can be propagated with supplied perturbations model parameters (akin to OPM content), or both.

## 2.5 EXCHANGE OF MULTIPLE MESSAGES

For a given object, multiple OPM, OMM, or OEM messages may be provided in a message exchange session to achieve ephemeris fidelity requirements, whereas a single, self-contained OCM may be sufficient. If ephemeris information for multiple objects is to be exchanged, then multiple OPM, OMM, OEM, or OCM files must be used, with the exception that the OCM supports parent/child deployment scenario specifications in a single message.

## 2.6 DEFINITIONS

Definitions of time systems, reference frames, planetary models, maneuvers, and other fundamental topics related to the interpretation and processing of state vectors and spacecraft ephemerides are provided in reference [H1].

### 3 ORBIT PARAMETER MESSAGE

#### 3.1 GENERAL

**3.1.1** Orbit information may be exchanged between two participants by sending a state vector (see reference [H1]) for a specified epoch using an OPM. The message recipient must have an orbit propagator available that is able to propagate the OPM state vector to compute the orbit at other desired epochs. For this propagation, additional ancillary information (spacecraft properties such as mass, area, and maneuver planning data, if applicable) may be included with the message.

**3.1.2** Osculating Keplerian elements and the Gravitational Coefficient may be included in the OPM in addition to the Cartesian state to aid the message recipient in performing consistency checks. If any Keplerian element is included, the entire set of elements must be provided.

**3.1.3** If participants wish to exchange mean element information, then the OMM or OCM should be the selected message type (see sections 4 and 6.)

**3.1.4** The use of the OPM is best applicable under the following conditions:

- a) an orbit propagator consistent with the models used to develop the orbit data should be available at the receiver's site.
- b) the receiver's modeling of gravitational forces, solar radiation pressure, atmospheric drag, and thrust phases (see reference [H1]) should fulfill accuracy requirements established between the exchange partners.

**3.1.5** The OPM shall be a plain text file consisting of orbit data for a single object.

NOTE – A sequence of OPMs for either a single object or for multiple objects can be aggregated into a single NDM XML file as described in 8.12 and shown in annex G.

**3.1.6** The OPM file-naming scheme should be mutually agreed between message exchange partners.

**3.1.7** The method of exchanging OPMs should be mutually agreed between message exchange partners.

#### NOTES

- 1 Detailed syntax rules for the OPM are specified in section 7.
- 2 Example OPMs and associated supplementary (non-normative) information are provided in annex G.

## 3.2 OPM CONTENT/STRUCTURE

### 3.2.1 GENERAL

The OPM shall be represented as a combination of the following:

- a) a header;
- b) metadata (data about data);
- c) data; and
- d) optional comments (explanatory information).

### 3.2.2 OPM HEADER

3.2.2.1 Table 3-1 specifies for each header item:

- a) the keyword to be used;
- b) a short description of the item;
- c) examples of allowed values; and
- d) whether the item is Mandatory (M), Optional (O), or Conditional (C). Conditional indicates that the item is mandatory if specified conditions are met (e.g., providing all covariance matrix elements if any are provided).

3.2.2.2 Only those keywords shown in table 3-1 shall be used in an OPM header.

**Table 3-1: OPM Header**

<b>Keyword</b>	<b>Description</b>	<b>Examples of Values</b>	<b>M/O/C</b>
CCSDS_OPM_VERS	Format version in the form of 'x.y', where 'y' is incremented for corrections and minor changes, and 'x' is incremented for major changes.	3.0	M
COMMENT	Comments (allowed in the OPM Header only immediately after the OPM version number). (See 7.8 for formatting rules.)	This is a comment	O
CREATION_DATE	File creation date/time in UTC. (For format specification, see 7.5.10.)	2001-11-06T11:17:33 2002-204T15:56:23Z	M
ORIGINATOR	Creating agency or operator. Select from the accepted set of values indicated in annex B, subsection B1 from the 'Abbreviation' column (when present), or the 'Name' column when an Abbreviation column is not populated. If desired organization is not listed there, follow procedures to request that originator be added to SANA registry.	CNES, ESOC, GSFC, GSOC, JPL, JAXA, INTELSAT, USAF, INMARSAT	M
MESSAGE_ID	ID that uniquely identifies a message from a given originator. The format and content of the message identifier value are at the discretion of the originator.	OPM 201113719185 ABC-12_34	O

### 3.2.3 OPM METADATA

Table 3-2 specifies for each metadata item:

- a) the keyword to be used;
- b) a short description of the item;
- c) examples of allowed values; and
- d) whether the item is Mandatory (M), Optional (O), or Conditional (C). Conditional indicates that the item is mandatory if specified conditions are met (e.g., providing all covariance matrix elements if any are provided).

**3.2.3.1** Only those keywords shown in table 3-2 shall be used in OPM metadata.

NOTE – For some keywords (OBJECT\_NAME, OBJECT\_ID) there are no definitive lists of authorized values maintained by a control authority; references [3] and [11] and the organizations provided on the SANA Registry (annex B, subsection B1) are the best known sources for authorized values to date. (For the TIME\_SYSTEM keyword, see annex B, subsection B3, for guidance and a link to the approved set of values.)



**Table 3-2: OPM Metadata**

Keyword	Description	Examples of Values	M/O/C
COMMENT	Comments (allowed at the beginning of the OPM Metadata). (See 7.8 for formatting rules.)	This is a comment	O
OBJECT_NAME	Spacecraft name for which the orbit state is provided. While there is no CCSDS-based restriction on the value for this keyword, it is recommended to use names from the UN Office of Outer Space Affairs designator index (reference [3], which include Object name and international designator of the participant). Where OBJECT_NAME is not known or cannot be disclosed, the value should be set to UNKNOWN.	EUTELSAT W1 MARS PATHFINDER STS 106 NEAR UNKNOWN	M
OBJECT_ID	Object identifier of the object for which the orbit state is provided. There is no CCSDS-based restriction on the value for this keyword, but values should be the international spacecraft designator as published in the UN Office of Outer Space Affairs designator index (reference [3]). Recommended values have the format YYYY-NNNP{PP}, where: YYYY = Year of launch. NNN = Three-digit serial number of launch in year YYYY (with leading zeros). P{PP} = At least one letter for the identification of the part brought into space by the launch. In cases when the asset is not listed in reference [3], the UN Office of Outer Space Affairs designator index format is not used, or the content cannot be disclosed, the value should be set to UNKNOWN.	2000-052A 1996-068A 2000-053A 1996-008A UNKNOWN	M
CENTER_NAME	<b>Origin of the OPM reference frame</b> , which shall be a natural solar system body (planets, asteroids, comets, and natural satellites), including any planet barycenter or the solar system barycenter. Natural bodies shall be selected from the accepted set of values indicated in annex B, subsection B2.	EARTH EARTH BARYCENTER MOON SOLAR SYSTEM BARYCENTER SUN JUPITER BARYCENTER STS 106 EROS	M
REF_FRAME	Reference frame in which the state vector and optional Keplerian element data are given. Use of values other than those in 3.2.3.3 should be documented in an ICD.	ICRF ITRF2000 EME2000 TEME	M
REF_FRAME_EPOCH	Epoch of reference frame, if not intrinsic to the definition of the reference frame. (See 7.5.10 for formatting rules.)	2001-11-06T11:17:33 2002-204T15:56:23Z	C
TIME_SYSTEM	Time system used for state vector, maneuver, and covariance data. Use of values other than those in 3.2.3.2 should be documented in an ICD.	UTC, TAI, TT, GPS, TDB, TCB	M

**3.2.3.2** Values for the TIME\_SYSTEM keyword should be selected from the following set:

<b>Time System Value</b>	<b>Meaning</b>
GMST	Greenwich Mean Sidereal Time
GPS	Global Positioning System
MET	Mission Elapsed Time (note)
MRT	Mission Relative Time (note)
SCLK	Spacecraft Clock (receiver) (requires rules for interpretation in ICD)
TAI	International Atomic Time
TCB	Barycentric Coordinate Time
TDB	Barycentric Dynamical Time
TCG	Geocentric Coordinate Time
TT	Terrestrial Time
UT1	Universal Time
UTC	Coordinated Universal Time

If MET or MRT is chosen as the TIME\_SYSTEM, then the epoch of either the start of the mission for MRT, or of the event for MET, should either be given in a comment in the message or provided in an ICD. The time system for the start of the mission or the event should also be provided in the comment or the ICD. If these values are used for the TIME\_SYSTEM, then the times given in the file denote a duration from the mission start or event. However, for clarity, an ICD should be used to fully specify the interpretation of the times if these values are to be used. The time format should only utilize three-digit days from the MET or MRT epoch, not months and days of the months.

**3.2.3.3** Values for the REF\_FRAME keyword should be selected from the following set:

<b>REF_FRAME Value</b>	<b>Meaning</b>
EME2000	Earth Mean Equator and Equinox of J2000
GCRF	Geocentric Celestial Reference Frame
GRC	Greenwich Rotating Coordinates
ICRF	International Celestial Reference Frame
ITRF2000	International Terrestrial Reference Frame 2000
ITRF-93	International Terrestrial Reference Frame 1993
ITRF-97	International Terrestrial Reference Frame 1997
MCI	Mars Centered Inertial
TDR	True of Date, Rotating
TEME	True Equator Mean Equinox (only used in OMMs)
TOD	True of Date

### 3.2.4 OPM DATA

3.2.4.1 Table 3-3 provides an overview of the six logical blocks in the OPM Data section (State Vector, Osculating Keplerian Elements, Spacecraft Parameters, Position/Velocity Covariance Matrix, Maneuver Parameters, and User-Defined Parameters), and specifies for each data item:

- a) the keyword to be used;
- b) a short description of the item;
- c) the units to be used;
- d) whether the item is Mandatory (M), Optional (O), or Conditional (C). An ‘M’ denotes mandatory keywords that must be included in this section if that particular data section is included. Conditional indicates that the item is mandatory if specified conditions are met (e.g., providing all covariance matrix elements if any are provided).

3.2.4.2 Only those keywords shown in table 3-3 shall be used in OPM data.

NOTE – Requirements relating to the keywords in table 3-3 appear after the table.

**Table 3-3: OPM Data**

Keyword	Description	Units	M/O/C
<b>State Vector Components in the Specified Coordinate System</b>			
COMMENT	(see 7.8 for formatting rules)		O
EPOCH	Epoch of state vector & optional Keplerian elements (see 7.5.10 for formatting rules)		M
X	Position vector X-component	km	M
Y	Position vector Y-component	km	M
Z	Position vector Z-component	km	M
X_DOT	Velocity vector X-component	km/s	M
Y_DOT	Velocity vector Y-component	km/s	M
Z_DOT	Velocity vector Z-component	km/s	M
<b>Osculating Keplerian Elements in the Specified Reference Frame (none or all parameters of this block must be given)</b>			
COMMENT	(see 7.8 for formatting rules)		O
SEMI_MAJOR_AXIS	Semi-major axis	km	C
ECCENTRICITY	Eccentricity		C
INCLINATION	Inclination	deg	C
RA_OF_ASC_NODE	Right ascension of ascending node	deg	C
ARG_OF_PERICENTER	Argument of pericenter	deg	C
TRUE ANOMALY or MEAN ANOMALY	True anomaly or mean anomaly	deg	C
GM	Gravitational Coefficient (Gravitational Constant × Central Mass)	km <sup>3</sup> /s <sup>2</sup>	C
<b>Spacecraft Parameters (if maneuver is specified, then mass must be provided)</b>			
COMMENT	(see 7.8 for formatting rules)		O

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Keyword	Description	Units	M/O/C
MASS	Spacecraft mass	kg	C
SOLAR_RAD_AREA	Solar Radiation Pressure Area ( $A_R$ )	$m^{**2}$	O
SOLAR_RAD_COEFF	Solar Radiation Pressure Coefficient ( $C_R$ )		O
DRAG_AREA	Drag Area ( $A_D$ )	$m^{**2}$	O
DRAG_COEFF	Drag Coefficient ( $C_D$ )		O
Position/Velocity Covariance Matrix (6x6 Lower Triangular Form. None or all parameters of the matrix must be given. COV_REF_FRAME may be omitted if it is the same as REF_FRAME.)			
COMMENT	(see 7.8 for formatting rules)		O
COV_REF_FRAME	Reference frame in which the covariance data are given. Select from the accepted set of values indicated in 3.2.4.9.		C
CX_X	Covariance matrix [1,1]	$km^{**2}$	C
CY_X	Covariance matrix [2,1]	$km^{**2}$	C
CY_Y	Covariance matrix [2,2]	$km^{**2}$	C
CZ_X	Covariance matrix [3,1]	$km^{**2}$	C
CZ_Y	Covariance matrix [3,2]	$km^{**2}$	C
CZ_Z	Covariance matrix [3,3]	$km^{**2}$	C
CX_DOT_X	Covariance matrix [4,1]	$km^{**2}/s$	C
CX_DOT_Y	Covariance matrix [4,2]	$km^{**2}/s$	C
CX_DOT_Z	Covariance matrix [4,3]	$km^{**2}/s$	C
CX_DOT_X_DOT	Covariance matrix [4,4]	$km^{**2}/s^{**2}$	C
CY_DOT_X	Covariance matrix [5,1]	$km^{**2}/s$	C
CY_DOT_Y	Covariance matrix [5,2]	$km^{**2}/s$	C
CY_DOT_Z	Covariance matrix [5,3]	$km^{**2}/s$	C
CY_DOT_X_DOT	Covariance matrix [5,4]	$km^{**2}/s^{**2}$	C
CY_DOT_Y_DOT	Covariance matrix [5,5]	$km^{**2}/s^{**2}$	C
CZ_DOT_X	Covariance matrix [6,1]	$km^{**2}/s$	C
CZ_DOT_Y	Covariance matrix [6,2]	$km^{**2}/s$	C
CZ_DOT_Z	Covariance matrix [6,3]	$km^{**2}/s$	C
CZ_DOT_X_DOT	Covariance matrix [6,4]	$km^{**2}/s^{**2}$	C
CZ_DOT_Y_DOT	Covariance matrix [6,5]	$km^{**2}/s^{**2}$	C
CZ_DOT_Z_DOT	Covariance matrix [6,6]	$km^{**2}/s^{**2}$	C
Maneuver Parameters (Repeat for each maneuver)			
COMMENT	(see 7.8 for formatting rules)		O
MAN_EPOCH_IGNITION	Epoch of ignition (see 7.5.10 for formatting rules)		O
MAN_DURATION	Maneuver duration (If = 0, impulsive maneuver)	s	O
MAN_DELTA_MASS	Mass change during maneuver (value is < 0)	kg	O
MAN_REF_FRAME	Reference frame in which the velocity increment vector data are given. The user must select from the accepted set of values indicated in 3.2.4.9.		O
MAN_DV_1	1 <sup>st</sup> component of the velocity increment	km/s	O
MAN_DV_2	2 <sup>nd</sup> component of the velocity increment	km/s	O
MAN_DV_3	3 <sup>rd</sup> component of the velocity increment	km/s	O

Keyword	Description	Units	M/O/C
User-Defined Parameters (all parameters in this section must be described in an ICD).			
USER_DEFINED_x	User-defined parameter, where 'x' is replaced by a variable-length user-specified character string. Any number of user-defined parameters may be included, if necessary, to provide essential information that cannot be conveyed in COMMENT statements. Example: USER_DEFINED_EARTH_MODEL = WGS-84		O

**3.2.4.3** All values except Maneuver Parameters in the OPM data are ‘at epoch’, that is, the value of the parameter at the time specified in the EPOCH keyword.

**3.2.4.4** Table 3-3 is broken into six logical blocks, each of which has a descriptive heading. These descriptive headings shall not be included in an OPM, unless they appear in a properly formatted COMMENT statement.

**3.2.4.5** If the solar radiation coefficient,  $C_R$ , is set to zero, no solar radiation pressure shall be considered.

NOTE – It is recommended that  $C_R$  and solar radiation pressure area be provided for GEO spacecraft.

**3.2.4.6** If the atmospheric drag coefficient,  $C_D$ , is set to zero, no atmospheric drag shall be considered.

NOTE – It is recommended that  $C_D$  and drag area be provided for LEO spacecraft.

**3.2.4.7** Parameters for thrust phases may be optionally given for the computation of the trajectory during or after maneuver execution (see reference [H1] for the simplified modeling of such maneuvers). For impulsive maneuvers, MAN\_DURATION must be set to zero. MAN\_DELTA\_MASS may be used for both finite and impulsive maneuvers; the value must be a negative number.

**3.2.4.8** Multiple sets of maneuver parameters may appear. For each maneuver, all the maneuver parameters shall be repeated in the order shown in table 3-3.

**3.2.4.9** If the OPM contains a maneuver definition, then the Conditional elements of the Spacecraft Parameters section (designated with a ‘C’) must be included.

**3.2.4.10** Values in the covariance matrix shall be expressed in the applicable reference frame (COV\_REF\_FRAME keyword) and shall be presented sequentially from upper left [1,1] to lower right [6,6], lower triangular form, row by row, left to right. Variance and covariance values shall be expressed in standard double precision as related in 7.5. This logical block of the OPM may be useful for risk assessment and establishing maneuver and mission margins. The intent is to provide causal connections between output orbit data and both physical hypotheses and measurement uncertainties. These causal relationships guide operators’ corrective actions and mitigations.

**3.2.4.11** Values for the MAN\_REF\_FRAME and COV\_REF\_FRAME keyword may be selected from the following set:

Reference Frame Value	Meaning
RSW	Another name for ‘Radial, Transverse, Normal’
RTN	Radial, Transverse, Normal
TNW	A local orbital coordinate frame that has the x-axis along the velocity vector, W along the orbital angular momentum vector, and N completing the right-handed system

**3.2.4.12** A section of User-Defined Parameters may be provided if necessary. In principle, this provides flexibility, but also introduces complexity, non-standardization, potential ambiguity, and potential processing errors. Accordingly, if used, the keywords and their meanings must be described in an ICD. User-Defined Parameters, if included, should be used as sparingly as possible; their use is not encouraged.

## 4 ORBIT MEAN-ELEMENTS MESSAGE

### 4.1 GENERAL

**4.1.1** Orbit information may be exchanged between two participants by sending an orbital state based on mean Keplerian elements (see reference [H1]) for a specified epoch using an OMM. The message recipient must use appropriate orbit propagator algorithms to correctly propagate the OMM state to compute the orbit at other desired epochs.

**4.1.2** The OMM is intended to allow replication of the data content of an existing TLE in a CCSDS standard format, but the message can also accommodate other implementations of mean elements. All essential fields of the ‘de facto standard’ TLE are included in the OMM in a style that is consistent with that of the other ODMs (i.e., the OPM and OEM). From the fields in the OMM, it is possible to generate a TLE (see reference [H2]). Programs that convert OMMs to TLEs must be aware of the structural requirements of the TLE, including the checksum algorithm and the formatting requirements for the values in the TLE. The checksum and formatting requirements of the TLE do not apply to the values in an OMM.

**4.1.3** If participants wish to exchange osculating element information, then the OPM or the OCM should be the selected message type. (See sections 3 and 6.)

**4.1.4** The use of the OMM is best applicable under the following conditions:

- a) an orbit propagator consistent with the models used to develop the orbit data should be run at the receiver’s site;
- b) the receiver’s modeling of gravitational forces, solar radiation pressure, atmospheric drag, etc. (see reference [H1]), should fulfill accuracy requirements established between the exchange partners.

**4.1.5** The OMM shall be a plain text file consisting of orbit data for a single object.

NOTE – A sequence of OMMs for either a single object or for multiple objects can be aggregated into a single NDM XML file as described in 8.12 and shown in annex G.

**4.1.6** The OMM file-naming scheme should be mutually agreed upon between message exchange partners.

**4.1.7** The method of exchanging OMMs should be mutually agreed upon between message exchange partners.

### NOTES

- 1 Detailed syntax rules for the OMM are specified in section 7.
- 2 Example OMMs and associated supplementary (non-normative) information are provided in annex G.

## **4.2 OMM CONTENT/STRUCTURE**

### **4.2.1 GENERAL**

The OMM shall be represented as a combination of the following:

- a) a header;
- b) metadata (data about data);
- c) data; and
- d) optional comments (explanatory information).

### **4.2.2 OMM HEADER**

**4.2.2.1** Table 4-1 specifies for each header item:

- a) the keyword to be used;
- b) a short description of the item;
- c) examples of allowed values; and
- d) whether the item is Mandatory (M), Optional (O), or Conditional (C). An 'M' denotes mandatory keywords that must be included in this section if that data section is included. Conditional indicates that the item is mandatory if specified conditions are met (e.g., providing all covariance matrix elements if any are provided).

**4.2.2.2** Only those keywords shown in table 4-1 shall be used in an OMM header.



**Table 4-1: OMM Header**

<b>Keyword</b>	<b>Description</b>	<b>Examples of Values</b>	<b>M/O/C</b>
CCSDS_OMM_VERS	Format version in the form of 'x.y', where 'y' is incremented for corrections and minor changes, and 'x' is incremented for major changes.	3.0	M
COMMENT	Comments (allowed in the OMM Header only immediately after the OMM version number). (See 7.8 for formatting rules.)	This is a comment	O
CREATION_DATE	File creation date/time in UTC. (For format specification, see 7.5.10.)	2001-11-06T11:17:33 2002-204T15:56:23Z	M
ORIGINATOR	Creating agency or operator. Select from the accepted set of values indicated in annex B, subsection B1 from the 'Abbreviation' column (when present), or the 'Name' column when an Abbreviation column is not populated. If desired organization is not listed there, follow procedures to request that originator be added to SANA registry.	CNES, ESOC, GSFC, GSOC, JPL, JAXA, INTELSAT, USAF, INMARSAT	M
MESSAGE_ID	ID that uniquely identifies a message from a given originator. The format and content of the message identifier value are at the discretion of the originator.	OMM 201113719185 ABC-12_34	O

### 4.2.3 OMM METADATA

4.2.3.1 Table 4-2 specifies for each metadata item:

- a) the keyword to be used;
- b) a short description of the item;
- c) examples of allowed values; and
- d) whether the item is Mandatory (M), Optional (O), or Conditional (C). Conditional indicates that the item is mandatory if specified conditions are met (e.g., providing all covariance matrix elements if any are provided).

4.2.3.2 Only those keywords shown in 4-2 shall be used in OMM metadata.

NOTE – For some keywords (OBJECT\_NAME and OBJECT\_ID), there are no definitive lists of authorized values maintained by a control authority; references [3] and [11] and the organizations provided on the SANA Registry (annex B, subsection B1) are the best known sources for authorized values to date.

**Table 4-2: OMM Metadata**

Keyword	Description	Examples of Values	M/O/C
COMMENT	Comments (allowed at the beginning of the OMM Metadata). (See 7.8 for formatting rules.)	This is a comment	O
OBJECT_NAME	Spacecraft name for which the orbit state is provided. There is no CCSDS-based restriction on the value for this keyword, but it is recommended to use names from the UN Office of Outer Space Affairs designator index (reference [3]), which include Object name and international designator of the participant. In cases in which OBJECT_NAME is not known or cannot be disclosed, the value should be set to UNKNOWN.	TelKom 2 Spaceway 2 INMARSAT 4-F2 UNKNOWN	M
OBJECT_ID	Object identifier of the object for which the orbit state is provided. There is no CCSDS-based restriction on the value for this keyword, but values should be the international spacecraft designator as published in the UN Office of Outer Space Affairs designator index (reference [3]). Recommended values have the format YYYY-NNNP{PP}, where: YYYY = Year of launch. NNN = Three-digit serial number of launch in year YYYY (with leading zeros). P{PP} = At least one capital letter for the identification of the part brought into space by the launch. In cases in which the asset is not listed in reference [3], the UN Office of Outer Space Affairs designator index format is not used, or the content cannot be disclosed, the value should be set to UNKNOWN.	2005-046A 2005-046B 2003-022A UNKNOWN	M
CENTER_NAME	Origin of the OMM reference frame, which shall be a natural solar system body (planets, asteroids, comets, and natural satellites), including any planet barycenter or the solar system barycenter. Natural bodies shall be selected from the accepted set of values indicated in annex B, subsection B2.	EARTH MARS MOON	M
REF_FRAME	Reference frame in which the Keplerian element data are given. Use of values other than those in 3.2.3.3 should be documented in an ICD.  NOTE – NORAD Two Line Element Sets and corresponding Simplified General Perturbations (SGP) orbit propagator ephemeris outputs are explicitly defined to be in the True Equator Mean Equinox of Date (TEME of Date) reference frame. Therefore, TEME of date shall be used for OMMs based on NORAD Two Line Element sets, rather than the almost imperceptibly different TEME of Epoch (see reference [H2] or [H3] for further details).	ICRF ITRF2000 EME2000 TEME	M

Keyword	Description	Examples of Values	M/O/C
REF_FRAME_EPOCH	Epoch of reference frame, if not intrinsic to the definition of the reference frame. (See 7.5.10 for formatting rules.)	2001-11-06T11:17:33 2002-204T15:56:23Z	C
TIME_SYSTEM	Time system used for Keplerian elements and covariance data. Use of values other than those in 3.2.3.2 should be documented in an ICD.	UTC	M
MEAN_ELEMENT_THEORY	Description of the Mean Element Theory. Indicates the proper method to employ to propagate the state.	SGP4 DSST USM	M

#### 4.2.4 OMM DATA

4.2.4.1 Table 4-3 provides an overview of the five logical blocks in the OMM Data section (Mean Keplerian Elements, Spacecraft Parameters, TLE Related Parameters, Position/Velocity Covariance Matrix, and User-Defined Parameters), and specifies for each data item:

- a) the keyword to be used;
- b) a short description of the item;
- c) the units to be used; and
- d) whether the item is Mandatory (M), Optional (O), or Conditional (C). Conditional indicates that the item is mandatory if specified conditions are met (e.g., providing all covariance matrix elements if any are provided).

4.2.4.2 Only those keywords shown in table 4-3 shall be used in OMM data.

NOTE – Requirements relating to the keywords in table 4-3 appear after the table.

**Table 4-3: OMM Data**

Keyword	Description	Units	M/O/C
Mean Keplerian Elements in the Specified Reference Frame			
COMMENT	(see 7.8 for formatting rules)		O
EPOCH	Epoch of Mean Keplerian elements (see 7.5.10 for formatting rules)		M
SEMI_MAJOR_AXIS or MEAN_MOTION	Semi-major axis in kilometers (preferred), or, if MEAN_ELEMENT_THEORY = SGP/SGP4, the Keplerian Mean motion in revolutions per day	km rev/day	M
ECCENTRICITY	Eccentricity		M
INCLINATION	Inclination	deg	M
RA_OF_ASC_NODE	Right ascension of ascending node	deg	M
ARG_OF_PERICENTER	Argument of pericenter	deg	M
MEAN_ANOMALY	Mean anomaly	deg	M
GM	Gravitational Coefficient (Gravitational Constant × Central Mass)	km <sup>3</sup> /s <sup>2</sup>	O

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Keyword	Description	Units	M/O/C
<b>Spacecraft Parameters</b>			
COMMENT	(see 7.8 for formatting rules.)		O
MASS	Spacecraft Mass	kg	O
SOLAR_RAD_AREA	Solar Radiation Pressure Area ( $A_R$ )	$m^{**2}$	O
SOLAR_RAD_COEFF	Solar Radiation Pressure Coefficient ( $C_R$ )		O
DRAG_AREA	Drag Area ( $A_D$ )	$m^{**2}$	O
DRAG_COEFF	Drag Coefficient ( $C_D$ )		O
<b>TLE Related Parameters (This section is only required if MEAN_ELEMENT_THEORY=SGP/SGP4)</b>			
COMMENT	(see 7.8 for formatting rules.)		O
EPHEMERIS_TYPE	Default value = 0. (See 4.2.4.7.)		O
CLASSIFICATION_TYPE	Default value = U. (See 4.2.4.7.)		O
NORAD_CAT_ID	NORAD Catalog Number ('Satellite Number') an integer of up to nine digits. This keyword is only required if MEAN_ELEMENT_THEORY=SGP/SGP4.		O
ELEMENT_SET_NO	Element set number for this satellite. Normally incremented sequentially but may be out of sync if it is generated from a backup source. Used to distinguish different TLEs, and therefore only meaningful if TLE-based data is being exchanged (i.e., MEAN_ELEMENT_THEORY = SGP/SGP4).		O
REV_AT_EPOCH	Revolution Number		O
BSTAR	SGP/SGP4 drag-like coefficient (in units 1/[Earth radii]). Only required if MEAN_ELEMENT_THEORY=SGP/SGP4	1/ER	O
MEAN_MOTION_DOT	First Time Derivative of the Mean Motion (only required if MEAN_ELEMENT_THEORY = SGP). (See 4.2.4.7 for important details).	rev/day**2	O
MEAN_MOTION_DDOT	Second Time Derivative of Mean Motion (only required if MEAN_ELEMENT_THEORY = SGP). (See 4.2.4.7 for important details).	rev/day**3	O
<b>Position/Velocity Covariance Matrix (6x6 Lower Triangular Form. None or all parameters of the matrix must be given. COV_REF_FRAME may be omitted if it is the same as the REF_FRAME.)</b>			
COMMENT	(see 7.8 for formatting rules.)		O
COV_REF_FRAME	Reference frame in which the covariance data are given. Select from the accepted set of values indicated in 3.2.4.9.		C
CX_X	Covariance matrix [1,1]	$km^{**2}$	C
CY_X	Covariance matrix [2,1]	$km^{**2}$	C
CY_Y	Covariance matrix [2,2]	$km^{**2}$	C
CZ_X	Covariance matrix [3,1]	$km^{**2}$	C
CZ_Y	Covariance matrix [3,2]	$km^{**2}$	C
CZ_Z	Covariance matrix [3,3]	$km^{**2}$	C
CX_DOT_X	Covariance matrix [4,1]	$km^{**2}/s$	C
CX_DOT_Y	Covariance matrix [4,2]	$km^{**2}/s$	C
CX_DOT_Z	Covariance matrix [4,3]	$km^{**2}/s$	C
CX_DOT_X_DOT	Covariance matrix [4,4]	$km^{**2}/s^{**2}$	C
CY_DOT_X	Covariance matrix [5,1]	$km^{**2}/s$	C
CY_DOT_Y	Covariance matrix [5,2]	$km^{**2}/s$	C
CY_DOT_Z	Covariance matrix [5,3]	$km^{**2}/s$	C
CY_DOT_X_DOT	Covariance matrix [5,4]	$km^{**2}/s^{**2}$	C
CY_DOT_Y_DOT	Covariance matrix [5,5]	$km^{**2}/s^{**2}$	C
CZ_DOT_X	Covariance matrix [6,1]	$km^{**2}/s$	C

Keyword	Description	Units	M/O/C
CZ_DOT_Y	Covariance matrix [6,2]	km**2/s	C
CZ_DOT_Z	Covariance matrix [6,3]	km**2/s	C
CZ_DOT_X_DOT	Covariance matrix [6,4]	km**2/s**2	C
CZ_DOT_Y_DOT	Covariance matrix [6,5]	km**2/s**2	C
CZ_DOT_Z_DOT	Covariance matrix [6,6]	km**2/s**2	C
User-Defined Parameters (all parameters in this section must be described in an ICD).			
USER_DEFINED_x	User-defined parameter, where 'x' is replaced by a variable length user specified character string. Any number of user-defined parameters may be included, if necessary, to provide essential information that cannot be conveyed in COMMENT statements. Example: USER_DEFINED_EARTH_MODEL = WGS-84		O

**4.2.4.3** All values in the OMM are ‘at epoch’, that is, the value of the parameter at the time specified in the EPOCH keyword.

**4.2.4.4** Table 4-3 is broken into five logical blocks, each of which has a descriptive heading. These descriptive headings shall not be included in an OMM, unless they appear in a properly formatted COMMENT statement.

**4.2.4.5** Values in the covariance matrix shall be expressed in the applicable reference frame (COV\_REF\_FRAME keyword if used, or REF\_FRAME keyword if not), and shall be presented sequentially from upper left [1,1] to lower right [6,6], lower triangular form, row by row left to right. Variance and covariance values shall be expressed in standard double precision as related in 7.5. This logical block of the OMM may be useful for risk assessment and establishing maneuver and mission margins.

**4.2.4.6** For operations in Earth orbit with a TLE-based OMM, some special conventions must be observed, as follows:

- The value associated with the CENTER\_NAME keyword shall be ‘EARTH’.
- The value associated with the REF\_FRAME keyword shall be ‘TEME’.
- The value associated with the TIME\_SYSTEM keyword shall be ‘UTC’.
- The format of the OBJECT\_NAME and OBJECT\_ID keywords shall be that of the UN Office of Outer Space Affairs designator index (reference [3]).
- The MEAN\_MOTION keyword must be used instead of SEMI\_MAJOR\_AXIS.

**4.2.4.7** For those who wish to use the OMM to represent a TLE, there are several considerations that apply with respect to precision of angle representation, use of certain fields by the propagator, reference frame, etc. Some sources suggest the coding for the EPHEMERIS\_TYPE keyword: 1=SGP, 2=SGP4, 3=SDP4, 4=SGP8, 5=SDP8. Some sources suggest the following coding for the CLASSIFICATION\_TYPE keyword: U=unclassified, S=secret. (For further information see references [H2] and [H3]).

NOTE – If the origin of MEAN\_MOTION\_DOT and MEAN\_MOTION\_DDOT is a TLE or if these values are intended to be used as a TLE, then these values need to be divided by 2 and 6 respectively to reflect the SGP theory Taylor Series expansion terms.

**4.2.4.8** Maneuvers are not accommodated in the OMM. Users of the OMM who wish to model maneuvers may use several OMM files to describe the orbit at applicable epochs.

**4.2.4.9** NORAD Two Line Element Sets are implicitly in a TEME of Date reference frame, which is ill-defined in international standard or convention. TEME may be used only for OMMs based on NORAD Two Line Element sets, and in no other circumstances. There are subtle differences between TEME of Epoch and TEME of Date (see references [H2] and [H3]). The effect is very small relative to TLE accuracy. The preferred option is TEME of Date. Users should specify in the ICD if their assumption is TEME of Epoch.

**4.2.4.10** A section of User-Defined Parameters may be provided if necessary. In principle, this provides flexibility, but also introduces complexity, non-standardization, potential ambiguity, and potential processing errors. Accordingly, if used, the keywords and their meanings must be described in an ICD. User-Defined Parameters, if included, should be used as sparingly as possible; their use is not encouraged.

## 5 ORBIT EPHEMERIS MESSAGE

### 5.1 GENERAL

**5.1.1** Orbit information may be exchanged between two participants by sending an ephemeris in the form of a series of state vectors (Cartesian vectors providing position and velocity, and optionally accelerations) using an OEM. The message recipient must have a means of interpolating across these state vectors to obtain the state at an arbitrary time contained within the span of the ephemeris.

**5.1.2** The OEM may be used for assessing mutual physical or electromagnetic interference among Earth-orbiting spacecraft, developing collaborative maneuvers, and representing the orbits of active satellites, inactive man-made objects, near-Earth debris fragments, etc. The OEM reflects the dynamic modeling of any users' approach to conservative and non-conservative phenomena.

**5.1.3** The OEM shall be a plain text file consisting of orbit data for a single object.

NOTE – A sequence of OEMs for either a single object or for multiple objects can be aggregated into a single NDM XML file as described in 8.12 and shown in annex G.

**5.1.4** The OEM file-naming scheme should be mutually agreed between message exchange partners.

**5.1.5** The method of exchanging OEMs should be mutually agreed between message exchange partners.

#### NOTES

- 1 Detailed syntax rules for the OEM are specified in section 7.
- 2 Example OEMs and associated supplementary (non-normative) information are provided in annex G.

### 5.2 OEM CONTENT/STRUCTURE

#### 5.2.1 GENERAL

**5.2.1.1** The OEM shall be represented as a combination of the following:

- a) a header;
- b) metadata (data about data);
- c) ephemeris data;
- d) optional covariance matrix data; and
- e) optional comments (explanatory information).

**5.2.1.2** OEM files must have a set of minimum required sections; some may be repeated. Table 5-1 outlines the contents of an OEM.

**Table 5-1: OEM File Layout Specifications**

<b>Required Sections</b>	Header Metadata Ephemeris Data (Appropriate comments should also be included, although they are not required.)
<b>Allowable Repetitions of Sections</b>	Covariance Matrix (optional) Metadata Ephemeris Data Covariance Matrix (optional) Metadata Ephemeris Data Covariance Matrix (optional) Metadata Ephemeris Data Covariance Matrix (optional) ...etc. (Appropriate comments should also be included.)

**5.2.2 OEM HEADER**

**5.2.2.1** The OEM header assignments are shown in table 5-2, which specifies for each item:

- a) the keyword to be used;
- b) a short description of the item;
- c) examples of allowed values; and
- d) whether the item is Mandatory (M), Optional (O), or Conditional (C). An ‘M’ denotes mandatory keywords that must be included in this section if that particular data section is included. Conditional indicates that the item is mandatory if specified conditions are met (e.g., providing all covariance matrix elements if any are provided).

**5.2.2.2** Only those keywords shown in table 5-2 shall be used in an OEM header.



**Table 5-2: OEM Header**

<b>Keyword</b>	<b>Description</b>	<b>Examples of Values</b>	<b>M/O/C</b>
CCSDS_OEM_VERS	Format version in the form of 'x.y', where 'y' is incremented for corrections and minor changes, and 'x' is incremented for major changes.	3.0	M
COMMENT	Comments (allowed in the OEM Header only immediately after the OEM version number). (See 7.8 for formatting rules.)	COMMENT This is a comment	O
CREATION_DATE	File creation date and time in UTC. (For format specification, see 7.5.10.)	2001-11-06T11:17:33 2002-204T15:56:23	M
ORIGINATOR	Creating agency or operator. Select from the accepted set of values indicated in annex B, subsection B1 from the 'Abbreviation' column (when present), or the 'Name' column when an Abbreviation column is not populated. If desired organization is not listed there, follow procedures to request that originator be added to SANA registry.	CNES, ESOC, GSFC, GSOC, JPL, JAXA, INTELSAT, USAF, INMARSAT	M
MESSAGE_ID	ID that uniquely identifies a message from a given originator. The format and content of the message identifier value are at the discretion of the originator.	OEM 201113719185 ABC-12_34	O

### 5.2.3 OEM METADATA

5.2.3.1 The OEM metadata assignments are shown in table 5-3, which specifies for each item:

- a) the keyword to be used;
- b) a short description of the item;
- c) examples of allowed values; and
- d) whether the item is Mandatory (M), Optional (O), or Conditional (C). Conditional indicates that the item is mandatory if specified conditions are met (e.g., providing all covariance matrix elements if any are provided).

5.2.3.2 Only those keywords shown in table 5-3 shall be used in OEM metadata.

NOTE – For some keywords (OBJECT\_NAME and OBJECT\_ID) there are no definitive lists of authorized values maintained by a control authority; references [3] and [11] and the organizations provided on the SANA Registry (annex B, subsection B1) are the best known sources for authorized values to date. (For the TIME\_SYSTEM keyword, see annex B, subsection B3, for guidance and a link to the approved set of values.)

**5.2.3.3** A single metadata group shall precede each ephemeris data block. Multiple occurrences of a metadata group followed by an ephemeris data block may be used. Before each metadata group the string ‘META\_START’ shall appear on a separate line and after each metadata group (and before the associated ephemeris data block) the string ‘META\_STOP’ shall appear on a separate line.

**Table 5-3: OEM Metadata**

Keyword	Description	Examples of Values	M/O/C
META_START	The OEM message contains metadata, ephemeris data, and covariance data; this keyword is used to delineate the start of a metadata block within the message (metadata are provided in a block, surrounded by ‘META_START’ and ‘META_STOP’ markers to facilitate file parsing). This keyword must appear on a line by itself.	n/a	M
COMMENT	Comments allowed only immediately after the META_START keyword. (See 7.8 for formatting rules.)	COMMENT This is a comment.	O
OBJECT_NAME	The name of the object for which the ephemeris is provided. There is no CCSDS-based restriction on the value for this keyword, but it is recommended to use names from the UN Office of Outer Space Affairs designator index (reference [3]), which include Object name and international designator of the participant. In cases in which OBJECT_NAME is not known or cannot be disclosed, the value should be set to UNKNOWN.	EUTELSAT W1 MARS PATHFINDER STS 106 NEAR UNKNOWN	M
OBJECT_ID	Object identifier of the object for which the ephemeris is provided. There is no CCSDS-based restriction on the value for this keyword, but values should be the international spacecraft designator as published in the UN Office of Outer Space Affairs designator index (reference [3]). Recommended values have the format YYYY-NNNP{PP}, where: YYYY = Year of launch. NNN = Three-digit serial number of launch in year YYYY (with leading zeros). P{PP} = At least one capital letter for the identification of the part brought into space by the launch.  In cases in which the asset is not listed in reference [3], the UN Office of Outer Space Affairs designator index format is not used, or the content cannot be disclosed, the value should be set to UNKNOWN.	2000-052A 1996-068A 2000-053A 1996-008A UNKNOWN	M

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Keyword	Description	Examples of Values	M/O/C
CENTER_NAME	<p>Origin of the OEM reference frame, which may be a natural solar system body (planets, asteroids, comets, and natural satellites), including any planet barycenter or the solar system barycenter, or another reference frame center (such as a spacecraft, formation flying reference 'chief' spacecraft, etc.).</p> <p>Natural bodies shall be selected from the accepted set of values indicated in annex B, subsection B2.</p> <p>For spacecraft, it is recommended to use either the OBJECT_ID or international designator of the participant as catalogued in the UN Office of Outer Space Affairs designator index (reference [3]).</p>	<p>EARTH EARTH BARYCENTER MOON SOLAR SYSTEM BARYCENTER SUN JUPITER BARYCENTER STS 106 EROS</p>	M
REF_FRAME	Reference frame in which the ephemeris data are given. Use of values other than those in 3.2.3.3 should be documented in an ICD.	<p>ICRF ITRF2000 EME2000 TEME</p>	M
REF_FRAME_EPOCH	Epoch of reference frame, if not intrinsic to the definition of the reference frame. (See 7.5.10 for formatting rules.)	<p>2001-11-06T11:17:33 2002-204T15:56:23Z</p>	C
TIME_SYSTEM	Time system used for ephemeris and covariance data. Use of values other than those in 3.2.3.2 should be documented in an ICD.	<p>UTC, TAI, TT, GPS, TDB, TCB</p>	M
START_TIME	Start of TOTAL time span covered by ephemeris data and covariance data immediately following this metadata block. (For format specification, see 7.5.10.)	<p>1996-12-18T14:28:15.1172 1996-277T07:22:54</p>	M
USEABLE_START_TIME	<p>Start time of USEABLE time span covered by ephemeris data immediately following this metadata block. (For format specification, see 7.5.10.)</p> <p>This optional keyword allows the message creator to introduce fictitious (but numerically smooth) data nodes prior to the actual data time history to support interpolation methods requiring more than two nodes (e.g., pure higher-order Lagrange interpolation methods). The use of this keyword and introduction of fictitious node points are optional and may not be necessary.</p>	<p>1996-12-18T14:28:15.1172 1996-277T07:22:54</p>	O

Keyword	Description	Examples of Values	M/O/C
USEABLE_STOP_TIME	Stop time of USEABLE time span covered by ephemeris data immediately following this metadata block. (For format specification, see 7.5.10.)  This optional keyword allows the message creator to introduce fictitious (but numerically smooth) data nodes following the actual data time history to support interpolation methods requiring more than two nodes (e.g., pure higher-order Lagrange interpolation methods). The use of this keyword and introduction of fictitious node points are optional and may not be necessary.	1996-12- 18T14:28:15.1172 1996-277T07:22:54	O
STOP_TIME	End of TOTAL time span covered by ephemeris data and covariance data immediately following this metadata block. (For format specification, see 7.5.10.)	1996-12- 18T14:28:15.1172 1996-277T07:22:54	M
INTERPOLATION	This keyword may be used to specify the recommended interpolation method for ephemeris data in the immediately following set of ephemeris lines.	HERMITE LINEAR LAGRANGE	O
INTERPOLATION_DEGREE	Recommended interpolation degree for ephemeris data in the immediately following set of ephemeris lines. Must be an integer value. This keyword must be used if the 'INTERPOLATION' keyword is used.	5 1	C
META_STOP	The OEM message contains metadata, ephemeris data, and covariance data; this keyword is used to delineate the end of a metadata block within the message (metadata are provided in a block, surrounded by 'META_START' and 'META_STOP' markers to facilitate file parsing). This keyword must appear on a line by itself.	n/a	M

## 5.2.4 OEM DATA: EPHEMERIS DATA LINES

**5.2.4.1** Each set of ephemeris data, including the time tag, must be provided on a single line. The order in which data items are given shall be fixed: **Epoch, X, Y, Z, X\_DOT, Y\_DOT, Z\_DOT, X\_DDOT, Y\_DDOT, Z\_DDOT**.

**5.2.4.2** The position and velocity terms shall be mandatory; acceleration terms may be provided.

**5.2.4.3** At least one space character must be used to separate the items in each ephemeris data line.

**5.2.4.4** Repeated time tags may occur in consecutive ephemeris data blocks if the STOP\_TIME of the first ephemeris data block is greater than the START\_TIME of the second ephemeris data block. Although the USEABLE\_STOP\_TIME and USEABLE\_START\_TIME of the consecutive ephemeris data blocks must not overlap (except for a possibly shared endpoint), the STOP\_TIME of the first ephemeris data block may be greater than the START\_TIME of the second ephemeris data block if the extra data is required for interpolation purposes.

**5.2.4.5** The TIME\_SYSTEM value must remain fixed within an OEM.

**5.2.4.6** The occurrence of a second (or greater) metadata block after some ephemeris data indicates that interpolation using succeeding ephemeris data with ephemeris data occurring prior to that metadata block shall not be done. This method may be used for proper modeling of propulsive maneuvers or any other source of a discontinuity such as eclipse entry or exit.

**5.2.4.7** Details about interpolation method should be specified using the INTERPOLATION and INTERPOLATION\_DEGREE keywords within the OEM. All data blocks must contain enough ephemeris data records to allow the recommended interpolation method to be carried out consistently throughout the OEM.

## **5.2.5 OEM DATA: COVARIANCE MATRIX LINES**

**5.2.5.1** A single covariance matrix data section may optionally follow each ephemeris data block.

**5.2.5.2** If present, the covariance matrix data lines in the OEM are separated from the ephemeris data by means of two mandatory keywords as specified in table 5-4: 'COVARIANCE\_START' and 'COVARIANCE\_STOP'. The COVARIANCE\_START keyword must appear before the first line of the covariance matrix data. The COVARIANCE\_STOP keyword must appear after the last line of covariance data. Each of these keywords shall appear on a line by itself with no time tags or values.

**5.2.5.3** The epoch of the navigation solution related to the covariance matrix must be provided via the 'EPOCH' keyword. The reference frame of the covariance matrix, if different from that of the states in the ephemeris, must be provided via the 'COV\_REF\_FRAME' keyword.

**5.2.5.4** Values in the covariance matrix shall be expressed in the applicable reference frame (COV\_REF\_FRAME keyword if used, or REF\_FRAME keyword if not), and shall be presented sequentially from upper left [1,1] to lower right [6,6], lower triangular form, row by row left to right. Variance and covariance values shall be expressed in standard double precision as related in 7.5.

**5.2.5.5** At least one space character must be used to separate the items in each covariance matrix data line.

**5.2.5.6** Multiple covariance matrices may appear in the covariance matrix section; they may appear with any desired frequency (one for each navigation solution that makes up the overall ephemeris is recommended). The OEM may also contain propagated covariances, not just individual covariances associated with navigation solutions.

**5.2.5.7** If there are multiple covariance matrices in the data section, they must be ordered by increasing time tag.

**Table 5-4: OEM Covariance Keywords**

<b>Keyword</b>	<b>Description</b>	<b>Examples of Values</b>	<b>M/O/C</b>
COVARIANCE_START	This keyword is used to delineate the start of a covariance data block within the message.	n/a	M
EPOCH	Epoch of covariance matrix. (See 7.5.10 for formatting rules.)	2019-12-28T21:29:07.267	C
COV_REF_FRAME	Reference frame in which the covariance data are given. Select from the accepted set of values indicated in 3.2.3.2 or 3.2.4.9.	EME2000	C
COVARIANCE_STOP	This keyword is used to delineate the end of a covariance data block within the message.	n/a	M

## 6 ORBIT COMPREHENSIVE MESSAGE

### 6.1 GENERAL

**6.1.1** Comprehensive orbit information may be exchanged between two participants by sending orbit data/content for one or more epochs using an OCM. The OCM aggregates and extends OMM, OPM, and OEM content in a single hybrid message. The OCM simultaneously emphasizes flexibility and message conciseness by offering extensive optional standardized content while minimizing mandatory content.

**6.1.2** The OCM shall be a plain text file consisting of orbit data for a single space object, or in the case of a parent/child satellite deployment scenario, a single parent object.

NOTE – A sequence of OCMs for either a single object or for multiple objects can be aggregated into a single NDM XML file as described in 8.12 and shown in annex G.

**6.1.3** Orbit information may be exchanged between two or more participants by sending an ephemeris in the form of one or more time series of orbital states (selectable as orbital elements and/or Cartesian vectors providing position and optionally velocity and accelerations) using an OCM. If orbital states are desired at arbitrary time(s) contained within the span of the provided orbit or covariance time histories, the message recipient must use a suitable interpolation method. For times outside of supplied orbit or covariance state time spans or if the step size between time points is too large to support interpolation (reference [H6]), optional perturbations parameters should be included in this message to allow the message recipient to use a suitably compatible orbit and covariance propagator.

**6.1.4** The OCM may be used for assessing mutual physical or electromagnetic interference among Earth-orbiting spacecraft, developing collaborative maneuvers, and representing the orbits of active satellites, inactive man-made objects, near-Earth debris fragments, etc. The OCM reflects the dynamic modeling of any users' approach to conservative and non-conservative phenomena.

**6.1.5** The OCM file-naming scheme should be mutually agreed between message exchange partners.

**6.1.6** The method of exchanging OCMs should be mutually agreed between message exchange partners.

#### NOTES

- 1 Detailed syntax rules for the OCM are specified in section 7.
- 2 Example OCMs and associated supplementary (non-normative) information are provided in annex G.

## 6.2 OCM STRUCTURE AND OVERARCHING REQUIREMENTS

### 6.2.1 GENERAL STRUCTURE

**6.2.1.1** The OCM shall be represented as the combination of the following **mandatory (M)** and **optional (O)** data blocks, which (where provided) shall be ordered as listed in table 6-1.

**6.2.1.2** Within the tables of each OCM section, each keyword is labeled as being Mandatory (M), Optional (O), or Conditional (C). **An ‘M’ denotes mandatory keywords that must be included in this section if this data section is included.** Keywords that have a pre-defined (default) value are listed as ‘O’ (optional), because if the keyword is not provided, then that default value as defined in the corresponding table shall be used in OCM processing. **A ‘C’ denotes keywords that are mandatory if this data block is included and certain conditions are met**, as specified in the keyword description.

NOTE – One can observe in table 6-1 that the OCM fully supports what might be considered as a ‘degenerate’ case, where the message is constructed without any data blocks. This was an intentional choice, given that the many metadata elements the OCM can accommodate are very useful (e.g., to convey phonebook information, link disparate messages together, and convey timing source information).

**6.2.1.3** In some cases, default values have been provided for mandatory ‘M’ and conditional ‘C’ content. Where such defaults exist and those default values match what the message creator intends, the message creator is not required to explicitly provide those mandatory or conditional keywords in that particular data block, and the default values shall implicitly be adopted by the message recipient.

**6.2.1.4** No defaults are supplied for Optional ‘O’ content. If an optional keyword/tag is not supplied by the message creator, then no value is intended and shall be treated simply as ‘null’ (not set), and no value shall be assumed or used in OCM processing.



**Table 6-1: OCM File Layout and Ordering Specification**

	Section	Content	Status M/O
	OCM Header	A single header of the message	M
	OCM Metadata	A single metadata section (data about data)	M
Data	orbit data 1 □ orbit data $n_{orbit}$	data description data lines One or more orbit state time histories (consisting of one or more orbit states) data description data lines	O
	physical properties	A single space object physical characteristics section	O
	covariance data 1 □ covariance data $n_{covariance}$	data description data lines One or more covariance time histories (each consisting of one or more covariance matrices) data description data lines	O
	maneuver data 1 □ maneuver data $n_{maneuver}$	data description data lines One or more maneuver specifications for either impulsive or finite burns or acceleration profiles data description data lines	O
	perturbations parameters	A single perturbations parameters section (required if an orbit determination section is provided)	C
	orbit determination	A single orbit determination data section	O
	user-defined parameters	A single user-defined parameters section containing data and supplemental comments (explanatory information)	O

## 6.2.2 GENERAL REQUIREMENTS

The following requirements apply to all OCM sections and content:

**6.2.2.1** The order of occurrence of OCM keywords shall be fixed as listed in the keyword value tables in the OCM section descriptions.

**6.2.2.2** If the message creator does not have a value for a mandatory keyword, a value of 'UNKNOWN' may be used.

**6.2.2.3** All time-tags may be specified by either a (signed) double precision relative time (e.g., 20157.26) measured in SI seconds with respect to the specified epoch time (EPOCH\_TZERO) or as an absolute time (e.g., 2018-11-13T11:13:20.5Z in CCSDS Time String A or B format, as specified in 7.5.10).

**6.2.2.4** Duplicate time tags shall not be used in any given OCM data block.

**6.2.2.5** Within an OCM data block, all time-tags must adhere to either relative time or absolute time for the entirety of that data block. Relative and absolute time shall not be used within the same data block.

**6.2.2.6** Time tags of information within ordered sequences of OCM sections may be separated by uniform or non-uniform step size(s).

**6.2.2.7** Time tags of one OCM section may or may not match those of another OCM section.

## 6.2.3 OCM HEADER

**6.2.3.1** Table 6-2 specifies the keywords for each header item.

**6.2.3.2** Only those keywords shown in table 6-2 shall be used in an OCM header.

**6.2.3.3** The order of occurrence of these OCM header keywords shall be fixed as shown in table 6-2.

**Table 6-2: OCM Header**

Keyword	Description	Examples of Values	M/O/C
CCSDS_OCM_VERS	Format version in the form of 'x.y', where 'y' is incremented for corrections and minor changes, and 'x' is incremented for major changes.	3.0	M
COMMENT	Comments (a contiguous set of one or more comment lines may be provided in the OCM Header only immediately after the OCM version number). (See 7.8 for formatting rules.)	This is a comment	O
CREATION_DATE	File creation date/time in UTC. (For format specification, see 7.5.10.)	2001-11-06T11:17:33 2002-204T15:56:23Z	M
ORIGINATOR	Creating agency or operator. Select from the accepted set of values indicated in annex B, subsection B1 from the 'Abbreviation' column (when present), or the 'Name' column when an Abbreviation column is not populated. If desired organization is not listed there, follow procedures to request that originator be added to SANA registry.	CNES, ESOC, GSFC, GSOC, JPL, JAXA, INTELSAT, USAF, INMARSAT	M
MESSAGE_ID	Free-text field containing an ID that uniquely identifies a message from this message originator. The format and content of the message identifier value are at the discretion of the originator.	OCM 201113719185 ABC-12_34	O

## 6.2.4 OCM METADATA

**6.2.4.1** Table 6-3 specifies the metadata keywords. Only those keywords shown in table 6-3 shall be used in OCM metadata.

**6.2.4.2** The metadata section must begin with keyword META\_START and end with keyword META\_STOP.

**6.2.4.3** At most, only one metadata section shall appear in the entire scope of an OCM.

**6.2.4.4** The order of occurrence of these OCM metadata keywords shall be fixed as shown in table 6-3.

### NOTES

- For some keywords (OBJECT\_NAME, OBJECT\_DESIGNATOR) there are no definitive lists of authorized values maintained by a control authority; references [3] and [11] and the organizations provided on the SANA Registry (annex B, subsection B1) are the best known sources for authorized values to date.
- While OBJECT\_NAME, OBJECT\_DESIGNATOR, and INTERNATIONAL\_DESIGNATOR are individually optional, it is recommended that at least one of these three keywords be supplied.

Table 6-3: OCM Metadata

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
META_START	Start of the metadata section.				M
COMMENT	Comments (a contiguous set of one or more comment lines may be provided in the OCM Metadata section; see 7.8 for comment formatting rules).			This is a comment	O
CLASSIFICATION	User-defined free-text message classification/caveats of this OCM. It is recommended that selected values be pre-coordinated between exchanging entities by mutual agreement.			SBU 'Operator-proprietary data; secondary distribution not permitted'	O
OBJECT_NAME	Free-text field containing the name of the object. There is no CCSDS-based restriction on the value for this keyword, but it is recommended to use names from either the UN Office of Outer Space Affairs designator index (reference [3], which include Object name and international designator of the participant), the spacecraft operator, or a State Actor or commercial Space Situational Awareness (SSA) provider maintaining the 'CATALOG_NAME' space catalog. If the object name is not known (uncorrelated object), 'UNKNOWN' may be used (or this keyword omitted).			SPOT-7 ENVISAT IRIDIUM NEXT-8 INTELSAT G-15 UNKNOWN	O

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
INTERNATIONAL_DESIGNATOR	<p>Free-text field containing an international designator for the object as assigned by the UN Committee on Space Research (COSPAR). Such designator values have the following COSPAR format: YYYY-NNNP{PP}, where: YYYY = Year of launch. NNN = Three-digit serial number of launch in year YYYY (with leading zeros). P{PP} = At least one capital letter for the identification of the part brought into space by the launch.</p> <p>In cases where the object has no international designator, the value 'UNKNOWN' may be used.</p> <p>NOTE – The international designator was typically specified by 'OBJECT_ID' in the OPM, OMM, and OEM.</p>			2000-052A 1996-068A 2000-053A 1996-008A UNKNOWN	O
CATALOG_NAME	<p>Free-text field containing the satellite catalog source (or source agency or operator, value to be drawn from the 'Abbreviation' column of the SANA Organizations registry at <a href="https://www.sanaregistry.org/r/organizations">https://www.sanaregistry.org/r/organizations</a>) from which 'OBJECT_DESIGNATOR' was obtained.</p>			CSPOC RFSA ESA COMSPOC	O
OBJECT_DESIGNATOR	<p>Free-text field specification of the unique satellite identification designator for the object, as reflected in the catalog whose name is 'CATALOG_NAME'. If the ID is not known (uncorrelated object), 'UNKNOWN' may be used (or this keyword omitted).</p>			22444 18SPCS 18571 2147483648_04ae[...]d84c UNKNOWN	O
ALTERNATE_NAMES	<p>Free-text comma-delimited field containing alternate name(s) of this space object, including assigned names used by spacecraft operator, State Actors, commercial SSA providers, and/or media.</p>			SV08, IN8	O
ORIGINATOR_POC	<p>Free-text field containing originator or programmatic Point-of-Contact (PoC) for OCM.</p>			Mr. Rodgers	O
ORIGINATOR_POSITION	<p>Free-text field containing contact position of the originator PoC.</p>			Flight Dynamics Mission Design Lead	O

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
ORIGINATOR_PHONE	Free-text field containing originator PoC phone number.			+12345678901	O
ORIGINATOR_EMAIL	Free-text field containing originator PoC email address.			JOHN.DOE@ SOMEWHERE.ORG	O
ORIGINATOR_ADDRESS	Free-text field containing originator's physical address information for OCM creator (suggest comma-delimited address lines).			5040 Spaceflight Ave., Cocoa Beach, FL, USA, 12345	O
TECH_ORG	Free-text field containing the creating agency or operator (value should be drawn from the 'Abbreviation' column of the SANA Organizations registry at <a href="https://www.sanaregistry.org/r/organizations">https://www.sanaregistry.org/r/organizations</a> ).			NASA	O
TECH_POC	Free-text field containing technical PoC for OCM.			Maxwell Smart	O
TECH_POSITION	Free-text field containing contact position of the technical PoC.			Flight Dynamics Mission Design Lead	O
TECH_PHONE	Free-text field containing technical PoC phone number.			+49615130312	O
TECH_EMAIL	Free-text field containing technical PoC email address.			JOHN.DOE@ SOMEWHERE.ORG	O
TECH_ADDRESS	Free-text field containing technical PoC physical address information for OCM creator (suggest comma-delimited address lines).			5040 Spaceflight Ave., Cocoa Beach, FL, USA, 12345	O
PREVIOUS_MESSAGE_ID	Free-text field containing an ID that uniquely identifies the previous message from this message originator for this space object. The format and content of the message identifier value are at the discretion of the originator.  NOTE – One may provide the previous message ID without supplying the 'PREVIOUS_MESSAGE_EPOCH' keyword, and vice versa.			OCM 201113719184 ABC-12_33	O

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
NEXT_MESSAGE_ID	Free-text field containing an ID that uniquely identifies the next message from this message originator for this space object. The format and content of the message identifier value are at the discretion of the originator.  NOTE – One may provide the next message ID without supplying the ‘NEXT_MESSAGE_EPOCH’ keyword, and vice versa.			OCM 201113719186 ABC-12_35	O
ADM_MSG_LINK	Free-text field containing a unique identifier of Attitude Data Message (ADM) (reference [10]) that are linked (relevant) to this Orbit Data Message.			ADM_MSG_35132.txt ADM_ID_0572	O
CDM_MSG_LINK	Free-text field containing a unique identifier of Conjunction Data Message (CDM) (reference [14]) that are linked (relevant) to this Orbit Data Message.			CDM_MSG_35132.txt CDM_ID_8257	O
PRM_MSG_LINK	Free-text field containing a unique identifier of Pointing Request Message (PRM) (reference [13]) that are linked (relevant) to this Orbit Data Message.			PRM_MSG_35132.txt PRM_ID_6897	O
RDM_MSG_LINK	Free-text field containing a unique identifier of Reentry Data Message (RDM) (reference [12]) that are linked (relevant) to this Orbit Data Message.			RDM_MSG_35132.txt RDM_ID_1839	O
TDM_MSG_LINK	Free-text string containing a comma-separated list of file name(s) and/or associated identification number(s) of Tracking Data Message (TDM) (reference [9]) observations upon which this OD is based.			TDM_MSG_37.txt TDM_835, TDM_836	O
OPERATOR	Free-text field containing the operator of the space object.			INTELSAT	O
OWNER	Free-text field containing the owner of the space object.			SIRIUS	O
COUNTRY	Free-text field containing the name of the country, country code, or country abbreviation where the space object owner is based.			US SPAIN	O
CONSTELLATION	Free-text field containing the name of the constellation to which this space object belongs.			SPIRE	O

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
OBJECT_TYPE	Specification of the type of object. Select from the accepted set of values indicated in annex B, subsection B11.			PAYLOAD ROCKET BODY DEBRIS UNKNOWN OTHER	O
TIME_SYSTEM	Time system for all absolute time stamps in this OCM including EPOCH_TZERO. Select from the accepted set of values indicated in annex B, subsection B3. This field is used by all OCM data blocks.  If the SCLK timescale is selected, then 'EPOCH_TZERO' shall be interpreted as the spacecraft clock epoch and both SCLK_OFFSET_AT_EPOCH and SCLK_SEC_PER_SI_SEC shall be supplied.		UTC	UTC	M
EPOCH_TZERO	Default epoch to which all relative times are referenced in data blocks (for format specification, see 7.5.10). The time scale of EPOCH_TZERO is controlled via the 'TIME_SYSTEM' keyword, with the exception that for the SCLK timescale, EPOCH_TZERO shall be interpreted as being in the UTC timescale. This field is used by all OCM data blocks.			2001-11-06T11:17:33	M
OPS_STATUS	Specification of the operational status of the space object. Select from the accepted set of values indicated in annex B, subsection B12.			OPERATIONAL	O
ORBIT_CATEGORY	Specification of the type of orbit. Select from the accepted set of values indicated in annex B, subsection B14.			GEO LEO	O



Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
OCM_DATA_ELEMENTS	Comma-delimited list of elements of information data blocks included in this message. The order shall be the same as the order of the data blocks in the message. Values shall be confined to the following list: ORB, PHYS, COV, MAN, PERT, OD, and USER. If the OCM contains multiple ORB, COV, or MAN data blocks (as allowed by table 6-1), the corresponding ORB, COV, or MAN entry shall be duplicated to match.			ORB, ORB, PHYS, COV, MAN, MAN, PERT, OD, USER	O
SCLK_OFFSET_AT_EPOCH	Defines the number of spacecraft clock counts existing at EPOCH_TZERO. This is only used if the SCLK timescale is employed by the user.	s	0.0	-5000.0	C
SCLK_SEC_PER_SI_SEC	Defines the current number of clock seconds occurring during one SI second. It should be noted that this clock rate may vary with time and is the current approximate value. This is only used if the SCLK timescale is employed by the user.	s	1.0	2.5	C
PREVIOUS_MESSAGE_EPOCH	Creation epoch of the previous message from this originator for this space object. (For format specification, see 7.5.10.)  NOTE – One may provide the previous message epoch without supplying the PREVIOUS_MESSAGE_ID, and vice versa.			2001-11-06T11:17:33	O
NEXT_MESSAGE_EPOCH	Anticipated (or actual) epoch of the next message from this originator for this space object. (For format specification, see 7.5.10.)  NOTE – One may provide the next message epoch without supplying the NEXT_MESSAGE_ID, and vice versa.			2001-11-07T11:17:33	O
START_TIME	Time of the earliest data contained in the OCM, specified as either a relative or absolute time tag.			0.0 2001-11-06T00:00:00	O
STOP_TIME	Time of the latest data contained in the OCM, specified as either a relative or absolute time tag.			86400.0 2001-11-08T00:00:00	O

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
TIME_SPAN	Span of time that the OCM covers, measured in days. TIME_SPAN is defined as (STOP_TIME-START_TIME), measured in days, irrespective of whether START_TIME or STOP_TIME are provided by the message creator.	d		20.0	O
TAIMUTC_AT_TZERO	Difference (TAI – UTC) in seconds (i.e., total number of leap seconds elapsed since 1958) as modeled by the message originator at epoch ‘EPOCH_TZERO’.	s		36	O
NEXT_LEAP_EPOCH	Epoch of next leap second, specified as an absolute time tag.			2016-12-31T23:59:60	O
NEXT_LEAP_TAIMUTC	Difference (TAI – UTC) in seconds (i.e., total number of leap seconds elapsed since 1958) incorporated by the message originator at epoch ‘NEXT_LEAP_EPOCH’. This keyword should be provided if NEXT_LEAP_EPOCH is supplied.	s		37	C
UT1MUTC_AT_TZERO	Difference (UT1 – UTC) in seconds, as modeled by the originator at epoch ‘EPOCH_TZERO’.	s		0.357	O
EOP_SOURCE	Free-text field specifying the source and version of the message originator’s Earth Orientation Parameters (EOP) used in the creation of this message, including leap seconds, TAI – UT1, etc.			CELESTRAK_20201028	O
INTERP_METHOD_EOP	Free-text field specifying the method used to select or interpolate sequential EOP data.			PRECEDING_VALUE NEAREST_NEIGHBOR LINEAR LAGRANGE_ORDER_5	O
CELESTIAL_SOURCE	Free-text field specifying the source and version of the message originator’s celestial body (e.g., Sun/Earth/Planetary) ephemeris data used in the creation of this message.			JPL_DE_FILES	O
META_STOP	End of the metadata section.				M

## **6.2.5 OCM DATA: TRAJECTORY STATE TIME HISTORY**

**6.2.5.1** Table 6-4 provides an overview of the OCM trajectory state time history ('ephemeris') section. Only those keywords shown in table 6-4 shall be used in the OCM trajectory state time history data specification.

**6.2.5.2** Each trajectory state time history data block must begin with keyword TRAJ\_START and end with keyword TRAJ\_STOP.

**6.2.5.3** Multiple trajectory state data blocks shall appear in an OCM only if they are delimited by separate TRAJ\_START and TRAJ\_STOP keywords.

**6.2.5.4** Each trajectory state time history data block should differ from all others in at least one of the following respects:

- a) the selected element set (TRAJ\_TYPE);
- b) the orbit basis, that is, the orbit determination, navigation solution, or simulation (TRAJ\_BASIS\_ID);
- c) the reference frame is unique (TRAJ\_REF\_FRAME);
- d) the orbit center is unique (CENTER\_NAME);
- e) the data interval timespan is unique (i.e., has no overlap with any other data interval(s)).

**6.2.5.5** Where multiple trajectory state time history data blocks are provided for the same TRAJ\_BASIS and TRAJ\_BASIS\_ID, the top-most depiction shall be adopted as the true or master depiction.

**6.2.5.6** Each trajectory state time history shall be time-ordered to be monotonically increasing.

**6.2.5.7** Positionally discontinuous trajectory states (i.e., separated by a gap in the trajectory state time history data across which interpolation should not be performed) shall be represented by separate trajectory state time history data blocks.

**6.2.5.8** Velocity-discontinuous trajectory states (i.e., by introduction of an impulsive maneuver) shall be represented by separate trajectory state time history data blocks.

**6.2.5.9** All trajectory state time history blocks must contain enough data records to allow the recommended interpolation method to be carried out consistently throughout the time span.

**6.2.5.10** If the user includes trajectory states at key mission events or times, it may be useful to provide times, names, and significance for such mission events in the descriptive comment line(s) immediately following the TRAJ\_START keyword.

**6.2.5.11** Each line of orbit ephemeris data shall be provided in fixed order beginning with an absolute or relative time tag, followed by the corresponding trajectory state elements as defined by TRAJ\_TYPE (see SANA registry, reference [11], and annex B, subsection B7).

**6.2.5.12** At least one space character must be used to separate the items in each orbit ephemeris data line.

**6.2.5.13** The number of significant figures and time steps suitable for interpolation of an orbit ephemeris time history should be chosen according to best practice to avoid positional and interpolation loss of precision (reference [H6]).

**6.2.5.14** If a trajectory state time history section is included in the message, a corresponding perturbations section should be included as well to specify the perturbations incorporated in these trajectory states.

**6.2.5.15** The CENTER\_NAME shall be used to specify the origin of the reference frame that the trajectory state time history is specified in. The specified center may either be a natural, gravitationally attracting body such as is provided in annex B, subsection B2, or it may be a non-gravitationally attracting origin to allow relative positional time histories. If a non-gravitationally attracting origin is selected, however, then the specified TRAJ\_TYPE shall be confined to Cartesian or spherical coordinates, where the reference frame may be rotating or inertially fixed.

**Table 6-4: OCM Data: Trajectory State Time History**

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
TRAJ_START	Start of a trajectory state vector or time history section.			n/a	M
COMMENT	Comments (a contiguous set of one or more comment lines may be provided in the Trajectory State Time History section only immediately after the TRAJ_START keyword; see 7.8 for comment formatting rules).			This is a comment	O
TRAJ_ID	Free-text field containing the identification number for this trajectory state time history block.			TRAJ_20160402_XYZ	O
TRAJ_PREV_ID	Free-text field containing the identification number for the previous trajectory state time history, contained either within this message or presented in a previous OCM.  NOTE – If this message is not part of a sequence of orbit time histories or if this trajectory state time history is the first in a sequence of orbit time histories, then TRAJ_PREV_ID should be excluded from this message.			ORB20160305A	O
TRAJ_NEXT_ID	Free-text field containing the identification number for the next trajectory state time history, contained either within this message, or presented in a future OCM.  NOTE – If this message is not part of a sequence of orbit time histories or if this trajectory state time history is the last in a sequence of orbit time histories, then TRAJ_NEXT_ID should be excluded from this message.			ORB20160305C	O

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
TRAJ_BASIS	The basis of this trajectory state time history data. This is a free-text field with the following suggested values: a) 'PREDICTED'. b) 'DETERMINED' when estimated from observation-based determination, reconstruction, and/or calibration. For definitive OD performed onboard spacecraft whose solutions have been telemetered to the ground for inclusion in an OCM, the TRAJ_BASIS shall be DETERMINED. c) 'TELEMETRY' when the trajectory states are read directly from telemetry, for example, based on inertial navigation systems or GNSS data. d) 'SIMULATED' for generic simulations, future mission design studies, and optimization studies. e) 'OTHER' for other bases of this data.			PREDICTED	O
TRAJ_BASIS_ID	Free-text field containing the identification number for the telemetry dataset, orbit determination, navigation solution, or simulation upon which this trajectory state time history block is based. When a matching orbit determination block accompanies this trajectory state time history, the TRAJ_BASIS_ID should match the corresponding OD_ID (see table 6-11).			OD_5910	O
INTERPOLATION	This keyword may be used to specify the recommended interpolation method for ephemeris data in the immediately following set of ephemeris lines. PROPAGATE indicates that orbit propagation is the preferred method to obtain states at intermediate times, via either a midpoint-switching or endpoint switching approach.			HERMITE LINEAR LAGRANGE PROPAGATE	O
INTERPOLATION_DEGREE	Recommended interpolation degree for ephemeris data in the immediately following set of ephemeris lines. Must be an integer value. This keyword must be provided if the 'INTERPOLATION' keyword is used and set to anything other than PROPAGATE.		3	5 1	C

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
ORB_AVERAGING	<p>Specifies whether the provided trajectory state/elements are based upon either osculating or mean elements, and if so, which mean element definition is employed. The intent of this field is to allow the user to correctly interpret how to use the provided orbit elements and know how to use them operationally.</p> <p>Values should be selected from the accepted set indicated in annex B, subsection B13. If an alternate single- or double-averaging formulation other than that provided is used, the user may name as mutually agreed upon by message exchange participants.</p>		OSCULATING	OSCULATING BROUWER KOZAI (other...)	M
PROPAGATOR	Free-text field containing the name of the orbit propagator used to create this trajectory state time history.			HPOP SP SGP4	O
CENTER_NAME	<p>Origin of the orbit reference frame, which may be a natural solar system body (planets, asteroids, comets, and natural satellites), including any planet barycenter or the solar system barycenter, or another reference frame center (such as a spacecraft, formation flying reference ‘chief’ spacecraft, etc.).</p> <p>Natural bodies shall be selected from the accepted set of values indicated in annex B, subsection B2.</p> <p>For spacecraft, it is recommended to use either the ‘OBJECT_NAME’ or ‘INTERNATIONAL_DESIGNATOR’ of the participant as catalogued in the UN Office of Outer Space Affairs designator index (reference [3]). Alternately, the ‘OBJECT_DESIGNATOR’ may be used.</p> <p>For other reference frame origins, this field is a free-text descriptor which may draw upon other naming conventions and sources.</p>		EARTH	EARTH MOON SOLAR SYSTEM BARYCENTER SUN ISS EROS EARTH_SUN_L2 EGLIN	M

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
TRAJ_REF_FRAME	Reference frame of the trajectory state time history. Select from the accepted set of values indicated in annex B, subsection B4.		ICRF3	J2000	M
TRAJ_FRAME_EPOCH	Epoch of the orbit data reference frame, if not intrinsic to the definition of the reference frame. (See 7.5.10 for formatting rules.)		EPOCH_TZERO	2001-11-06T11:17:33 2002-204T15:56:23Z	C
USEABLE_START_TIME	<p>Start time of USEABLE time span covered by ephemeris data immediately following this metadata block. (For format specification, see 7.5.10.)</p> <p>NOTES</p> <p>1 This optional keyword allows the message creator to introduce fictitious (but numerically smooth) data nodes following the actual data time history to support interpolation methods requiring more than two nodes (e.g., pure higher-order Lagrange interpolation methods). The use of this keyword and introduction of fictitious node points are optional and may not be necessary.</p> <p>2 If this keyword is not supplied, then all data shall be assumed to be valid.</p>			1996-12-18T14:28:15.1172 1996-277T07:22:54	
USEABLE_STOP_TIME	<p>Stop time of USEABLE time span covered by ephemeris data immediately following this metadata block. (For format specification, see 7.5.10.)</p> <p>NOTES</p> <p>1 This optional keyword allows the message creator to introduce fictitious (but numerically smooth) data nodes following the actual data time history to support interpolation methods requiring more than two nodes (e.g., pure higher-order Lagrange interpolation methods). The use of this keyword and introduction of fictitious node points are optional and may not be necessary.</p> <p>2 If this keyword is not supplied, then all data shall be assumed to be valid.</p>			1996-12-18T14:28:15.1172 1996-277T07:22:54	



Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
ORB_REVNUM	<p>The integer orbit revolution number associated with first trajectory state in this trajectory state time history block.</p> <p>NOTE – The first ascending node crossing that occurs AFTER launch or deployment is designated to be the beginning of orbit revolution number = one ('1').</p>			1500 30007	O
ORB_REVNUM_BASIS	<p>Specifies the message creator's basis for their orbit revolution counter, with</p> <ul style="list-style-type: none"> <li>– '0', designating that the first launch or deployment trajectory state corresponds to a revolution number of 0.XXXX, where XXXX represents the fraction of an orbit revolution measured from the equatorial plane, and orbit revolution 1.0 begins at the very next (subsequent) ascending node passage;</li> <li>– '1', designating that the first launch or deployment trajectory state corresponds to a revolution number of 1.XXXX, and orbit revolution 2.0 begins at the very next ascending node passage.</li> </ul> <p>This keyword shall be provided if ORB_REVNUM is specified.</p>		0	0 1	C
TRAJ_TYPE	Specifies the trajectory state type; selected per annex B, subsection B7.		CARTPV	CARTP	M

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
TRAJ_UNITS	<p>A comma-delimited set of SI unit designations for each element of the trajectory state time history following the trajectory state time tag solely for informational purposes, provided as a free-text field enclosed in square brackets. When provided, each trajectory state element shall have a corresponding units entry, with non-dimensional values (such as orbit eccentricity) denoted by 'n/a'.</p> <p>NOTE – The listing of units via the TRAJ_UNITS keyword does not override the mandatory units specified for the selected TRAJ_TYPE (links to the relevant SANA registries provided in annex B, subsection B7).</p>			[km,km,km,km/s,km/s,km/s] [km,n/a,deg, deg, deg, deg]	O
... < Insert trajectory state time history here >	Trajectory state time history line(s) shall be formatted as specified in 6.2.5.11, containing time and orbit elements formatted as specified in 7.4.1.5 and corresponding to the selected TRAJ_TYPE in the SANA Orbital Elements registry (annex B, subsection B7). Units are as specified in this registry.				M
TRAJ_STOP	End of a trajectory state vector or time history section.			n/a	M

## **6.2.6 OCM DATA: SPACE OBJECT PHYSICAL CHARACTERISTICS**

**6.2.6.1** Table 6-5 gives an overview of the OCM space object physical characteristics section. Only those keywords shown in table 6-5 shall be used in OCM space object physical characteristics data.

**6.2.6.2** At most, only one space object physical characteristics section shall appear in an OCM.

**6.2.6.3** The space object physical characteristics data section in the OCM shall be indicated by two keywords: PHYS\_START and PHYS\_STOP.

**6.2.6.4** The Space Object Optimally Encompassing Box (OEB) parameters are defined in further detail in informative annex F, subsection F1.

**6.2.6.5** Modeling of cross-sectional area and the contributions of relevant parameters (DRAG\_CONST\_AREA, SRP\_CONST\_AREA, AREA\_ALONG\_OEB\_MAX, AREA\_ALONG\_OEB\_INT, and AREA\_ALONG\_OEB\_MIN) to total cross-sectional area is provided in informative annex F, subsection F1.

**Table 6-5: OCM Data: Space Object Physical Characteristics**

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
PHYS_START	Start of a Space Object Physical Characteristics section.				M
COMMENT	Comments (a contiguous set of one or more comment lines may be provided in the OCM Space Object Physical Characteristics only immediately after the PHYS_START keyword; see 7.8 for comment formatting rules).			This is a comment	O
MANUFACTURER	Free-text field containing the satellite manufacturer's name.			BOEING	O
BUS_MODEL	Free-text field containing the satellite manufacturer's spacecraft bus model name.			702	O
DOCKED_WITH	Free-text field containing a comma-separated list of other space objects that this object is docked to.			ISS	O
DRAG_CONST_AREA	Attitude-independent drag cross-sectional area (AD) facing the relative wind vector, not already incorporated into the attitude-dependent 'AREA_ALONG_OEB' parameters.	m**2		2.5	O
DRAG_COEFF_NOM	Nominal drag Coefficient ( $C_{D_{Nom}}$ ). If the atmospheric drag coefficient, CD, is set to zero, no atmospheric drag shall be considered.			2.2	O
DRAG_UNCERTAINTY	Drag coefficient one sigma ( $1\sigma$ ) percent uncertainty, where the actual range of drag coefficients to within $1\sigma$ shall be obtained from $\left[1.0 \pm \frac{DRAGUNCERTAINTY}{100.0}\right] (C_{D_{Nom}})$ . This factor is intended to allow operators to supply the nominal ballistic coefficient components while accommodating ballistic coefficient uncertainties.	%		10.0	O

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
INITIAL_WET_MASS	Space object total mass at beginning of life.	kg		500	O
WET_MASS	Space object total mass (including propellant, i.e., 'wet mass') at the current reference epoch 'EPOCH_TZERO'.	kg		472.3	O
DRY_MASS	Space object dry mass (without propellant).	kg		300	O
OEB_PARENT_FRAME	Parent reference frame that maps to the OEB frame via the quaternion-based transformation defined in annex F, subsection F1. Select from the accepted set of values indicated in B, subsections B4 and B5.  This keyword shall be provided if OEB_Q1,2,3,4 are specified.		RSW_ROTATING	ITRF1997	C
OEB_PARENT_FRAME_EPOCH	Epoch of the OEB parent frame, if OEB_PARENT_FRAME is provided and its epoch is not intrinsic to the definition of the reference frame. (See 7.5.10 for formatting rules.)		EPOCH_TZERO	2001-11-06T11:17:33 2002-204T15:56:23Z	C
OEB_Q1	$q_1 = e_1 * \sin(\phi/2)$ , where per reference [H1], $\phi$ = Euler rotation angle and $e_1$ = 1st component of Euler rotation axis for the rotation that maps from the OEB_PARENT_FRAME (defined above) to the frame aligned with the OEB (defined in annex F, subsection F1). A value of '-999' denotes a tumbling space object.			-0.575131822	O
OEB_Q2	$q_2 = e_2 * \sin(\phi/2)$ , where per reference [H1], $\phi$ = Euler rotation angle and $e_2$ = 2nd component of Euler rotation axis for the rotation that maps from the OEB_PARENT_FRAME (defined above) to the frame aligned with the Optimally Encompassing Box (defined in annex F, subsection F1). A value of '-999' denotes a tumbling space object.			-0.280510532	O
OEB_Q3	$q_3 = e_3 * \sin(\phi/2)$ , where per reference [H1], $\phi$ = Euler rotation angle and $e_3$ = 3rd component of Euler rotation axis for the rotation that maps from the OEB_PARENT_FRAME (defined above) to the frame aligned with the Optimally Encompassing Box (defined in annex F, subsection F1). A value of '-999' denotes a tumbling space object.			-0.195634856	O

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
OEB_QC	$q_c = \cos(\varphi/2)$ , where per reference [H1], $\varphi$ = the Euler rotation angle for the rotation that maps from the OEB_PARENT_FRAME (defined above) to the frame aligned with the Optimally Encompassing Box (annex F, subsection F1). $q_c$ shall be made non-negative by convention. A value of '-999' denotes a tumbling space object.			0.743144825	O
OEB_MAX	Maximum physical dimension (along $\hat{x}_{OEB}$ ) of the OEB.	m		1	O
OEB_INT	Intermediate physical dimension (along $\hat{y}_{OEB}$ ) of OEB normal to OEB_MAX direction.	m		0.5	O
OEB_MIN	Minimum physical dimension (along $\hat{z}_{OEB}$ ) of OEB in direction normal to both OEB_MAX and OEB_INT directions.	m		0.3	O
AREA_ALONG_OEB_MAX	Attitude-dependent cross-sectional area of space object (not already included in DRAG_CONST_AREA and SRP_CONST_AREA) when viewed along max OEB ( $\hat{x}_{OEB}$ ) direction as defined in annex F.	m**2		0.15	O
AREA_ALONG_OEB_INT	Attitude-dependent cross-sectional area of space object (not already included in DRAG_CONST_AREA and SRP_CONST_AREA) when viewed along intermediate OEB ( $\hat{y}_{OEB}$ ) direction as defined in annex F.	m**2		0.3	O
AREA_ALONG_OEB_MIN	Attitude-dependent cross-sectional area of space object (not already included in DRAG_CONST_AREA and SRP_CONST_AREA) when viewed along minimum OEB ( $\hat{z}_{OEB}$ ) direction as defined in annex F.	m**2		0.5	O
AREA_MIN_FOR_PC	Minimum cross-sectional area for collision probability estimation purposes.	m**2		1.0	O
AREA_MAX_FOR_PC	Maximum cross-sectional area for collision probability estimation purposes.	m**2		1.0	O
AREA_TYP_FOR_PC	Typical (50 <sup>th</sup> percentile) cross-sectional area sampled over all space object orientations for collision probability estimation purposes.	m**2		1.0	O
RCS	Typical (50 <sup>th</sup> percentile) effective Radar Cross Section of the space object sampled over all possible viewing angles.	m**2		1.25	O

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
RCS_MIN	Minimum Radar Cross Section observed for this object.	m**2		1.1	O
RCS_MAX	Maximum Radar Cross Section observed for this object.	m**2		2.5	O
SRP_CONST_AREA	Attitude-independent solar radiation pressure cross-sectional area (AR) facing the Sun, not already incorporated into the attitude-dependent 'AREA_ALONG_OEB' parameters computed from  $\{ \text{AREA\_ALONG\_OEB\_MAX} \cos(\theta_1) + \text{AREA\_ALONG\_OEB\_INT} \cos(\theta_2) + \text{AREA\_ALONG\_OEB\_MIN} \cos(\theta_3) \}$ Where $\theta_i$ represents the angle between the normal to each MAX/INT/MIN face and the direction to the Sun.	m**2		1.0	O
SOLAR_RAD_COEFF	Nominal Solar Radiation Pressure Coefficient ( $C_{RNom}$ ). NOTE – If the solar radiation coefficient, CR, is set to zero, no solar radiation pressure shall be considered.			1.7	O
SOLAR_RAD_UNCERTAINTY	SRP one sigma ( $1\sigma$ ) percent uncertainty, where the actual range of SRP coefficients to within $1\sigma$ shall be obtained from $\left[ 1.0 \pm \frac{SRP\_UNCERTAINTY}{100.0} \right] (C_{RNom})$ . This factor is intended to allow operators to supply the nominal ballistic coefficient components while accommodating ballistic coefficient uncertainties.	%		1.0	O
VM_ABSOLUTE	Typical (50 <sup>th</sup> percentile) absolute Visual Magnitude of the space object sampled over all possible viewing angles and 'normalized' as specified in informative annex F, subsection F2 to a 1 AU Sun-to-target distance, a phase angle of 0°, and a 40,000 km target-to-sensor distance (equivalent of GEO satellite tracked at 15.6° above local horizon).			15.0	O
VM_APPARENT_MIN	Minimum apparent Visual Magnitude observed for this space object.			19.0	O
VM_APPARENT	Typical (50 <sup>th</sup> percentile) apparent Visual Magnitude observed for this space object.			15.0	O

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
VM_APPARENT_MAX	Maximum apparent Visual Magnitude observed for this space object. NOTE – The ‘MAX’ value represents the brightest observation, which associates with a lower Vmag.			16.0	O
REFLECTANCE	Typical (50 <sup>th</sup> percentile) coefficient of REFLECTANCE of the space object over all possible viewing angles, ranging from 0 (none) to 1 (perfect reflectance).			0.7	O
ATT_CONTROL_MODE	Free-text specification of primary mode of attitude control for the space object. Suggested examples include: – THREE_AXIS – SPIN – DUAL_SPIN – TUMBLING – GRAVITY_GRADIENT			SPIN	O
ATT_ACTUATOR_TYPE	Free-text specification of type of actuator for attitude control. Suggested examples include: – ATT_THRUSTERS – ACTIVE_MAG_TORQUE – PASSIVE_MAG_TORQUE – REACTION_WHEELS – MOMENTUM_WHEELS – CONTROL_MOMENT_GYROSCOPE – NONE – OTHER			ATT_THRUSTERS	O
ATT_KNOWLEDGE	Accuracy of attitude knowledge.	deg		0.3	O
ATT_CONTROL	Accuracy of attitude control system (ACS) to maintain attitude, assuming attitude knowledge was perfect (i.e., deadbands).	deg		2.0	O
ATT_POINTING	Overall accuracy of spacecraft to maintain attitude, including attitude knowledge errors and ACS operation.	deg		2.3	O
AVG_MANEUVER_FREQ	Average maneuver frequency, measured in the number of orbit- or attitude-adjust maneuvers per year.	#/yr		20.0	O
MAX_THRUST	Maximum composite thrust the spacecraft can accomplish in any single body-fixed direction.	N		1.0	O



<b>Keyword</b>	<b>Description</b>	<b>Units</b>	<b>Default (if any)</b>	<b>Examples of Values</b>	<b>M/O/C</b>
DV_BOL	Total $\Delta V$ capability of the spacecraft at beginning of life.	km/s		1.0	O
DV_REMAINING	Total $\Delta V$ remaining for the spacecraft.	km/s		0.2	O
IXX	Moment of Inertia about the X-axis of the space object's primary body frame (e.g., SC_Body_1) (see reference [H1]).	kg*m**2		1000.0	O
IYY	Moment of Inertia about the Y-axis.	kg*m**2		800.0	O
IZZ	Moment of Inertia about the Z-axis.	kg*m**2		400.0	O
IXY	Inertia Cross Product of the X & Y axes.	kg*m**2		20.0	O
IXZ	Inertia Cross Product of the X & Z axes.	kg*m**2		40.0	O
IYZ	Inertia Cross Product of the Y & Z axes.	kg*m**2		60.0	O
PHYS_STOP	End of the Space Object Physical Characteristics section.				M

## 6.2.7 OCM DATA: COVARIANCE TIME HISTORY

**6.2.7.1** Table 6-6 provides an overview of the OCM covariance time history section. Only those keywords shown in table 6-6 shall be used in the OCM covariance time history data specification.

**6.2.7.2** Each covariance time history data block must begin with keyword COV\_START and end with keyword COV\_STOP.

**6.2.7.3** Multiple trajectory state covariance data blocks shall appear in an OCM only if they are delimited by separate COV\_START and COV\_STOP keywords.

**6.2.7.4** Each covariance time history data block should differ from all others in at least one of the following respects:

- a) the selected element set (COV\_TYPE);
- b) the orbit basis (COV\_BASIS);
- c) the orbit determination, navigation solution, or simulation (COV\_BASIS\_ID);
- d) the reference frame is unique (COV\_REF\_FRAME);
- e) the data interval timespan is unique (i.e., has no overlap with any other data interval(s)).

**6.2.7.5** When multiple covariance time history data blocks are provided for the same COV\_BASIS and COV\_BASIS\_ID, the top-most depiction shall be adopted as the true or master depiction.

**6.2.7.6** Each covariance time history shall be time-ordered to be monotonically increasing.

**6.2.7.7** Discontinuous covariance time segments shall be represented by separate covariance time history data blocks.

**6.2.7.8** If the user includes trajectory state covariances at key mission events or times, it may be useful to provide times, names, and significance for such mission events in descriptive comment line(s) immediately following the COV\_START keyword.

**6.2.7.9** Values in the trajectory state covariance matrix shall be expressed in the applicable reference frame specified via the COV\_REF\_FRAME keyword.

**6.2.7.10** If a covariance time history section is included in the message:

- a) a corresponding perturbations section should be included as well to specify the perturbations incorporated in these covariances;

- b) each covariance time history line shall begin with either a relative or absolute time tag corresponding to the specified covariance matrix.

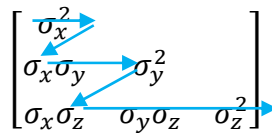
**6.2.7.11** For all covariance representations, the covariance time tag and covariance matrix elements (or dispersions and eigenvectors, in the case of COV\_TYPE= SIG3EIGVEC3 or TSIG3EIGVEC3) shall all be presented on a single line.

**6.2.7.11.1** The composition of the covariance matrix shall be commensurate with the specified COV\_TYPE value.

**6.2.7.11.2** Directly following the time tag specification on the same line as the time tag, all elements of the 'NxN' covariance shall be presented in row wise fashion.

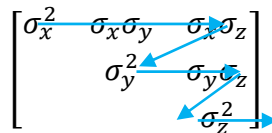
**6.2.7.11.3** On each covariance line, the ordering of the covariance values shall be governed by the 'COV\_ORDERING' keyword. Acceptable values (illustrated with a 3x3 matrix example) are:

**6.2.7.11.3.1** LTM: Lower Triangular Matrix beginning with element [1,1], followed by [2,1], [2,2], [3,1], [3,2] and so on, until all  $\sum_{i=1}^N i$  of the LTM entries have been provided as shown and ordered in figure 6-1.



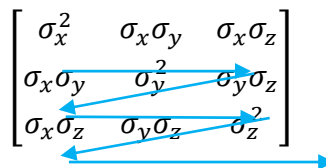
**Figure 6-1: LTM Covariance Element Ordering following Time Tag**

**6.2.7.11.3.2** UTM: Upper Triangular Matrix beginning with element [1,1], followed by [1,2], [1,3], [2,2], [2,3] and so on, until all  $\sum_{i=1}^N i$  of the UTM entries have been provided as shown and ordered in figure 6-2.



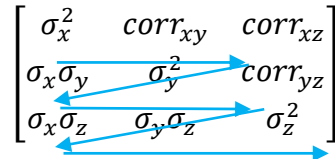
**Figure 6-2: UTM Covariance Element Ordering following Time Tag**

**6.2.7.11.3.3** FULL: The full, symmetric covariance matrix, beginning with element [1,1], followed by [1,2], [1,3], [2,1], [2,2], [2,3], [3,1], [3,2] [3,3] and so on, until all covariance entries (there are  $N^2$  entries in total) have been provided as shown and ordered in figure 6-3.



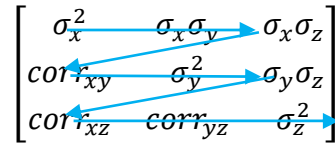
**Figure 6-3: Full Covariance Element Ordering following Time Tag**

**6.2.7.11.3.4 LTMWCC:** Lower Triangular Matrix conflated with cross-correlation terms, where correlation is obtained by dividing the covariance of the two variables by the product of their standard deviations. This combined matrix shall be provided beginning with covariance element [1,1], followed by correlation<sub>xy</sub>, correlation<sub>xz</sub>, covariance [2,1], [2,2], correlation<sub>yz</sub>, covariance [3,1], [3,2] [3,3], and so on until all covariance entries (there are  $N^2$  entries in total) have been provided as shown and ordered in figure 6-4.



**Figure 6-4: LTM Covariance/Correlation Element Ordering following Time Tag**

**6.2.7.11.3.5 UTMWCC:** Upper Triangular Matrix conflated with cross-correlation terms, provided beginning with covariance element [1,1], followed by [1,2] and [1,3]; then correlation<sub>xy</sub>, covariance [2,2], and [2,3]; then correlation<sub>xz</sub>, correlation<sub>yz</sub>, and covariance; [3,3] and so on until all covariance entries (there are  $N^2$  entries in total) have been provided as shown and ordered in figure 6-5.



**Figure 6-5: UTM Covariance/Correlation Element Ordering following Time Tag**

**6.2.7.11.4** At least one space character must be used to separate the items in each covariance matrix data line as related in 7.4.1.6.

**6.2.7.12** Variance and covariance values shall be expressed in floating point or scientific notation as related in 7.5. The number of significant figures and time steps suitable for interpolation of a covariance time history should be chosen according to best practice to avoid covariance interpolation loss of precision (references [H6] and [H13]).

NOTE – It is strongly recommended that covariance matrix time history interpolation be done by using either (1) orbit-dynamics-aware numerical methods as provided in reference [H9] or (2) eigenvalue/vector decomposition, and linear (or higher-order) interpolation of neighboring eigenvalues; Euler axis/angle rotation of eigenvectors at intermediate time(s) of interest; and re-composition of attained eigenvalues and eigenvectors into covariances at time(s) of interest (see annex F, subsection F5 and references [H10], [H11], and [H12]). Direct interpolation of covariance matrix components or failure to incorporate enough significant figures on the interpolated covariance elements can produce invalid (non-positive-semidefinite) covariances.

**Table 6-6: OCM Data: Covariance Time History**

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
COV_START	Start of a covariance time history section.			n/a	M
COMMENT	Comments (a contiguous set of one or more comment lines may be provided in the OCM covariance time history section only immediately after the COV_START keyword; see 7.8 for comment formatting rules).			This is a comment	O
COV_ID	Free-text field containing the identification number for this covariance time history block.			COV_20160402_XYZ	O
COV_PREV_ID	Free-text field containing the identification number for the previous covariance time history, contained either within this message or presented in a previous OCM.  NOTE – If this message is not part of a sequence of covariance time histories or if this covariance time history is the first in a sequence of covariance time histories, then COV_PREV_ID should be excluded from this message.			COV_20160305a	O
COV_NEXT_ID	Free-text field containing the identification number for the next covariance time history, contained either within this message, or presented in a future OCM.  NOTE – If this message is not part of a sequence of covariance time histories or if this covariance time history is the last in a sequence of covariance time histories, then COV_NEXT_ID should be excluded from this message.			COV_20160305C	O

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
COV_BASIS	Basis of this covariance time history data. This is free-text field with the following suggested values: a) 'PREDICTED'. b) 'DETERMINED' when estimated from observation-based orbit determination reconstruction and/or calibration. For definitive OD performed onboard whose solutions have been telemetered to the ground for inclusion in an OCM, the COV_BASIS shall be considered to be DETERMINED. c) EMPIRICAL (for empirically determined such as overlap analyses). d) SIMULATED for simulation-based (including Monte Carlo) estimations, future mission design studies, and optimization studies. e) 'OTHER' for other bases of this data.			PREDICTED EMPIRICAL DETERMINED SIMULATED OTHER	O
COV_BASIS_ID	Free-text field containing the identification number for the orbit determination, navigation solution, or simulation upon which this covariance time history block is based. When a matching orbit determination block accompanies this covariance time history, the COV_BASIS_ID should match the corresponding OD_ID (see table 6-11).			OD_5910	O
COV_REF_FRAME	Reference frame of the covariance time history. Select from the accepted set of values indicated in annex B, subsection B4 and B5.		TNW_INERTIAL	J2000	M
COV_FRAME_EPOCH	Epoch of the covariance data reference frame, if not intrinsic to the definition of the reference frame. (See 7.5.10 for formatting rules.)		EPOCH_TZERO	2001-11-06T11:17:33 2002-204T15:56:23Z	C
COV_SCALE_MIN	Minimum scale factor to apply to this covariance data to achieve realism.			0.5	O
COV_SCALE_MAX	Maximum scale factor to apply to this covariance data to achieve realism.			5.0	O
COV_CONFIDENCE	A measure of the confidence in the covariance errors matching reality, as characterized via a Wald test, a Chi-squared test, the log of likelihood, or a numerical representation per mutual agreement.	%		50	O

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
COV_TYPE	Indicates covariance composition. Select from annex B, subsections B7 and B8.		CARTPV	CARTP CARTPV ADBARV	M
COV_ORDERING	Indicates covariance ordering as being either LTM, UTM, Full covariance, LTM covariance with cross-correlation information provided in upper triangle off-diagonal terms (LTMWCC), or UTM covariance with cross-correlation information provided in lower triangle off-diagonal terms (UTMWCC).		LTM	LTM UTM FULL LTMWCC UTMWCC	M
COV_UNITS	<p>A comma-delimited set of SI unit designations for each element of the covariance time history following the covariance time tag, solely for informational purposes, provided as a free-text field enclosed in square brackets. When provided, these units designations shall correspond to the units of the standard deviations (or square roots) of each of the covariance matrix diagonal elements (or variances), respectively, and all diagonal elements shall have a corresponding units entry, with non-dimensional values (such as dispersion in orbit eccentricity) denoted by 'n/a'.</p> <p><b>NOTE – The listing of units via the COV_UNITS keyword does not override the mandatory units specified for the selected COV_TYPE</b> (links to the relevant SANA registries provided in annex B, subsections B7 and B8).</p>			[km,km,km,km/s,km/s,km/s]	O
...< Insert covariance data here>	Covariance time history line(s) shall be formatted as specified in b) through 6.2.7.12 and 7.4.1.6 and corresponding to the selected time and covariance elements by COV_TYPE.				M
COV_STOP	End of a covariance time history section.			n/a	M

## 6.2.8 OCM DATA: MANEUVER SPECIFICATION

**6.2.8.1** Table 6-7 provides an overview of the OCM maneuver specification section. Only those keywords shown in table 6-7 shall be used in the OCM maneuver time history data specification.

**6.2.8.2** The order of occurrence of these OCM Maneuver Specification keywords shall be fixed as shown in table 6-7.

**6.2.8.3** Maneuver data in the OCM shall be indicated by two keywords: MAN\_START and MAN\_STOP.

**6.2.8.4** Multiple maneuver data blocks shall appear in an OCM only when delimited by separate MAN\_START and MAN\_STOP keywords.

**6.2.8.5** The time intervals of multiple maneuver data blocks may be separated in time, abutted, or overlapped.

NOTE – This is done to accommodate multiple maneuver reference frames, multiple thrusters in simultaneous operation, deployments during thrusting phases, multiple basis definitions (MAN\_BASIS), etc.

**6.2.8.6** Each maneuver data block shall be assigned a maneuver device ID (MAN\_DEVICE\_ID) value, which specifies the unique thruster (or other propulsive device) used in this maneuver sequence time history data block.

**6.2.8.7** Except for the special values ‘ALL’ and ‘DEPLOY’, MAN\_DEVICE\_ID is a free-text field that allows the user to identify which specific thruster or other propulsive device performed this maneuver time history.

**6.2.8.8** A MAN\_DEVICE\_ID value of ‘ALL’ shall be used to indicate that this maneuver sequence represents an aggregation of thrust, acceleration, and/or velocity increments imparted by any/all thrusters utilized in the maneuver that are not attributed to a single specific propulsive device.

**6.2.8.9** A MAN\_DEVICE\_ID value of ‘DEPLOY’ shall be used to indicate that this maneuver data block represents ONLY maneuvers caused by a series of one or more deployments from this host vehicle.

**6.2.8.10** Multiple maneuver data blocks may invoke the same maneuver device ID (MAN\_DEVICE\_ID) value.

**6.2.8.11** All specified maneuver constituents having a common MAN\_ID, MAN\_BASIS, and MAN\_REF\_FRAME shall be added together to obtain the total composite maneuver description.



**6.2.8.12** Each maneuver data block should differ from all other maneuver data blocks in at least one of the following respects:

- a) the maneuver device ID (MAN\_DEVICE\_ID) is unique;
- b) the maneuver device ID is the same, but the 'ON' time intervals are unique and do not overlap with any other data interval(s) for this maneuver device ID (e.g., during multiple interleaved duty cycle 'ON' firings);
- c) the maneuver basis (MAN\_BASIS) is unique;
- d) the reference frame is unique (MAN\_REF\_FRAME);
- e) the maneuver is based upon a unique orbit determination, navigation solution, or simulation (e.g., MAN\_BASIS\_ID);
- f) the data interval timespan is unique (i.e., has no overlap).

**6.2.8.13** If the only difference between multiple maneuver time history data blocks is the selected maneuver composition (MAN\_COMPOSITION) or reference frame (MAN\_REF\_FRAME), the top-most depiction (i.e., the time history occurring first in the OCM) shall be adopted as the official depiction, and those subsequent data blocks shall be treated as containing informational derivative depictions.

**6.2.8.14** The MAN\_COMPOSITION keyword shall specify the individual maneuver time history elements to follow the maneuver time tag.

**6.2.8.15** The MAN\_COMPOSITION keyword shall contain a comma-separated list of values taken from either table 6-8 or table 6-9 (i.e., keywords unique to table 6-8 or table 6-9 shall not be commingled within a single maneuver data block).

**6.2.8.16** The values contained in the MAN\_COMPOSITION keyword shall appear in the order fixed in either table 6-8 or table 6-9.

**6.2.8.17** Maneuver time history lines shall be confined to only one spacecraft object.

**6.2.8.18** Only one of the time tag types (TIME\_ABSOLUTE or TIME\_RELATIVE) shall be selected as the first element of the MAN\_COMPOSITION specification sequence.

**6.2.8.19** Within a single maneuver time history line, acceleration, impulsive  $\Delta V$ , and thrust parameters shall not be additive, but rather shall be interpreted as alternate representations of the same underlying propulsive phenomenology.

**6.2.8.20** Time tag(s) on each maneuver line shall represent the start of the maneuver, with the exception that impulsive  $\Delta V$  entries in the propulsive representation (table 6-8) shall be interpreted as occurring at a time tag of  $T_{\text{start}} + \frac{1}{2} (\text{MAN\_DURA})$ .

NOTE – While one could artificially make  $T_{\text{start}}$  and the impulsive maneuver time be the same value by setting `MAN_DURA` equal to zero, the actual duration of the maneuver is typically nonzero and providing it if/when known facilitates improved modeling and maneuver reconstruction.

**6.2.8.20.1** When invoked, interpolation of acceleration (`ACC_INTERP=ON`) and/or thrust vectors (`THR_INTERP=ON`) shall be done using a suitable interpolation scheme such as the Euler axis/angle formulation discussed in informative annex F, subsection F5.

**6.2.8.20.2** Thrust and acceleration levels for any propulsive device shall be presumed to be ‘OFF’ until explicitly turned ‘ON’ by setting one or more thrust or acceleration components to a non-zero value.

**6.2.8.20.3** Thrust and acceleration shall be set back to ‘OFF’ after the maneuver duration has elapsed [ $T_{\text{start}} + \text{MAN\_DURA}$ ]. Thrusters may also be turned ‘OFF’ by setting all thrust and acceleration components to zero.

**6.2.8.20.4** If thrust is continuous (not affected by a duty cycle), then none of the duty cycle keywords (`DC_XXXX`) are required.

**6.2.8.20.5** If thrust is not continuous (`DC_TYPE`  $\neq$  `CONTINUOUS`), thruster duty cycles shall be triggered either by a reference direction or a reference time.

NOTE – This duty cycle specification imposes cut-outs of non-thrust periods onto the thrust (finite burn) parameters to reflect the periods of duty cycle inactivity.

**6.2.8.20.6** If the value of the `DC_TYPE` keyword is `TIME`, then the following duty cycle parameters shall be present: `DC_WIN_OPEN`, `DC_WIN_CLOSE`, `DC_EXEC_START`, `DC_EXEC_STOP`, `DC_REF_TIME`, `DC_TIME_PULSE_DURATION`, and `DC_TIME_PULSE_PERIOD`.

**6.2.8.20.7** If the value of the `DC_TYPE` keyword is `TIME_AND_ANGLE`, then the following duty cycle parameters shall be present: `DC_WIN_OPEN`, `DC_WIN_CLOSE`, `DC_EXEC_START`, `DC_EXEC_STOP`, `DC_REF_TIME`, `DC_TIME_PULSE_DURATION`, `DC_TIME_PULSE_PERIOD`, `DC_REF_DIR`, `DC_BODY_FRAME`, `DC_BODY_TRIGGER`, `DC_PA_START_ANGLE`, and `DC_PA_STOP_ANGLE`.

**6.2.8.20.8** `DC_MIN_CYCLES` and `DC_MAX_CYCLES` may be specified to constrain the number of duty cycles performed in either `TIME` or `TIME_AND_ANGLE` mode. These parameters may override the duty cycle maneuver stop time (`DC_EXEC_STOP`).

NOTE

1 Relationships between such duty cycle parameters are described in informative annex F, subsection F3.

- 2 The effects of using a pulse width modulation thruster controller can be accommodated by applying a reduced constant thrust level or by invoking the duty cycle parameters, or a combination thereof (being careful to avoid double-booking of thruster degradations).

**Table 6-7: OCM Data: Maneuver Specification**

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
MAN_START	Start of a maneuver time history section.				M
COMMENT	Comments (a contiguous set of one or more comment lines may be provided in the OCM Maneuver Specification only immediately after the MAN_START keyword; see 7.8 for comment formatting rules).			This is a comment	O
MAN_ID	Free-text field containing the unique maneuver identification number for this maneuver. All supplied maneuver 'constituents' within the same MAN_BASIS and MAN_REF_FRAME categories shall be added together to represent the total composite maneuver description.			E_W_20160305B STAGE2	M
MAN_PREV_ID	Free-text field containing the identification number of the previous maneuver for this MAN_BASIS, contained either within this message, or presented in a previous OCM. If this message is not part of a sequence of maneuver messages or if this maneuver is the first in a sequence of maneuvers, then MAN_PREV_ID should be excluded from this message.			E_W_20160305A	O
MAN_NEXT_ID	Free-text field containing the identification number of the next maneuver for this MAN_BASIS, contained either within this message, or presented in a future OCM. If this message is not part of a sequence of maneuver messages or if this maneuver is the last in a sequence of maneuvers, then MAN_NEXT_ID should be excluded from this message.			E_W_20160305C	O

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
MAN_BASIS	Basis of this maneuver time history data, which shall be selected from one of the following values: a) 'CANDIDATE' for a proposed operational or a hypothetical (i.e., mission design and optimization studies) future maneuver b) 'PLANNED' for a currently planned future maneuver. c) 'ANTICIPATED' for a non-cooperative future maneuver that is anticipated (i.e., likely) to occur (e.g., based upon patterns-of-life analysis). d) 'TELEMETRY' when the maneuver is determined directly from telemetry (e.g., based on inertial navigation systems or accelerometers). e) 'DETERMINED' when a past maneuver is estimated from observation-based orbit determination reconstruction and/or calibration. f) 'SIMULATED' for generic maneuver simulations, future mission design studies, and optimization studies. g) 'OTHER' for other bases of this data.			TELEMETRY CANDIDATE	O
MAN_BASIS_ID	Free-text field containing the identification number for the orbit determination, navigation solution, or simulation upon which this maneuver time history block is based. Where a matching orbit determination block accompanies this maneuver time history, the MAN_BASIS_ID should match the corresponding OD_ID (see table 6-11).			OD_20181122A	O
MAN_DEVICE_ID	Free-text field containing the maneuver device identifier used for this maneuver. 'ALL' indicates that this maneuver represents the summed acceleration, velocity increment, or thrust imparted by any/all thrusters utilized in the maneuver.			THR_02 DEPLOYMENT ALL	M
MAN_PREV_EPOCH	Identifies the completion time of the previous maneuver for this MAN_BASIS.			50.0 2001-11-06T11:17:33 2002-204T15:56:23Z	O
MAN_NEXT_EPOCH	Identifies the start time of the next maneuver for this MAN_BASIS.			50.0 2001-11-06T11:17:33 2002-204T15:56:23Z	O

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
MAN_PURPOSE	<p>A free-text field used to specify the intention(s) of the maneuver. Multiple maneuver purposes can be provided as a comma-delimited list, and could include:</p> <ul style="list-style-type: none"> <li>Aerobraking (AEROBRAKE)</li> <li>Attitude adjust (ATTITUDE)</li> <li>Collision avoidance (COLA)</li> <li>Deployment (DEPLOY)</li> <li>Disposal (DISPOSAL)</li> <li>Gravity assist flyby (GRAV_ASSIST_FROM_XXXX, where XXXX=body center name, e.g., SANA Registry, reference [11])</li> <li>Inclination adjustment (INCLINATION)</li> <li>Launch &amp; Early Orbit (LEOP)</li> <li>Maneuver cleanup (MNVR_CLEANUP)</li> <li>Mass adjust (MASS_ADJUST)</li> <li>Momentum desaturation (DESAT)</li> <li>Orbit adjust (ORBIT)</li> <li>Orbit trim (TRIM)</li> <li>Other (OTHER)</li> <li>Period adjustment (PERIOD)</li> <li>Pointing Request Message (PRM_ID_XXXX)</li> <li>Relocation (RELOCATION)</li> <li>Science objective (SCI_OBJ)</li> <li>Spin rate adjust (SPIN_RATE)</li> <li>Station-keeping (SK)</li> <li>Trajectory correction (TRAJ_CORR)</li> </ul>			DISPOSAL	O
MAN_PRED_SOURCE	<p>For future maneuvers, specifies the source of the orbit and/or attitude state(s) upon which the maneuver is based. While there is no CCSDS-based restriction on the value for this free-text keyword, it is suggested to consider using TRAJ_ID and OD_ID keywords as described in tables 6-4 and 6-11, respectively, or a combination thereof.</p>			OD_5	O

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
MAN_REF_FRAME	Reference frame in which all maneuver vector direction data is provided in this maneuver data block. Select from the accepted set of values indicated in annex B, subsections B4 and B5. The reference frame must be the same for all data elements within a given maneuver time history block.		TNW_INERTIAL	J2000	M
MAN_FRAME_EPOCH	Epoch of the maneuver data reference frame, if not intrinsic to the definition of the reference frame. (See 7.5.10 for formatting rules.)			2001-11-06T11:17:33 2002-204T15:56:23Z	C
GRAV_ASSIST_NAME	Origin of maneuver gravitational assist body, which may be a natural solar system body (planets, asteroids, comets, and natural satellites), including any planet barycenter or the solar system barycenter. (See annex B, subsection B2, for acceptable GRAV_ASSIST_NAME values and the procedure to propose new values.)			EARTH MOON EROS JUPITER	O
DC_TYPE	<ul style="list-style-type: none"> <li>– Duty cycle type to use for this maneuver time history section:</li> <li>– CONTINUOUS denotes full/continuous thrust &lt;default&gt;;</li> <li>– TIME denotes a time-based duty cycle driven by time past a reference time and the duty cycle ON and OFF durations;</li> <li>– TIME_AND_ANGLE denotes a duty cycle driven by the phasing/clocking of a space object body frame ‘trigger’ direction past a reference direction.</li> </ul>		CONTINUOUS	CONTINUOUS TIME TIME_AND_ANGLE	M
DC_WIN_OPEN	<p>Start time of the duty cycle-based maneuver window that occurs on or prior to the actual maneuver execution start time. For example, this may identify the time at which the satellite is first placed into a special duty-cycle-based maneuver mode.</p> <p>This keyword shall be set if DC_TYPE ≠ ‘CONTINUOUS’.</p>			50.0 2001-11-06T11:17:33 2002-204T15:56:23Z	C

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
DC_WIN_CLOSE	<p>End time of the duty cycle-based maneuver window that occurs on or after the actual maneuver execution end time. For example, this may identify the time at which the satellite is taken out of a special duty-cycle-based maneuver mode.</p> <p>This keyword shall be set if DC_TYPE ≠ 'CONTINUOUS'.</p>			100.0 2001-11-07T51:17:33 2002-204T15:58:03Z	C
DC_MIN_CYCLES	<p>Minimum number of 'ON' duty cycles (may override DC_EXEC_STOP). This value is optional even if DC_TYPE ≠ 'CONTINUOUS'.</p>			5	O
DC_MAX_CYCLES	<p>Maximum number of 'ON' duty cycles (may override DC_EXEC_STOP). This value is optional even if DC_TYPE ≠ 'CONTINUOUS'.</p>			200	O
DC_EXEC_START	<p>Start time of the initial duty cycle-based maneuver sequence execution. DC_EXEC_START is defined to occur on or prior to the first maneuver 'ON' portion within the duty cycle sequence. DC_EXEC_START must occur coincident with or after DC_WIN_OPEN.</p> <p>This keyword shall be set if DC_TYPE ≠ 'CONTINUOUS'.</p>			50.0 2001-11-06T11:17:33 2002-204T15:56:23Z	C
DC_EXEC_STOP	<p>End time of the final duty cycle-based maneuver sequence execution. DC_EXEC_STOP typically occurs on or after the end of the final maneuver 'ON' portion within the duty cycle sequence. DC_EXEC_STOP occurs prior to or coincident with DC_WIN_CLOSE.</p> <p>This keyword shall be set if DC_TYPE ≠ 'CONTINUOUS'.</p>			100.0 2001-11-07T51:17:33 2002-204T15:58:03Z	C



Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
DC_REF_TIME	<p>Reference time for the THRUST duty cycle, specified as either time in seconds (relative to EPOCH_TZERO), or as an absolute '&lt;epoch&gt;' (see 7.5.10 for formatting rules).</p> <p>NOTE – Depending upon EPOCH_TZERO, DC_REF_TIME relative times may be negative.</p> <p>This keyword shall be set if DC_TYPE ≠ 'CONTINUOUS'.</p>			8000.0 2001-11-06T11:17:33	C
DC_TIME_PULSE_DURATION	<p>Thruster pulse 'ON' duration, initiated at first satisfaction of the burn 'ON' time constraint or upon completion of the previous DC_TIME_PULSE_PERIOD cycle.</p> <p>This keyword shall be set if DC_TYPE ≠ 'CONTINUOUS'.</p>	s		10.0	C
DC_TIME_PULSE_PERIOD	<p>Elapsed time between the start of one pulse and the start of the next. Must be greater than or equal to DC_TIME_PULSE_DURATION.</p> <p>This keyword shall be set if DC_TYPE ≠ 'CONTINUOUS'.</p>	s		200.0	C
DC_REF_DIR	<p>For phase angle thruster duty cycles (DC_TYPE=TIME_AND_ANGLE); specifies the reference vector direction in the 'MAN_REF_FRAME' reference frame at which, when mapped into the space object's spin plane (normal to the spin axis), the duty cycle is triggered (see DC_PA_START_ANGLE for phasing).</p> <p>This (<b>tripartite</b>, or three-element vector) value shall be provided if DC_TYPE = 'TIME_AND_ANGLE'.</p> <p>This reference direction does <u>NOT</u> represent the duty cycle midpoint.</p>			1.0 0.0 0.0	C

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
DC_BODY_FRAME	For phase angle thruster duty cycles (DC_TYPE=TIME_AND_ANGLE); specifies the body reference frame in which DC_BODY_TRIGGER will be specified. Select from the accepted set of values indicated in annex B, subsection B6. This keyword shall be set if DC_TYPE = 'TIME_AND_ANGLE'.			SC_BODY_1 SENSOR_3	C
DC_BODY_TRIGGER	For phase angle thruster duty cycles (DC_TYPE=TIME_AND_ANGLE); specifies the body frame reference vector direction in the 'DC_BODY_FRAME' reference frame at which, when its projection onto the spin plane crosses the corresponding projection of DC_REF_DIR onto the spin plane, this angle-based duty cycle is initiated (see DC_PA_START_ANGLE for phasing).  This tripartite value shall be provided if DC_TYPE = 'TIME_AND_ANGLE'.			0.707 0.0 0.707	C
DC_PA_START_ANGLE	For phase angle thruster duty cycles (DC_TYPE=TIME_AND_ANGLE); specifies the phase angle offset of thruster pulse start, measured with respect to the occurrence of a DC_BODY_TRIGGER crossing of the DC_REF_DIR direction when both are projected into the spin plane (normal to the body spin axis). This phase angle offset can be positive or negative to allow the duty cycle to begin prior to the next crossing of the DC_REF_DIR. As this angular direction is to be used in a modulo sense, there is no requirement for the magnitude of the phase angle offset to be less than 360 degrees.  This keyword shall be set if DC_TYPE = 'TIME_AND_ANGLE'.	deg		25.0	C

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
DC_PA_STOP_ANGLE	For phase angle thruster duty cycles (DC_TYPE=TIME_AND_ANGLE); specifies the phase angle of thruster pulse stop, measured with respect to the DC_BODY_TRIGGER crossing of the DC_REF_DIR direction when both are projected into the spin plane. This phase angle offset can be positive or negative to allow the duty cycle to end after to the next crossing of the DC_REF_DIR. As this angular direction is to be used in a modulo sense, there is no requirement for the magnitude of the phase angle offset to be less than 360 degrees.  This keyword shall be set if DC_TYPE = 'TIME_AND_ANGLE'.	deg		35.0	C
MAN_COMPOSITION	The comma-delimited ordered set of maneuver elements of information contained on every maneuver time history line, with values selected from table 6-8. Within this maneuver data section, the maneuver composition shall include only one TIME specification (TIME_ABSOLUTE or TIME_RELATIVE).			TIME_RELATIVE, THR_X, THR_Y, THR_Z, THR_ISP, THR EFFIC, DELTA_MASS, DV_X, DV_Y, DV_Z, DV_MAG_SIGMA	M
MAN_UNITS	A comma-delimited set of SI unit designations for each and every element of the maneuver time history following the maneuver time tag(s), solely for informational purposes, provided as a free-text field enclosed in square brackets. When MAN_UNITS is provided, all elements of MAN_COMPOSITION must have a corresponding units entry; percentages shall be denoted by '%', and control switches, non-dimensional values, and text strings shall be labelled as 'n/a'.  NOTE – The listing of units via the MAN_UNITS keyword does not override the mandatory units for the selected MAN_COMPOSITION, as specified in table 6-8 or table 6-9.			[N,N,N,s,%,kg, m/s,m/s,m/s, %,n/a]	O
... < Insert maneuver data here>	<Maneuver time history data, content, and units shall be provided, formatted as specified in 7.4.1.7 and corresponding to the selected as specified by MAN_COMPOSITION.>				M
MAN_STOP	End of a maneuver time history section.				M

**6.2.8.20.9** Maneuver elements of information shall be drawn from table 6-8 or table 6-9.

NOTE – Each set of data has practical benefits when applied to maneuver scenarios:

**6.2.8.20.10** Specification of a time history of acceleration parameters ACC\_X, ACC\_Y, and ACC\_Z allows the OCM originator to portray and share the net effects of maneuvers without the OCM recipient needing to do complex finite burn modeling.

**6.2.8.21** Specification of  $\Delta V$  parameters allows simplified modeling assuming impulsive (instantaneous velocity change) maneuvers.

**6.2.8.22** Specification of thrust parameters provides a finite burn capability. In the case of low-thrust and/or long-duration burns, sequential low-thrust interval maneuver lines may be used to reflect the evolution of the low-thrust maneuver thrust parameters.

**Table 6-8: OCM Data: Selectable Propulsive (i.e., Non-Deployment) Maneuver Fields in the Maneuver Time History Data**

Keyword	Description	Units	Examples of Values
TIME_ABSOLUTE	Absolute epoch time as formatted in 7.5.10.	n/a	2018-11-13T11:13:20.5Z
TIME_RELATIVE	Relative epoch time measured in SI seconds with respect to the epoch time specified via the EPOCH_TZERO keyword.	s	20157.26
MAN_DURA	The maneuver duration associated with this impulsive $\Delta V$ , thrust, and/or acceleration-imparted event.	s	200.0
DELTA_MASS	Mass change (where a negative number denotes a mass decrement/loss to the host) associated with this portion ('time slice') of the maneuver. For 'thrust' specification, this mass change shall include the mass change prescribed by the rocket equation.	kg	-5.0
ACC_X	Acceleration component $A_X$ in the selected maneuver frame.	km/s**2	0.000734092785
ACC_Y	Acceleration component $A_Y$ in the selected maneuver frame.	km/s**2	0.000189779834
ACC_Z	Acceleration component $A_Z$ in the selected maneuver frame.	km/s**2	0.0000794872502
ACC_INTERP	Acceleration vector Euler axis/angle interpolation mode between current and next acceleration line.	n/a	OFF ON
ACC_MAG_SIGMA	One-sigma percent error on acceleration magnitude.	%	1.0
ACC_DIR_SIGMA	One-sigma angular off-nominal acceleration vector direction.	deg	5.0
DV_X	Velocity increment $\Delta V_X$ in the selected maneuver reference frame. The actual $\Delta V$ should be impulsively applied at a time of <time tag> + $\frac{1}{2}$ (MAN_DURA).	km/s	0.025
DV_Y	Velocity increment $\Delta V_Y$ in the selected maneuver reference frame. The actual $\Delta V$ should be impulsively applied at a time of <time tag> + $\frac{1}{2}$ (MAN_DURA).	km/s	0.0015
DV_Z	Velocity increment $\Delta V_Z$ in the selected maneuver reference frame. The actual $\Delta V$ should be impulsively applied at a time of <time tag> + $\frac{1}{2}$ (MAN_DURA).	km/s	0.00029
DV_MAG_SIGMA	One-sigma percent error on $\Delta V$ magnitude.	%	2.0
DV_DIR_SIGMA	One-sigma angular off-nominal $\Delta V$ vector direction.	deg	5.0
THR_X	Thrust component $T_X$ measured in the selected maneuver reference frame.	N	1.0

Keyword	Description	Units	Examples of Values
THR_Y	Thrust component $T_Y$ measured in the selected maneuver reference frame.	N	2.0
THR_Z	Thrust component $T_Z$ measured in the selected maneuver reference frame.	N	3.0
THR_EFFIC	Thrust efficiency ' $\eta$ ,' typically ranging between 0.0 and 1.0, that must be applied to the nominal thrust X, Y, and Z constituents to obtain the net resultant thrust applied to the vehicle.	n/a	0.95
THR_INTERP	Thrust vector Euler axis/angle interpolation mode between current and next thrust line; values shall be selected as either 'OFF' or 'ON'.	n/a	OFF ON
THR_ISP	Thrust specific impulse.	s	330.0
THR_MAG_SIGMA	One-sigma percent error on thrust magnitude.	%	2.0
THR_DIR_SIGMA	One-sigma angular off-nominal thrust vector direction.	deg	5.0

**Table 6-9: OCM Data: Selectable Deployment Fields in the Maneuver Time History Data**

Keyword	Description	Units	Examples of Values
TIME_ABSOLUTE	Absolute epoch time as formatted in 7.5.10.	n/a	2018-11-13T11:13:20.5Z
TIME_RELATIVE	Relative epoch time measured in SI seconds with respect to the epoch time specified via the EPOCH_TZERO keyword.	s	20157.26
DEPLOY_ID	Free-text identifier of the resulting 'child' object deployed from this host at this time tag. Setting DEPLOY_ID to zero (value = 0) indicates that a deployment did not occur.	n/a	CubeSat_001
DEPLOY_DV_X	Velocity increment $\Delta V_X$ of the deployed 'child' object measured in the selected maneuver reference frame, applied instantaneously at the time tag of deployment.	km/s	0.0001
DEPLOY_DV_Y	Velocity increment $\Delta V_Y$ of the deployed 'child' object measured in the selected maneuver reference frame, applied instantaneously at the time tag of deployment.	km/s	0.00003
DEPLOY_DV_Z	Velocity increment $\Delta V_Z$ of the deployed 'child' object measured in the selected maneuver reference frame, applied instantaneously at the time tag of deployment.	km/s	0.00002
DEPLOY_MASS	Decrement in host mass as a result of deployment (shall be $\leq 0.0$ ).	kg	-1.0
DEPLOY_DV_SIGMA	One-sigma percent error on deployment $\Delta V$ magnitude.	%	5.0
DEPLOY_DIR_SIGMA	One-sigma angular off-nominal deployment vector direction.	deg	5.0

Keyword	Description	Units	Examples of Values
DEPLOY_DV_RATIO	Ratio of child-to-host $\Delta V$ vectors, such that:  $\overline{\Delta v}_{host} = \text{DEPLOY\_DV\_RATIO} \times \overline{\Delta v}_{child}$ NOTE – As an opposite $\Delta V$ is typically imparted to the host during deployment, this number is typically less than or equal to zero. This ratio allows the user to specify how much $\Delta V$ is imparted to the host vehicle. This is usually not -1.0 (i.e., an equal-and-opposite imparted velocity), to account for the mass fraction between the child and the host as well as any rotational torque acted on the host as a result of deployment direction centerline offsets as compared to the host’s center of gravity.	n/a	-0.05
DEPLOY_DV_CDA	Typical (50 <sup>th</sup> percentile) product of drag coefficient (Cd) times cross-sectional area for the deployed ‘child’ object.	m**2	0.022

## 6.2.9 OCM DATA: PERTURBATIONS SPECIFICATION

**6.2.9.1** Table 6-10 provides an overview of the OCM Perturbations Specification section. Only those keywords shown in table 6-10 shall be used in an OCM Perturbations Specification.

**6.2.9.2** Only one OCM Perturbations Specification section shall appear in an OCM.

**6.2.9.3** The OCM Perturbations Specification section shall be delimited by two keywords: PERT\_START and PERT\_STOP.

**Table 6-10: OCM Data: Perturbations Specification**

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
PERT_START	Start of the perturbations data section.				M
COMMENT	Comments (a contiguous set of one or more comment lines may be provided in the OCM Perturbations Specification only immediately after the PERT_START keyword; see 7.8 for comment formatting rules).			This is a comment	O
ATMOSPHERIC_MODEL	Name of atmosphere model, which shall be selected from the accepted set of values indicated in annex B, subsection B9.			MSISE90 NRLMSIS00 J70 J71 JROBERTS DTM JB2008	O
GRAVITY_MODEL	The gravity model (selected from the accepted set of gravity model names indicated in annex B, subsection B10), followed by the degree (D) and order (O) of the applied spherical harmonic coefficients used in the simulation.  NOTE – Specifying a zero value for ‘order’ (e.g., 2D 0O) denotes zonals (J2 ... JD).			EGM-96: 36D 36O WGS-84: 8D 0O GGM-01: 36D 36O TEG-4: 36D 36O	O
EQUATORIAL_RADIUS	Oblate spheroid equatorial radius of the central body used in the message, if different from the gravity model.	km		6378.137	O
GM	Gravitational coefficient of attracting body (Gravitational Constant x Central Mass), if different from the gravity model.	km**3/ s**2		3.986004e5	O

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
N_BODY_PERTURBATIONS	One OR MORE (N-body) gravitational perturbations bodies used. Values, listed serially in comma-delimited fashion, denote a natural solar or extra-solar system body (stars, planets, asteroids, comets, and natural satellites).  NOTE – Only those entries (or those procedurally added to the CENTER_NAME content as specified in annex B) that are denoted as an ‘Attracting Body’ in annex B, subsection B2 are acceptable values.			MOON, SUN, JUPITER	O
CENTRAL_BODY_ROTATION	Central body angular rotation rate, measured about the major principal axis of the inertia tensor of the central body, relating inertial, and central-body-fixed reference frames.  NOTE – The rotation axis may be slightly offset from the inertial frame Z-axis definition.	deg/s		4.17807421629e-3	O
OBLATE_FLATTENING	Central body’s oblate spheroid oblateness for the polar-symmetric oblate central body model (e.g., for the Earth, it is approximately 1.0/298.257223563).			0.00335281066475	O
OCEAN_TIDES_MODEL	Name of ocean tides model (optionally specify order or constituent effects, diurnal, semi-diurnal, etc.). This is a free-text field, so if the examples on the right are insufficient, others may be used.			DIURNAL SEMI-DIURNAL	O
SOLID_TIDES_MODEL	Name of solid tides model (optionally specify order or constituent effects, diurnal, semi-diurnal, etc.).			DIURNAL SEMI-DIURNAL	O
REDUCTION_THEORY	Specification of the reduction theory used for precession and nutation modeling. This is a free-text field, so if the examples on the right are insufficient, others may be used.			IAU1976/FK5 IAU2010 IERS1996	O
ALBEDO_MODEL	Name of the albedo model.			STK	O
ALBEDO_GRID_SIZE	Number of grid points used in the albedo model.			100	O
SHADOW_MODEL	Shadow model used for Solar Radiation Pressure; dual cone uses both umbra/penumbra regions. Selected option should be one of ‘NONE’, ‘CYLINDRICAL’, ‘CONE’, or ‘DUAL CONE’.			NONE CYLINDRICAL CONE DUAL CONE	O



Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
SHADOW_BODIES	Comma-separated list of planetary bodies for which SRP shadowing is modeled, selected from annex B for CENTER_NAME values.			EARTH MOON	O
SRP_MODEL	Name of SRP model. This is a free-text field, so if the examples on the right are insufficient, others may be used.			GPS_ROCK BOX_WING CANNONBALL COD	O
SW_DATA_SOURCE	Free-text field specifying the source and version of the Space Weather data used in the creation of this message. Multiple space weather sources can be specified in a comma-delimited fashion.			CELESTRAK	O
SW_DATA_EPOCH	Epoch of the Space Weather data.			2001-11-08T00:00:00	O
SW_INTERP_METHOD	Free-text field specifying the method used to select or interpolate any and all sequential space weather data ( $K_p$ , $a_p$ , Dst, $F_{10.7}$ , $M_{10.7}$ , $S_{10.7}$ , $Y_{10.7}$ , etc.). While not constrained to specific entries, it is anticipated that the utilized method would match methods detailed in numerical analysis textbooks.			PRECEDING_VALUE NEAREST_NEIGHBOR LINEAR LAGRANGE_ORDER_5	O
FIXED_GEOMAG_KP	A fixed (time invariant) value of the planetary geomagnetic index $K_p$ used to override the normal time varying $K_p$ values (e.g., obtained from SW_DATA_SOURCE).  NOTE – The use of $K_p$ or $A_p$ would depend on the selected ATMOSPHERIC_MODEL.	nT		3.2	O
FIXED_GEOMAG_AP	A fixed (time invariant) value of the geomagnetic index $a_p$ used to override the normal time-varying $a_p$ values (e.g., obtained from SW_DATA_SOURCE).  NOTE – The use of $K_p$ or $A_p$ would depend on the selected ATMOSPHERIC_MODEL.	nT		21	O
FIXED_GEOMAG_DST	A fixed (time invariant) value of the planetary geomagnetic index Dst used to override the normal time varying daily Dst values (e.g., obtained from SW_DATA_SOURCE).	nT		-20	O

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
FIXED_F10P7	A fixed (time invariant) value of the Solar Flux Unit (SFU) daily proxy $F_{10.7}$ used to override the normal time varying daily $F_{10.7}$ values (e.g., obtained from SW_DATA_SOURCE).	SFU		120.0	O
FIXED_F10P7_MEAN	A fixed (time invariant) value of the solar flux proxy $F_{10.7}$ used to override the normal time varying averaged $F_{10.7}$ values (e.g., obtained from SW_DATA_SOURCE).	SFU		132.0	O
FIXED_M10P7	A fixed (time invariant) value of the solar flux daily proxy $M_{10.7}$ used to override the normal time varying daily $M_{10.7}$ values (e.g., obtained from SW_DATA_SOURCE).	SFU		120.0	O
FIXED_M10P7_MEAN	A fixed (time invariant) value of the solar flux proxy $M_{10.7}$ used to override the normal time varying averaged $M_{10.7}$ values (e.g., obtained from SW_DATA_SOURCE).	SFU		120.0	O
FIXED_S10P7	A fixed (time invariant) value of the solar flux proxy $S_{10.7}$ used to override the normal time varying daily $S_{10.7}$ values (e.g., obtained from SW_DATA_SOURCE).	SFU		120.0	O
FIXED_S10P7_MEAN	A fixed (time invariant) value of the solar flux proxy $S_{10.7}$ used to override the normal time varying averaged $S_{10.7}$ values (e.g., obtained from SW_DATA_SOURCE).	SFU		120.0	O
FIXED_Y10P7	A fixed (time invariant) value of the solar flux proxy $Y_{10.7}$ used to override the normal time varying daily $Y_{10.7}$ values (e.g., obtained from SW_DATA_SOURCE).	SFU		120.0	O
FIXED_Y10P7_MEAN	A fixed (time invariant) value of the solar flux proxy $Y_{10.7}$ used to override the normal time varying averaged $Y_{10.7}$ values (e.g., obtained from SW_DATA_SOURCE).	SFU		120.0	O
PERT_STOP	End of the perturbations section.				M

## **6.2.10 OCM DATA: ORBIT DETERMINATION DATA**

**6.2.10.1** Table 6-11 provides an overview of the OCM orbit determination data section. Only those keywords shown in table 6-11 shall be used in an OCM orbit determination data specification.

**6.2.10.2** At most, only one Orbit Determination Data section shall appear in an OCM.

**6.2.10.3** Orbit determination data in the OCM shall be indicated by two keywords: OD\_START and OD\_STOP.

**6.2.10.4** The values of the DAYS\_SINCE\_FIRST\_OBS, and DAYS\_SINCE\_LAST\_OBS keywords shall be specified as relative time, in days, to the value of the OD\_EPOCH keyword.

**6.2.10.5** If an orbit determination parameters section is included in the message, a corresponding perturbations section shall be included as well to specify the perturbations incorporated in the orbit determination.

**6.2.10.6** When these orbit determination settings match those used to generate an OCM orbit, covariance, and/or maneuver time history, the OD\_ID should match the TRAJ\_BASIS\_ID, COV\_BASIS\_ID, and/or MAN\_BASIS\_ID keyword values respectively.

**Table 6-11: OCM Data: Orbit Determination Data**

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
OD_START	Start of the orbit determination data section.			n/a	M
COMMENT	Comments (a contiguous set of one or more comment lines may be provided in the OCM Orbit Determination Data section only immediately after the OD_START keyword; see 7.8 for comment formatting rules).			This is a comment	O
OD_ID	Identification number for this orbit determination.			OD_20160402	M
OD_PREV_ID	Optional identification number for the previous orbit determination.  NOTE – If this orbit determination is the first one performed on this object, then OD_PREV_ID should be excluded from this message.			OD_20160401	O
OD_METHOD	Type of orbit determination method used to produce the orbit estimate. While this is a free-text field, it is suggested that it comprise the method, followed by a colon delimiter and the actual OD tool used to estimate the orbit (e.g., BAHN, ODIN, ODTK).  NOTE – Commonly used methods include Batch Weighted Least Squares (BWLS), Extended Kalman Filter (EKF), Sequential Filter (SF), Square Root Information Filter (SRIF), Sequential Simultaneous Estimation Method (SSEM).			BWLS: BAHN BWLS: ODIN SF: ODTK	M
OD_EPOCH	Relative or absolute time tag of the orbit determination solved-for state in the selected OCM time system specified by the TIME_SYSTEM keyword.			2001-11-06T11:17:33 27854.239	M
DAYS_SINCE_FIRST_OBS	Days elapsed between first accepted observation and OD_EPOCH. NOTE – May be positive or negative.	d		3.5	O
DAYS_SINCE_LAST_OBS	Days elapsed between last accepted observation and OD_EPOCH. NOTE – May be positive or negative.	d		1.2	O
RECOMMENDED_OD_SPAN	Number of days of observations recommended for the OD of the object ( <i>useful only for Batch OD systems</i> ).	d		5.2	O

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
ACTUAL_OD_SPAN	Actual time span in days used for the OD of the object. NOTE – Should equal (DAYS_SINCE_FIRST_OBS - DAYS_SINCE_LAST_OBS).	d		2.3	O
OBS_AVAILABLE	The number of observations available within the actual OD time span.			100	O
OBS_USED	The number of observations accepted within the actual OD time span.			90	O
TRACKS_AVAILABLE	The number of sensor tracks available for the OD within the actual time span (see definition of ‘tracks’, 1.5.2).			33	O
TRACKS_USED	The number of sensor tracks accepted for the OD within the actual time span (see definition of ‘tracks’, 1.5.2).			30	O
MAXIMUM_OBS_GAP	The maximum time between observations in the OD of the object.	d		1.0	O
OD_EPOCH_EIGMAJ	Positional error ellipsoid $1\sigma$ major eigenvalue at the epoch of the OD.	m		58.73	O
OD_EPOCH_EIGMED	Positional error ellipsoid $1\sigma$ intermediate eigenvalue at the epoch of the OD.	m		35.7	O
OD_EPOCH_EIGMIN	Positional error ellipsoid $1\sigma$ minor eigenvalue at the epoch of the OD.	m		21.5	O
OD_MAX_PRED_EIGMAJ	The resulting maximum <b>predicted</b> major eigenvalue of the $1\sigma$ positional error ellipsoid over the entire TIME_SPAN of the OCM, stemming from this OD.	m		21.5	O
OD_MIN_PRED_EIGMIN	The resulting minimum <b>predicted</b> minor eigenvalue of the $1\sigma$ positional error ellipsoid over the entire TIME_SPAN of the OCM, stemming from this OD.	m		21.5	O
OD_CONFIDENCE	OD confidence metric, which spans 0 to 100% (useful only for Filter-based OD systems). The OD confidence metric shall be as mutually defined by message exchange participants.	%		95.3	O

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
GDOP	Generalized Dilution Of Precision for this orbit determination, based on the observability grammian as defined in references [H15] and [H16] and expressed in informative annex F, subsection F4. GDOP provides a rating metric of the observability of the element set from the OD. Alternate GDOP formations may be used as mutually defined by message exchange participants.			.857	O
SOLVE_N	The number of solve-for states in the orbit determination.			6	O
SOLVE_STATES	Free-text comma-delimited description of the state elements solved for in the orbit determination.			POS[3], VEL[3]	O
CONSIDER_N	The number of consider parameters used in the orbit determination.			2	O
CONSIDER_PARAMS	Free-text comma-delimited description of the consider parameters used in the orbit determination.			DRAG, SRP	O
SEDR	The Specific Energy Dissipation Rate, which is the amount of energy being removed from the object's orbit by the non-conservative forces. This value is an average calculated during the OD. (See annex F, subsection F7 for definition.)	W/kg		4.54570E-05	O
SENSORS_N	The number of sensors used in the orbit determination.			3	O
SENSORS	Free-text comma-delimited description of the sensors used in the orbit determination.			EGLIN, FYLINGDALES	O
WEIGHTED_RMS	<p><i>(Useful/valid only for Batch OD systems.)</i></p> <p>The weighted RMS residual ratio, defined as:</p> $Weighted\ RMS = \sqrt{\frac{\sum_{i=1}^N w_i (y_i - \hat{y}_i)^2}{N}}$ <p>Where <math>y_i</math> is the <math>i</math>th observation measurement,  <math>\hat{y}_i</math> is the current estimate of <math>y_i</math>,  <math>w_i = \frac{1}{\sigma_i^2}</math> is the weight (sigma) associated with the measurement at the <math>i</math>th time and <math>N</math> is the number of observations.</p> <p>This is a value that can generally identify the quality of the most recent vector update and is used by the analyst in evaluating the OD process. A value of 1.00 is ideal.</p>	(measurement units)		1.3	O

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
DATA_TYPES	Comma-separated list of observation data types utilized in this orbit determination. Although this is a free-text field, it is recommended at a minimum to use data type descriptor(s) as provided in table 3-5 of the TDM standard (reference [9]) (excluding the DATA_START, DATA_STOP, and COMMENT keywords). Additional descriptors/detail is encouraged if the descriptors of table 3-5 are not sufficiently clear; for example, one could replace ANGLE_1 and ANGLE_2 with RADEC (e.g., from a telescope), AZEL (e.g., from a ground radar), RANGE (whether from radar or laser ranging), etc.			ANGLE_1, ANGLE_2	O
OD_STOP	End of the orbit determination data section.			n/a	M

**6.2.11 OCM DATA: USER-DEFINED PARAMETERS**

**6.2.11.1** A single section of User-Defined Parameters may be provided if necessary. In principle, this provides flexibility, but also introduces complexity, non-standardization, potential ambiguity, and potential processing errors. Accordingly, if used, the keywords and their meanings must be described in an ICD. User-Defined Parameters, if included, should be used as sparingly as possible; their use is not encouraged.

**6.2.11.2** At most, only one User-Defined Parameters section shall appear in an OCM.

**6.2.11.3** Table 6-12 provides an overview of the OCM user-defined data section. Only those keywords shown in table 6-12 shall be used in an OCM user-defined data specification.

**Table 6-12: OCM Data: User-Defined Parameters**

Keyword	Description	Units	Examples of Values	M/O/C
USER_START	Start of the User-Defined Parameters data block.			M
COMMENT	Comments (a contiguous set of one or more comment lines may be provided immediately following the USER_START keyword). (See 7.8 for formatting rules.)		This is a comment	O
USER_DEFINED_x	User-defined parameter, where 'x' is replaced by a variable-length user-specified character string. Any number of user-defined parameters may be included, if necessary, to provide essential information that cannot be conveyed in COMMENT statements.		USER_DEFINED_EARTH_MODEL = WGS-84	M
USER_STOP	End of the User-Defined Parameters data block.			M



## **7 ORBIT DATA MESSAGE SYNTAX**

### **7.1 OVERVIEW**

This section details the syntax requirements for each of the Orbit Data Messages.

### **7.2 GENERAL**

The Orbit Data Messages (OPM, OMM, OEM, and OCM) shall observe the syntax described in 7.3 through 7.8.

### **7.3 ODM LINES**

**7.3.1** Each OPM, OMM, OEM, or OCM line shall be one of the following:

- Header line;
- Metadata line;
- Data line;
- Comment line; or
- Blank line.

**7.3.2** Each OPM, OMM, or OEM line must not exceed 254 ASCII characters and spaces (excluding line termination character[s]).

**7.3.3** OCM lines may be of arbitrary length. If exchange between the two parties requires a maximum line length, that limit should be mutually agreed upon between message exchange partners.

**7.3.4** Only printable ASCII characters and blanks shall be used. Control characters (such as TAB, etc.) shall not be used, except for the line termination characters specified below.

**7.3.5** Blank lines may be used at any position within the file. Blank lines shall have no assignable meaning and may be ignored.

**7.3.6** The first header line must be the first non-blank line in the file.

**7.3.7** All lines shall be terminated by a single Carriage Return or a single Line Feed, a Carriage Return/Line Feed pair, or a Line Feed/Carriage Return pair.

## 7.4 ORBIT DATA MESSAGES IN ‘KEYWORD = VALUE NOTATION’ (I.E., NON-XML) AND ORDER OF ASSIGNMENT STATEMENTS

**7.4.1** For the OPM and OMM, all header, metadata, and data lines shall use ‘keyword = value’ notation, abbreviated as KVN.

**7.4.1.1** For the OEM, all header and metadata elements shall use KVN notation.

**7.4.1.2** OEM ephemeris data lines shall not use KVN format; rather, the OEM ephemeris data line has a fixed structure containing seven required fields (epoch time, three position components, three velocity components), and three optional acceleration components. (See 5.2.4.)

**7.4.1.3** OEM covariance matrix epoch and covariance reference frame (if used) shall use KVN format. The OEM covariance data lines shall not use KVN format; rather, the OEM covariance data line has a fixed structure containing from one to six required fields (a row from the 6x6 lower triangular form covariance matrix). (See 5.2.5.)

**7.4.1.4** For the OCM, all header and metadata elements shall use KVN notation.

**7.4.1.5** OCM trajectory state time history data lines shall not use KVN format; rather, the structure of such OCM trajectory state time history data shall comprise a contiguous set of lines, with the values on each line separated by at least one white space character, and those values consisting of the time tag followed by the parameters corresponding to the selected orbit set (see TRAJ\_TYPE, 6.2.5).

**7.4.1.6** OCM covariance matrix epoch and covariance reference frame (if used) shall use KVN format. The OCM covariance data lines shall not use KVN format; rather, OCM covariance time history data shall comprise a contiguous set of lines, with the values on each line separated by at least one white space character, and those values consisting of the time tag followed by the covariance matrix corresponding to the selected covariance type (see COV\_TYPE, 6.2.7, particularly 6.2.7.10 through 6.2.7.12).

**7.4.1.7** OCM maneuver data lines shall not use KVN format; rather, OCM maneuver data shall comprise a contiguous set of lines, the values on each line separated by at least one white space character, and with those values consisting of the specified maneuver parameters (see MAN\_COMPOSITION, 6.2.8.14).

**7.4.2** The keywords ‘COMMENT’, \*‘\_START’, and \*‘\_STOP’ are exceptions to the KVN syntax assignment.

**7.4.3** Only a single ‘keyword = value’ assignment shall be made on a line.

**7.4.4** Keywords must be uppercase and must not contain blanks.

**7.4.5** Any white space immediately preceding or following the keyword shall not be significant.

**7.4.6** Any white space immediately preceding or following the ‘equals’ sign shall not be significant.

**7.4.7** Any white space immediately preceding the end of line shall not be significant.

**7.4.8** The order of occurrence of mandatory and optional KVN assignments shall be fixed as shown in the tables in sections 3, 4, 5, and 6 that describe the OPM, OMM, OEM, and OCM keywords.

## **7.5 VALUES**

**7.5.1** A non-empty value field must be assigned to each mandatory keyword except for \*‘\_START’ and \*‘\_STOP’ keyword values.

**7.5.2** Comments and free-text value fields may be in any case (or mix of case) desired by the user.

**7.5.3** Apart from comments and free-text fields, normative text value fields shall be constructed using only exclusively all uppercase or exclusively all lowercase.

**7.5.4** Integer values shall consist of a sequence of decimal digits with an optional leading sign (‘+’ or ‘-’). If the sign is omitted, ‘+’ shall be assumed. Leading zeroes may be used. The range of values that may be expressed as an integer is:

$$-2,147,483,648 \leq x \leq +2,147,483,647 \text{ (i.e., } -2^{31} \leq x \leq 2^{31}-1\text{)}.$$

NOTE – The commas in the range of values above are thousands separators and are used only for readability. They should not appear in an actual message.

**7.5.5** Non-integer numeric values may be expressed in either fixed-point or floating-point notation. Both representations may be used within an OPM, OMM, OEM, or OCM.

**7.5.6** Non-integer numeric values expressed in fixed-point notation shall consist of a sequence of decimal digits separated by a period as a decimal point indicator, with an optional leading sign (‘+’ or ‘-’). If the sign is omitted, ‘+’ shall be assumed. Leading and trailing zeroes may be used. At least one digit shall appear before and after a decimal point. The number of digits shall be 16 or fewer.

**7.5.7** Non-integer numeric values expressed in floating point notation shall consist of a sign, a mantissa, an alphabetic character indicating the division between the mantissa and exponent, and an exponent, constructed according to the following rules:

- a) The sign may be ‘+’ or ‘-’. If the sign is omitted, ‘+’ shall be assumed.
- b) The mantissa must be a string of no more than 16 decimal digits with a decimal point (‘.’) in the second position of the ASCII string, separating the integer portion of the mantissa from the fractional part of the mantissa.

- c) The character used to denote exponentiation shall be ‘E’ or ‘e’. If the character indicating the exponent and the following exponent are omitted, an exponent value of zero shall be assumed (essentially yielding a fixed-point value).
- d) The exponent must be an integer and may have either a ‘+’ or ‘-’ sign (if the sign is omitted, then ‘+’ shall be assumed).
- e) The maximum positive floating-point value is approximately 1.798E+308, with 16 significant decimal digits precision. The minimum positive floating-point value is approximately 4.94E-324, with 16 significant decimal digits precision.

**7.5.8** Blanks shall not be permitted within numeric values and time strings.

**7.5.9** In value fields that are text, an underscore shall be equivalent to a single blank. Individual blanks shall be retained (shall be significant), but multiple contiguous blanks shall be equivalent to a single blank.

**7.5.10** In value fields that represent an absolute time tag or epoch, times shall be given in one of the following two formats:

YYYY-MM-DDThh:mm:ss[.d→d][Z]

or

YYYY-DDDThh:mm:ss[.d→d][Z],

where ‘YYYY’ is the year; ‘MM’ is the two-digit month; ‘DD’ is the two-digit day; ‘DDD’ is the three-digit day of year; ‘T’ is constant; ‘hh:mm:ss[.d→d]’ is the time in hours, minutes, seconds, and optional fractional seconds; and ‘Z’ is an optional time code terminator (the only permitted value is ‘Z’ for Zulu, i.e., UTC). As many ‘d’ characters to the right of the period as required may be used to obtain the required precision, up to the maximum allowed for a fixed-point number. All fields shall have leading zeros. (See reference [2], ASCII Time Code A or B.)

NOTE – During a leap second introduction, the value of the two-digit integer seconds (ss) field shall be ‘60’ as specified in reference [2].

**7.5.11** The time system for CREATION\_DATE is UTC; for all other keywords representing times or epochs, the time system is determined by the TIME\_SYSTEM metadata keyword.

## 7.6 OCM VECTOR DATA TYPE

**7.6.1** Several OCM keywords may be set to values containing more than one number. Examples include DC\_REF\_DIR and DC\_BODY\_TRIGGER. Such vectors shall be space-delimited and provided serially on a single line following the equals ‘=’ sign, adhering to the requirements for ‘numeric values’ provided in the previous sections.

## 7.7 UNITS IN THE ORBIT DATA MESSAGES

### 7.7.1 OPM/OMM UNITS

**7.7.1.1** For documentation purposes and clarity only, units may be included as ASCII text after a value in the OPM and OMM. If units are displayed, they must exactly match the units (including lower/upper case) as specified in tables 3-3 and 4-3. If units are displayed, then:

- a) there must be at least one blank character between the value and the units text;
- b) the units must be enclosed within square brackets (e.g., '[km]');
- c) combinations of units shall adhere to requirements listed in 1.5.

**7.7.1.2** Some of the items in the applicable tables are dimensionless. The table shows a unit value of 'n/a', which in this case means that there is no applicable units designator for these items (e.g., for ECCENTRICITY).

**7.7.1.3** The notation '[n/a]' shall not appear in an OPM or OMM.

### 7.7.2 OEM UNITS

**7.7.2.1** In an OEM ephemeris data line, units shall be km, km/s, and km/s\*\*2 for position, velocity, and acceleration components, respectively, but the units shall not be displayed.

**7.7.2.2** In an OEM covariance matrix line, units shall be km\*\*2, km\*\*2/s, or km\*\*2/s\*\*2, depending on whether the element is computed from two position components, one position component and one velocity component, or two velocity components. The units shall not be displayed.

### 7.7.3 OCM UNITS

**7.7.3.1** Apart from trajectory state, covariance, and maneuver data lines, units of OCM keyword values shall correspond to the normative 'Units' column of the accompanying Keyword Value Tables (i.e., tables 6-3 through 6-12) for each section definition.

NOTE – The units used throughout the OCM are generally a combination of kilometers for distance and seconds for time (e.g., km/s for velocity, km/s\*\*2 for acceleration, and so forth). Mass is in kilograms, and force is in Newtons.

**7.7.3.2** The units of orbit time state history data lines, when present, shall adhere to the specified units for trajectory states as provided in the SANA Registry (reference [11]) for Orbital Elements (annex B, subsection B7).

**7.7.3.3** The units of covariance time history data lines, when present, shall adhere to the specified units for covariance data as provided in the SANA Registry (reference [11]) for

Orbital Elements (annex B, subsection B7) and Additional Covariance Representations (annex B, subsection B8).

**7.7.3.4** The units of maneuver time history data lines, when present, shall adhere to the specified units for maneuver lines as provided in table 6-8 and table 6-9.

**7.7.3.5** For OCM keywords used to convey multipartite trajectory state, covariance, or maneuver data lines, units may accompany these data lines via the TRAJ\_UNITS, COV\_UNITS, and MAN\_UNITS keywords, respectively. Units shall not be displayed in OCM trajectory state, covariance, or maneuver data lines themselves.

**7.7.3.6** For OCM keywords that are not used to convey multipartite trajectory state, covariance, or maneuver data lines, units may be included as ASCII text after a value in the OCM for documentation purposes and clarity only. If units are displayed, then:

- a) there must be at least one blank character between the value and the units text;
- b) the units must be enclosed within square brackets (e.g., '[m]');
- c) combinations of units shall adhere to requirements listed in 1.5.

**7.7.3.7** Some of the items in the applicable tables are dimensionless. The table shows a unit value of 'n/a', which in this case means that there is no applicable units designator for these items (e.g., for ECCENTRICITY) and no units displayed.

## **7.8 COMMENTS IN THE ORBIT DATA MESSAGES**

**7.8.1** There are certain pieces of information that provide clarity and remove ambiguity about the interpretation of the information in a file, yet are not standardized so as to fit cleanly into the 'keyword = value' paradigm. Rather than force the information to fit into a space limited to one line, the ODM producer should put further specifications and information into comments. Static information should be separately shared and/or mutually agreed upon between message exchange partners outside of the ODM.

**7.8.2** Comments may be used to provide provenance information or to help describe dynamical events or other pertinent information associated with the data. This additional information is intended to aid in consistency checks and elaboration when needed but shall not be required for successful processing of a file.

**7.8.3** For the OPM, OMM, OEM, and OCM, comment lines shall be optional.

**7.8.4** Comment text may be in any case (or mix of case) desired by the user.

**7.8.5** All comment lines shall begin with the 'COMMENT' keyword followed by at least one space. This keyword must appear on every comment line, not just the first such line. The remainder of the line shall be the comment value. White space shall be retained (shall be significant) in comment values.

**7.8.6** Placement of comments shall be as specified in the tables in sections 3, 4, 5, and 6 that describe the OPM, OMM, OEM, and OCM keywords.

**7.8.7** Comments in the OPM may appear in the OPM Header immediately after the ‘CCSDS\_OPM\_VERS’ keyword, at the very beginning of the OPM Metadata section, and at the beginning of a logical block in the OPM Data section. Comments must not appear between the components of any logical block in the OPM Data section.

NOTE – The logical blocks in the OPM Data section are indicated in table 3-3.

**7.8.8** Comments in the OMM may appear in the OMM Header immediately after the ‘CCSDS\_OMM\_VERS’ keyword, at the very beginning of the OMM Metadata section, and at the beginning of a logical block in the OMM Data section. Comments must not appear between the components of any logical block in the OMM Data section.

NOTE – The logical blocks in the OMM Data section are indicated in table 4-3.

**7.8.9** Comments in the OEM may appear in the OEM Header immediately after the ‘CCSDS\_OEM\_VERS’ keyword, at the very beginning of the OEM Metadata section (after the ‘META\_START’ keyword), at the beginning of the OEM Ephemeris Data Section, and at the beginning of the OEM Covariance Data section (after the ‘COV\_START’ keyword). Comment lines must not appear within any block of ephemeris lines or covariance matrix lines.

**7.8.10** Comments may appear in all logical blocks of the OCM, but only at the positions shown in the defining tables (generally at the top of each section, following the \*\_START section delimiting keyword).

**7.8.11** Extensive comments in an ODM are recommended in cases when that content is germane to the message and changes from message to message.

**7.8.12** The following comments should be provided:

- a) Information regarding the genesis, history, interpretation, intended use, etc., of the state vector, spacecraft, maneuver, or ephemeris that may be of use to the receiver of the OPM, OMM, OEM, or OCM:

COMMENT Source: File created by JPL Multi-Mission Navigation Team as part  
COMMENT of Launch Operations Readiness Test held on 20 April 2001.

- b) Natural body ephemeris information: When the Earth is not the center of motion, the ephemerides of the planets, satellites, asteroids, and/or comets (including associated constants) consistent with the ODM should be identified so that the recipient can, in a consistent manner, make computations involving other centers:

COMMENT Based on latest orbit solution which includes observations  
COMMENT through 2000-May-15 relative to planetary ephemeris DE-0405.

- c) OEM accuracy vs. efficiency: If the covariance data section of the OEM is not utilized, the producer of an OEM should report in comment lines what the expected accuracy of the ephemeris is, so the user can smooth or otherwise compress the data without affecting the accuracy of the trajectory. The OEM producer also should strive to achieve not only the best accuracy possible, considering prediction errors, but also consider the efficiency of the trajectory representation (e.g., step sizes of fractional seconds between ephemeris lines may be necessary for precision scientific reconstruction of an orbit, but are excessive from the standpoint of antenna pointing predicts generation).

## 7.9 ORBIT DATA MESSAGE KEYWORDS

### 7.9.1 VERSION KEYWORDS

The Header of the OPM, OMM, OEM, and OCM shall provide a CCSDS Orbit Data Message version number that identifies the format version; this is included to anticipate future changes. The version keywords for the OPM, OMM, OEM, and OCM shall be `CCSDS_OPM_VERS`, `CCSDS_OMM_VERS`, `CCSDS_OEM_VERS`, and `CCSDS_OCM_VERS`, respectively. The value shall have the form of 'x.y', where 'y' shall be incremented for corrections and minor changes, and 'x' shall be incremented for major changes. Version x.0 shall be reserved for versions accepted by the CCSDS as an official Recommended Standard ('Blue Book'). Testing shall be conducted using OPM, OMM, OEM, and OCM version numbers less than 1.0 (e.g., 0.x). The specific OPM, OMM, OEM, and OCM version numbers to be used should be mutually agreed between message exchange partners. The following version numbers are supported (Blue Book) or have been supported in the past (Silver Book):

Version Keyword	Version Number	Applicable Recommendation
<code>CCSDS_OPM_VERS</code>	1.0	Silver Book 1.0, 09/2004
<code>CCSDS_OPM_VERS</code>	2.0	Silver Book 2.0, 11/2009
<code>CCSDS_OPM_VERS</code>	3.0	Blue Book 3.0 (this document)
<code>CCSDS_OMM_VERS</code>	2.0	Silver Book 2.0, 11/2009
<code>CCSDS_OMM_VERS</code>	3.0	Blue Book 3.0 (this document)
<code>CCSDS_OEM_VERS</code>	1.0	Silver Book 1.0, 09/2004
<code>CCSDS_OEM_VERS</code>	2.0	Silver Book 2.0, 11/2009
<code>CCSDS_OEM_VERS</code>	3.0	Blue Book 3.0 (this document)



CCSDS_OCM_VERS	3.0	Blue Book 3.0 (this document)
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## 7.9.2 GENERAL KEYWORDS

**7.9.2.1** Only those keywords shown in tables 3-1, 3-2, and 3-3 shall be used in an OPM. Some keywords represent mandatory items, and some are optional. KVN assignments representing optional items may be omitted.

**7.9.2.2** Only those keywords shown in tables 4-1, 4-2, and 4-3 shall be used in an OMM. Some keywords represent mandatory items, and some are optional. KVN assignments representing optional items may be omitted.

**7.9.2.3** Only those keywords shown in tables 5-2 and 5-3 shall be used in an OEM. Some keywords represent mandatory items, and some are optional. KVN assignments representing optional items may be omitted.

**7.9.2.4** Only those keywords shown in tables 6-3 through 6-12 shall be used in an OCM. Some keywords represent mandatory items, and some are optional. KVN assignments representing optional items may be omitted.

## 7.10 VALIDATION AND INGEST OF KVN CONTENT VIA REGULAR EXPRESSIONS (OR 'REGEX')

### 7.10.1 BENEFITS OF USING REGULAR EXPRESSIONS WITH THE ODM

Unlike the schema validation feature native to XML versions of this message as described in section 8, the KVN version of this message does not natively support such validations of KVN content. To accomplish validation and ingest of KVN versions of the ODM, the use of Regular Expressions (referred to as 'Regex') is strongly encouraged where possible. Most programming languages support the Regex feature, and Regex offers a detailed and rigorous way to ensure proper validation, interpretation, and conformance to Orbit Data Message content.

### 7.10.2 SAMPLE REGULAR EXPRESSIONS FOR CCSDS MESSAGES

To facilitate the use of Regular Expressions when processing CCSDS Navigation Messages, informative annex F, subsection F6 provides sample Regex patterns to rigorously match a variety of common KVN sequences.

## 8 CONSTRUCTING AN ODM/XML INSTANCE

### 8.1 OVERVIEW

This section provides detailed instructions for the user on how to create an XML message (reference [5]) based on one of the KVN-formatted messages described in sections 3, 4, 5, and 6. This section applies only to the XML representation of ODMs.

Overall information on using XML for Navigation Data Messages is provided in reference [5]. The ODM/XML schemas are available on the SANA Web site. SANA is the registrar for the protocol registries created under CCSDS. The ODM/XML schemas explicitly define the permitted data elements and values acceptable for the XML versions of the ODMs. The location of the ODM/XML schemas is:

OPM: [https://sanaregistry.org/files/ndmxml\\_unqualified/ndmxml-3.0.0-opm-3.0.xsd](https://sanaregistry.org/files/ndmxml_unqualified/ndmxml-3.0.0-opm-3.0.xsd)

OMM: [https://sanaregistry.org/files/ndmxml\\_unqualified/ndmxml-3.0.0-omm-3.0.xsd](https://sanaregistry.org/files/ndmxml_unqualified/ndmxml-3.0.0-omm-3.0.xsd)

OEM: [https://sanaregistry.org/files/ndmxml\\_unqualified/ndmxml-3.0.0-oem-3.0.xsd](https://sanaregistry.org/files/ndmxml_unqualified/ndmxml-3.0.0-oem-3.0.xsd)

OCM: [https://sanaregistry.org/files/ndmxml\\_unqualified/ndmxml-3.0.0-ocm-3.0.xsd](https://sanaregistry.org/files/ndmxml_unqualified/ndmxml-3.0.0-ocm-3.0.xsd)

Figure 8-1 illustrates the basic structure of an ODM/XML instance. Defined structural elements are the header and body. The body then consists of one or more segments depending on the message type (one for the OPM, OMM, and OCM; one or more for the OEM). Each segment consists of a <metadata>/<data> pair. In an OEM, which could have more than one segment, the metadata/data pair is repeated in each segment.

```

<header>
</header>
<body>
  <segment>
    <metadata>
    </metadata>
    <data>
    </data>
  </segment>
</body>

```

**Figure 8-1: ODM/XML Basic Structure**

ODM/XML tags for keywords defined in sections 3, 4, 5, and 6 appear just as in the KVN, that is, all capital letters. Tags related to the XML message structure (i.e., that do not correspond directly to a KVN keyword) appear in ‘lowerCamelCase’ (e.g., <header>, <segment>, <metadata>, <stateVector>, <covarianceMatrix>, etc.).

## 8.2 XML VERSION

This section describes the Extensible Markup Language (or XML) version of the Orbit Data Message. The first line of each XML instantiation shall specify the XML version:

```
<?xml version="1.0" encoding="UTF-8"?>
```

This line must appear on the first line of each instantiation, exactly as shown.

## 8.3 BEGINNING THE INSTANTIATION: ROOT ELEMENT TAG

**8.3.1** Each instantiation shall have a ‘root element tag’ that identifies the message type and other information such as where to find the applicable schema, required attributes, etc.

**8.3.2** The root element tag in an ODM/XML instantiation shall be one of those listed in table 8-1.

**Table 8-1: ODM/XML Root Element Tags**

Root Element Tag	Message Type
<opm></opm>	Orbit Parameter Message
<omm></omm>	Orbit Mean Elements Message
<oem></oem>	Orbit Ephemeris Message
<ocm></ocm>	Orbit Comprehensive Message

**8.3.3** The XML Schema Instance namespace attribute must appear in the root element tag of all ODM/XML instantiations, exactly as shown:

```
xmlns:xsi="http://www.w3.org/2001/XMLSchema-instance"
```

If it is desired to validate an instantiation against the CCSDS Web-based schema, the `xsi:noNamespaceSchemaLocation` attribute must be coded as a single string of non-blank characters, with no line breaks exactly as shown:

```
xsi:noNamespaceSchemaLocation=https://sanaregistry.org/r/ndmxml\_unqualified/ndmxml-3.0.0-master-3.0.xsd
```

and

```
xsi:noNamespaceSchemaLocation=https://sanaregistry.org/r/ndmxml\_qualified/ndmxml-3.0.0-master-3.0.xsd
```

NOTE – The value associated with the `xsi:noNamespaceSchemaLocation` attribute shown in this document is too long to appear on a single line.

**8.3.4** For use in a local operations environment, the schema set may be downloaded from the SANA website to a local server that meets local requirements for operations robustness.

**8.3.5** If a local version is used, the value associated with the `xsi:noNamespaceSchemaLocation` attribute must be changed to a URL that is accessible to the local server.

**8.3.6** Two attributes shall appear in the root element tag of an ODM/XML single message instantiation, specifically, the `CCSDS_XXX_VERS` keyword that is also part of the standard KVN header, and the Blue Book version number. The final attributes of the root element tag shall be 'id' and 'version'.

**8.3.7** The `CCSDS_XXX_VERS` keyword shall be supplied via the 'id' attribute of the root element tag. The 'id' attribute shall be `'id="CCSDS_XXX_VERS"'`, where `xxx` = OPM, OMM, OEM, or OCM.

**8.3.8** The version number of the Blue Book to which the schema applies shall be supplied via the 'version' attribute. The 'version' attribute shall be `'version="3.0"'`.

NOTE – The following example root element tag for an OPM instantiation combines all the directions in the preceding several sections:

```
<?xml version="1.0" encoding="UTF-8"?>
<opm xmlns:xsi="http://www.w3.org/2001/XMLSchema-instance"
xsi:noNamespaceSchemaLocation="https://sanaregistry.org/r/ndmxml_
unqualified/ndmxml-3.0.0-master-3.0.xsd"
id="CCSDS_OPM_VERS" version="3.0">
```

and

```
<?xml version="1.0" encoding="UTF-8"?>
<opm xmlns:xsi="http://www.w3.org/2001/XMLSchema-instance"
xsi:noNamespaceSchemaLocation="https://sanaregistry.org/r/ndmxml_
qualified/ndmxml-3.0.0-master-3.0.xsd"
id="CCSDS_OPM_VERS" version="3.0">
```

## 8.4 THE STANDARD ODM/XML HEADER SECTION

**8.4.1** The ODMs shall share a standard header section, with tags `<header>` and `</header>`.

**8.4.2** Immediately following the `<header>` tag, the message may have any number of `<COMMENT>` elements.

**8.4.3** The standard ODM header shall contain the `<CREATION_DATE>` and the `<ORIGINATOR>` elements.

**8.4.4** The standard ODM header may contain the `<MESSAGE_ID>` element.

NOTE – An example <header> section is shown immediately below.

```

<header>
  <COMMENT>This is the common ODM/XML header</COMMENT>
  <COMMENT>I can put as many comments here as I want,
    </COMMENT>
  <COMMENT>including none. </COMMENT>
  <CREATION_DATE>2004-281T17:26:06</CREATION_DATE>
  <ORIGINATOR>AGENCY-X</ORIGINATOR>
  <MESSAGE_ID>XYZ123-2019</MESSAGE_ID>
</header>

```

## 8.5 THE ODM/XML BODY SECTION

**8.5.1** After coding the <header>, the instantiation must include a <body> section.

**8.5.2** Inside the <body> section must appear at least one <segment> section.

**8.5.3** Each segment must be made up of one or more <metadata> and <data> sections, depending on the specific message type.

## 8.6 THE ODM/XML METADATA SECTION

**8.6.1** All ODMs must have a metadata section.

**8.6.2** The metadata section shall be delimited by the <metadata> element.

**8.6.3** Between the <metadata> and </metadata> tags, the keywords shall be the same as those in the metadata sections in sections 3, 4, 5, and 6, with possible exceptions as noted in the sections below that discuss creating instantiations of the specific messages.

## 8.7 THE ODM/XML DATA SECTION

**8.7.1** All ODMs must have a data section.

**8.7.2** The data section shall follow the metadata section and shall be delimited by the <data> element.

**8.7.3** Between the <data> and </data> tags, the keywords shall be the same as those in the data sections in sections 3, 4, 5, and 6, with possible exceptions as noted in the sections that discuss creating instantiations of the specific messages.

## 8.8 CREATING AN OPM INSTANTIATION

**8.8.1** An OPM instantiation shall be delimited with the `<opm></opm>` root element tags using the standard attributes documented in 8.3.

**8.8.2** The final attributes of the `<opm>` tag shall be `'id'` and `'version'`.

**8.8.3** The `'id'` attribute shall be `'id="CCSDS_OPM_VERS"'`.

**8.8.4** The `'version'` attribute shall be `'version="3.0"'`.

**8.8.5** The standard NDM header shall follow the `<opm>` tag.

**8.8.6** The OPM `<body>` shall consist of a single `<segment>`.

**8.8.7** The `<segment>` shall consist of a `<metadata>` section and a `<data>` section.

**8.8.8** The keywords in the `<metadata>` and `<data>` sections shall be those specified in section 3. The rules for including any of the keyword tags in the OPM/XML are the same as those specified for the OPM/KVN.

**8.8.9** Tags for keywords specified in section 3 shall be all uppercase.

**8.8.10** Several of the OPM/XML keywords may have the units attribute.

**8.8.11** In all cases, the units shall match those defined in the tables in section 3.

**8.8.12** Table 8-2 lists examples of the use of units in the OPM/XML.

**Table 8-2: Examples of Units in OPM/XML**

Keyword	Units	Example
INCLINATION	deg	<code>&lt;INCLINATION units="deg"&gt;numeric-value&lt;/INCLINATION&gt;</code>
MASS	kg	<code>&lt;MASS units="kg"&gt;numeric-value&lt;/MASS&gt;</code>
X	km	<code>&lt;X units="km"&gt;numeric-value&lt;/X&gt;</code>
CX X	km**2	<code>&lt;CX X units="km**2"&gt;numeric-value&lt;/CX X&gt;</code>
CX_DOT_X	km**2/s	<code>&lt;CX_DOT_X units="km**2/s"&gt;numeric-value&lt;/CX_DOT_X&gt;</code>
CX_DOT_X_DOT,	km**2/s**2	<code>&lt;CX_DOT_X_DOT units="km**2/s**2"&gt;numeric-value&lt;/CX_DOT_X_DOT&gt;</code>
GM	km**3/s**2	<code>&lt;GM units="km**3/s**2"&gt;numeric-value&lt;/GM&gt;</code>
X_DOT	km/s	<code>&lt;X_DOT units="km/s"&gt;numeric-value&lt;/X_DOT&gt;</code>
MAN_DV_1	km/s	<code>&lt;MAN_DV_1 units="km/s"&gt;numeric-value&lt;/MAN_DV_1&gt;</code>
SOLAR_RAD_AREA	m**2	<code>&lt;SOLAR_RAD_AREA units="m**2"&gt;numeric-value&lt;/SOLAR_RAD_AREA&gt;</code>
MAN_DURATION	s	<code>&lt;MAN_DURATION units="s"&gt;numeric-value&lt;/MAN_DURATION&gt;</code>

**8.8.13** In addition to the OPM keywords specified in section 3, there are several special tags associated with the OPM body as described in the next few sections. The information

content in the OPM is separated into ‘logical blocks’. Special tags in the OPM are used to encapsulate the information in the logical blocks of the OPM.

**8.8.14** The OPM/XML tags used to delimit the logical blocks of the OPM shall be drawn from table 8-3.

**Table 8-3: OPM/XML Tag Delimiters**

OPM Logical Block	Associated ODM/XML OPM Tag
State Vector	<stateVector>
Keplerian Elements	<keplerianElements>
Spacecraft Parameters	<spacecraftParameters>
Covariance Matrix	<covarianceMatrix>
Maneuver Parameters	<maneuverParameters>
User-Defined Parameters	<userDefinedParameters>

**8.8.15** Between the begin tag and end tag (i.e., between <spacecraftParameters> and </spacecraftParameters>), the user shall place the keywords required by the Spacecraft Parameters logical block as specified in table 3-3.

## 8.9 CREATING AN OMM INSTANTIATION

**8.9.1** An OMM instantiation shall be delimited with the <omm></omm> root element tags using the standard attributes documented in 8.3.

**8.9.2** The final attributes of the <omm> tag shall be ‘id’ and ‘version’.

**8.9.3** The ‘id’ attribute shall be ‘id="CCSDS\_OMM\_VERS"’.

**8.9.4** The ‘version’ attribute for the version of the OMM described in section 4 shall be ‘version="3.0"’.

**8.9.5** The standard NDM header shall follow the <omm> tag.

**8.9.6** The OMM <body> shall consist of a single <segment>.

**8.9.7** The <segment> shall consist of a <metadata> section and a <data> section.

**8.9.8** The keywords in the <metadata> and <data> sections shall be those specified in section 4. The rules for including any of the keyword tags in the OMM/XML are the same as those specified for the OMM/KVN in section 4.

**8.9.9** Tags for keywords specified in section 4 shall be all uppercase.

**8.9.10** Several of the OMM/XML keywords may have the unit attribute.

**8.9.11** In all cases, the units shall match those defined in the tables in section 4.

**8.9.12** Table 8-4 lists examples of the use of units in the OMM/XML.

**Table 8-4: Examples of Units in OMM/XML**

Keyword	Units	Example
BSTAR	1/ER	<BSTAR units="1/ER">numeric-value</BSTAR>
INCLINATION	deg	<INCLINATION units="deg">numeric-value</INCLINATION>
MASS	kg	<MASS units="kg">numeric-value</MASS>
SEMI_MAJOR_AXIS	km	<SEMI_MAJOR_AXIS units="km">numeric-value</SEMI_MAJOR_AXIS>
CX_X	km**2	<CX_X units="km**2">numeric-value</CX_X>
CX_DOT_X	km**2/s	<CX_DOT_X units="km**2/s">numeric-value</CX_DOT_X>
CX_DOT_X_DOT	km**2/s**2	<CX_DOT_X_DOT units="km**2/s**2">numeric-value</CX_DOT_X_DOT>
GM	km**3/s**2	<GM units="km**3/s**2">numeric-value</GM>
SOLAR_RAD_AREA	m**2	<SOLAR_RAD_AREA units="m**2">numeric-value</SOLAR_RAD_AREA>
MEAN_MOTION	rev/day	<MEAN_MOTION units="rev/day">numeric-value</MEAN_MOTION>
MEAN_MOTION_DOT	rev/day**2	<MEAN_MOTION_DOT units="rev/day**2">numeric-value</MEAN_MOTION_DOT>

**8.9.13** In addition to the OMM keywords specified in section 4, there are several special tags associated with the OMM body as described in the next few sections. The information content in the OMM is separated into constructs described in section 4 as ‘logical blocks’. Special tags in the OMM are used to encapsulate the information in the logical blocks of the OMM.

**8.9.14** The OMM/XML tags used to delimit the logical blocks of the OMM shall be drawn from table 8-5:



**Table 8-5: OMM/XML Tag Delimiters**

OMM Logical Block	Associated ODM/XML OMM Tag
Mean Keplerian Elements	<meanElements>
Spacecraft Parameters	<spacecraftParameters>
TLE Parameters	<tleParameters>
Covariance Matrix	<covarianceMatrix>
User-Defined Parameters	<userDefinedParameters>

**8.9.15** Between the begin tag and end tag (i.e., between <spacecraftParameters> and </spacecraftParameters>), the user must place the keywords required by the Spacecraft Parameters logical block as specified in table 4-3.

## 8.10 CREATING AN OEM INSTANTIATION

**8.10.1** An OEM instantiation shall be delimited with the <oem></oem> root element tags using the standard attributes documented in 8.3.

**8.10.2** The final attributes of the <oem> tag shall be 'id' and 'version'.

**8.10.3** The 'id' attribute shall be 'id="CCSDS\_OEM\_VERS"'.

**8.10.4** The 'version' attribute for the version of the OEM described in section 5 shall be 'version="3.0"'.

**8.10.5** The standard NDM header shall follow the <oem> tag.

**8.10.6** The OEM <body> shall consist of one or more <segment> constructs.

**8.10.7** Each <segment> shall consist of a <metadata> section and a <data> section.

**8.10.8** The keywords in the <metadata> and <data> sections shall be those specified in section 5. The rules for including any of the keyword tags in the OEM/XML are the same as those specified for the OEM/KVN in section 5.

**8.10.9** Tags for keywords specified in section 5 shall be all uppercase.

**8.10.10** Several of the OEM/XML keywords may have the unit attribute.

**8.10.11** In all cases, the units shall match those defined in section 5.

**8.10.12** In addition to the OEM keywords specified in section 5, there are some special tags associated with the OEM body as described in the next sections.

**8.10.13**The <stateVector> tag shall encapsulate the keywords associated with one of the ephemeris data lines in the OEM.

**8.10.14**In the XML representation of the OEM, the components of the <stateVector> ephemeris data line must be represented with keywords (i.e., a tag).

**8.10.15**The <stateVector> keywords shall be the same as those defined for the same construct in the OPM.

**8.10.16**The OEM/XML tags used within the <stateVector> structure shall be drawn from table 8-6.

**Table 8-6: Examples of Units in OEM/XML**

OEM Tag	Represents	Example
<EPOCH>	time tag of the state	<EPOCH>2007-09-20T17:41:00</EPOCH>
<X>	x component of position	<X units="km">6678.0</X>
<Y>	y component of position	<Y units="km">0.0</Y>
<Z>	z component of position	<Z units="km">0.0</Z>
<X_DOT>	x component of velocity	<X_DOT units="km/s">0</X_DOT>
<Y_DOT>	y component of velocity	<Y_DOT units="km/s">7.73</Y_DOT>
<Z_DOT>	z component of velocity	<Z_DOT units="km/s">0.0</Z_DOT>
<X_DDOT>	x component of acceleration	<X_DDOT units="km/s**2">0.0</X_DDOT>
<Y_DDOT>	y component of acceleration	<Y_DDOT units="km/s**2">0.50</Y_DDOT>
<Z_DDOT>	z component of acceleration	<Z_DDOT units="km/s**2">0.0</Z_DDOT>

**8.10.17**Between the begin tag and end tag (i.e., between <stateVector> and </stateVector>), the user shall place the values required by the ephemeris data line as specified in section 5.

**8.10.18**Since the state vector structure is shared by the OPM schema and OEM schema, units may optionally appear in the XML version of the OEM ephemeris data line.

**8.10.19**The <covarianceMatrix> tag shall encapsulate the keywords associated with the covariance matrix lines in the OEM.

**8.10.20**In the XML representation of the OEM, the covariance data line must be represented with keywords (i.e., a tag).

**8.10.21**The OEM <covarianceMatrix> keywords shall be the same as those defined for the same construct in the OPM and OMM.

**NOTE** – In the KVN representations of the OEM covariance matrix data lines, keywords are not used. Rather, the components of the covariance matrix data line appear in an order defined in section 5. Similarly, units are not used in the KVN version of the OEM covariance matrix; however, they are optional in the OPM and OMM.

**8.10.22** Since the covariance matrix structure is shared by the OPM, OMM, and OEM, units may optionally appear in the XML version of the OEM covariance matrix line.

**8.10.23** The OEM/XML tags used within the <covarianceMatrix> structure shall be drawn from table 8-7.

**Table 8-7: OEM/XML Tag Delimiters**

<b>Keyword</b>	<b>Units</b>	<b>Example</b>
CX_X, CY_X, CY_Y, CZ_X, CZ_Y, CZ_Z	km**2	<CX_X units="km**2">numeric-value</CX_X>
CX_DOT_X, CX_DOT_Y, CX_DOT_Z, CY_DOT_X, CY_DOT_Y, CY_DOT_Z, CZ_DOT_X, CZ_DOT_Y, CZ_DOT_Z	km**2/s	<CX_DOT_X units="km**2/s">numeric-value</CX_DOT_X>
CX_DOT_X_DOT, CY_DOT_X_DOT, CY_DOT_Y_DOT, CZ_DOT_X_DOT, CZ_DOT_Y_DOT, CZ_DOT_Z_DOT	km**2/s**2	<CX_DOT_X_DOT units="km**2/s**2">numeric-value</CX_DOT_X_DOT>

**8.10.24** Between the begin tag and end tag (i.e., between <covarianceMatrix> and </covarianceMatrix>), the user shall place the values required by the covariance matrix line type as specified in 5.2.5.4 and table 5-4.

**8.11 CREATING AN OCM INSTANTIATION**

**8.11.1** An OCM instantiation shall be delimited with the <ocm></ocm> root element tags using the standard attributes documented in 8.3.

**8.11.2** The final attributes of the <ocm> tag shall be ‘id’ and ‘version’.

**8.11.3** The ‘id’ attribute shall be ‘id="CCSDS\_OCM\_VERS"’.

**8.11.4** The ‘version’ attribute for the version of the OCM described in section 6 shall be ‘version="3.0"’.

**8.11.5** The standard NDM header shall follow the <ocm> tag.

**8.11.6** The OCM <body> shall consist of a single <segment> construct.

**8.11.7** The <segment> shall consist of a <metadata> section and a <data> section.

**8.11.8** The keywords in the <metadata> and <data> sections shall be those specified in section 6. The rules for including any of the keyword tags in the OCM/XML are the same as those specified for the OCM in section 6.

**8.11.9** Tags for keywords specified in section 6 shall be all uppercase.

**8.11.10** Several of the OCM/XML keywords may have the unit attribute.

**8.11.11** In all cases, the units shall match those defined in the SANA Registry (detailed in annex B) and tables in section 6, including orbit, covariance and maneuver data units as specified by TRAJ\_UNITS, COV\_UNITS, and MAN\_UNITS, respectively.

**8.11.12** Table 8-8 lists examples of the use of units in the OCM/XML.

**Table 8-8: Examples of Units in OCM/XML**

Keyword	Units	Example
ACTUAL_OD_SPAN	d	<ACTUAL_OD_SPAN units="d">numeric-value</ACTUAL_OD_SPAN>
DC_PA_START_ANGLE	deg	<DC_PA_START_ANGLE units="deg">numeric-value</DC_PA_START_ANGLE>
CENTRAL_BODY_ROTATION	deg/s	<CENTRAL_BODY_ROTATION units="deg/s">numeric-value</CENTRAL_BODY_ROTATION>
WET_MASS	kg	<MASS units="kg">numeric-value</WET_MASS>
IXX	kg*m**2	<IXX units="kg*m**2">numeric-value</IXX>
EQUATORIAL_RADIUS	km	<EQUATORIAL_RADIUS units="km">numeric-value</EQUATORIAL_RADIUS>
GM	km**3/s**2	<GM units="km**3/s**2">numeric-value</GM>
DV_BOL	km/s	<DV_BOL units="km/s">numeric-value</DV_BOL>
OEB_MAX	m	<OEB_MAX units="m">numeric-value</OEB_MAX>
DRAG_CONST_AREA	m**2	<DRAG_CONST_AREA units="m**2">numeric-value</DRAG_CONST_AREA>
MAX_THRUST	N	<MAX_THRUST units="N">numeric-value</MAX_THRUST>
FIXED_GEOMAG_KP	nT	<FIXED_GEOMAG_KP units="nT">numeric-value</FIXED_GEOMAG_KP>
UT1MUTC_AT_TZERO	s	<UT1MUTC_AT_TZERO units="s">numeric-value</UT1MUTC_AT_TZERO>
FIXED_F10P7	SFU	<FIXED_F10P7 units="SFU">numeric-value</FIXED_F10P7>
DRAG_UNCERTAINTY	%	<DRAG_UNCERTAINTY units="%">numeric-value</DRAG_UNCERTAINTY>

**8.11.13**In addition to the OCM keywords specified in section 6, there are some special tags associated with the OCM body, as listed in table 8-9, described in the next sections.

**Table 8-9: OCM/XML Tag Delimiters**

OCM Logical Block	ODM/XML OCM Section Tags	Data Line Tag
Trajectory Data	<traj>	<trajLine>
Space Object Physical Characteristics	<phys>	N/A
Covariance Data	<covar>	<covLine>
Maneuver Data	<man>	<manLine>
Perturbations Parameters	<pert>	N/A
Orbit Determination Data	<od>	N/A
User-Defined Parameters	<userDef>	N/A

**8.11.14**Between the begin tag and end tag (e.g., between <traj> and </traj>), the user must place the keywords required by the specific OCM section as specified in, for example, table 6-4.

**8.11.15**The data type of the <trajLine>, <covLine>, and <manLine> elements is 'xsd:string', that is, there is no validation of the contents, and the line must be parsed by the OCM recipient to access the individual components of the trajectory, covariance, or maneuver data line.

**8.11.16**The number of individual components in the multipartite <trajLine> shall be determined by the number of components in the value for the TRAJ\_TYPE keyword, plus one for the timetag.

**8.11.17**The number of individual components in the single multipartite <covLine> shall be 13 if COV\_TYPE=SIG3EIGVEC3 is selected, or 14 if COV\_TYPE=TSIG3EIGVEC3 is selected. Otherwise, if 'N' is the dimension of the covariance matrix, then the number of individual components in the single multipartite <covLine> shall either be  $1 + (N**2 + N)/2$  corresponding to the Lower or Upper Triangular Matrix formats plus one for the timetag, or  $1 + N**2$  for the Full Matrix format plus one for the timetag.

**8.11.18**The number of individual components in the multipartite <manLine> shall be determined by the number of comma-separated values in the MAN\_COMPOSITION keyword, plus one for the timetag.

## 8.12 CREATING A COMBINED INSTANTIATION

**8.12.1** An ODM user may create an XML instance that incorporates any number of messages from sections 3 through 6 of this document in a logical suite called an 'NDM

(Navigation Data Message) Combined Instantiation'. Such combined instantiations may be useful for some situations, for example:

**8.12.2** A maneuver scenario in which both 'no burn' and 'with burn' ephemerides are combined in a single message.

**8.12.3** A constellation scenario in which states (OPM, OMM) and/or ephemeris data (OEM, OCM) for all the spacecraft in the constellation are combined in a single XML message.

**8.12.4** A full OEM or OCM ephemeris with detail on important states reflected in some number of OPMs. The OEM/OCM and the multiple OPMs can be conveniently conveyed in a single NDM.

**8.12.5** An NDM combined instantiation shall be delimited with the `<ndm></ndm>` root element tags instead of one of the individual message tags described in 8.3.2.

**8.12.6** The standard attributes documented in 8.3 shall be used with the `<ndm>` tag, with the exception that neither 'id' nor 'version' attributes are associated with the `<ndm>` tag.

**8.12.7** In the NDM combined instantiation, the only attributes that shall appear on the constituent message tags (i.e., the tags listed in 8.3.2) are the 'id' and 'version' attributes.

**8.12.8** Between the `<ndm></ndm>` tags, the desired messages described in 8.8 through 8.11 may be combined as needed to meet user requirements.

**8.12.9** Any combination of constituent ODM types may be used in an NDM combined instantiation.

**8.12.10** Figures 8-2 and 8-3 illustrate the basic structure of an NDM combined instantiation. All detail has been removed from figure 8-1 to contrast the single message ODM with an NDM combined instantiation. As shown in figure 8-2, in an NDM combined instantiation the individual message tags still have the 'id' and 'version' attributes, but the namespace attributes and schema location attributes are associated with the `<ndm>` root element.

Single Message OPM	NDM Combined Instantiation
<pre> &lt;opm&gt;   &lt;header&gt; &lt;/header&gt;   &lt;body&gt; &lt;/body&gt; &lt;/opm&gt; </pre>	<pre> &lt;ndm&gt;   &lt;opm&gt;     &lt;header&gt; &lt;/header&gt;     &lt;body&gt; &lt;/body&gt;   &lt;/opm&gt;   .   .   .   &lt;opm&gt;     &lt;header&gt; &lt;/header&gt;     &lt;body&gt; &lt;/body&gt;   &lt;/opm&gt; &lt;/ndm&gt; </pre>

**Figure 8-2: Comparison of Single Message OPM with NDM Combined Instantiation**

**8.12.11** The OPMs shown in figure 8-2 may be replaced with any number of OMM, OEM, or OCM messages in any combination as needed to meet user requirements, as shown in figure 8-3, below.

```

<?xml version="1.0" encoding="UTF-8"?>
<ndm xmlns:
xsi:noNamespaceSchemaLocation=https://sanaregistry.org/r/ndmxml\_unqualified/ndmxml-3.0.0-
master-3.0.xsd>

<COMMENT>This figure combines multiple ODM/XML messages into a single message</COMMENT>
<COMMENT>Message detail is deleted in order to focus on the message structure</COMMENT>
<COMMENT>Note use of "<ndm>" root element, and ODM/version attributes</COMMENT>

  <opm id="CCSDS_OPM_VERS" version="3.0">
    <header>
    </header>
    <body>
    </body>
  </opm>

  <oem id="CCSDS_OEM_VERS" version="3.0">
    <header>
    </header>
    <body>
    </body>
  </oem>

  <opm id="CCSDS_OPM_VERS" version="3.0">
    <header>
    </header>
    <body>
    </body>
  </opm>

  <omm id="CCSDS_OMM_VERS" version="3.0">
    <header>
    </header>
    <body>
    </body>
  </omm>

  <ocm id="CCSDS_OCM_VERS" version="3.0">
    <header>
    </header>
    <body>
    </body>
  </ocm>

</ndm>

```

**Figure 8-3: NDM Combined Instantiation Showing Mix of ODMs and Use of Attributes**

NOTE – Figure G-21 shows a full example of a use case combining multiple ODMs in a single XML message. (For instructions on creating a combined instantiation that incorporates ODM/XML messages combined with other navigation related messages, see reference [5].)

## 8.13 SPECIAL SYNTAX RULES FOR ODM/XML

**8.13.1** Most of the KVN syntax rules apply for ODM/XML instantiations of an ODM; however, there are a few variations described in this section that shall be observed.

**8.13.2** Each mandatory XML tag must be present and contain a valid value.



**8.13.3** Integer values shall follow the conventions of the integer data type per reference [6]. Additional restrictions on the allowable range of values permitted for any integer data element may also be defined in the ODM/XML Schema.

NOTE – Examples of such restrictions may include a defined range (e.g., 0 - 100, 1 - 10, etc.), a set of enumerated values (e.g., 0,1,2,4,8), a pre-defined specific variation such as positiveInteger, or a user-defined data type variation.

**8.13.4** Non-integer numeric values may be expressed in either fixed-point or floating-point notation. Numeric values shall follow the conventions of the double data type per reference [6]. Additional restrictions on the allowable range of values permitted for any numeric data element may also be defined in the ODM/XML Schema.

NOTE – Examples of such restrictions may include a defined range (e.g., 0.0-100.0, etc.), or a user-defined data type variation.

**8.13.5** Text values shall follow the conventions of the string data type per reference [6]. Additional restrictions on the allowable range or values permitted for any data element may also be defined in the ODM/XML Schema.

NOTE – Examples of such restrictions may include a set of enumerated values (e.g., ‘YES’/‘NO’) or other user-defined data type variation.

**8.13.6** The units in the ODM/XML shall be the same units used in the KVN-formatted ODM described in 7.7, or as mandated in the SANA registry per annex B. XML attributes shall be used to explicitly define the units or other important information associated with the given data element. (See the tables in this section for the OPM, OMM, OEM, and OCM for examples of coding units in ODM/XML instantiations.)

**8.13.7** Comments must be displayed as values between the <COMMENT> and </COMMENT> tags.

**ANNEX A**

**IMPLEMENTATION CONFORMANCE**

**STATEMENT PROFORMA**

**(NORMATIVE)**

**A1 INTRODUCTION**

**A1.1 OVERVIEW**

This annex provides the Implementation Conformance Statement (ICS) Requirements List (RL) for an implementation of the Orbit Data Messages (CCSDS 502.0). The ICS for an implementation is generated by completing the RL in accordance with the instructions below. An implementation shall satisfy the mandatory conformance requirements referenced in the RL.

The RL in this annex is blank. An implementation's completed RL is called the ICS. The ICS states which capabilities and options have been implemented. The following can use the ICS:

- the implementer, as a checklist to reduce the risk of failure to conform to the standard through oversight;
- a supplier or potential acquirer of the implementation, as a detailed indication of the capabilities of the implementation, stated relative to the common basis for understanding provided by the standard ICS proforma;
- a user or potential user of the implementation, as a basis for initially checking the possibility of interworking with another implementation (it should be noted that, while interworking can never be guaranteed, failure to interwork can often be predicted from incompatible ICS lists);
- a tester, as the basis for selecting appropriate tests against which to assess the claim for conformance of the implementation.

**A1.2 ABBREVIATIONS AND CONVENTIONS**

The RL consists of information in tabular form. The status of features is indicated using the abbreviations and conventions described below.

Item Column

The item column contains sequential numbers for items in the table.

### Feature Column

The feature column contains a brief descriptive name for a feature. It implicitly means “Is this feature supported by the implementation?”

### Status Column

The status column uses the following notations:

- M            mandatory;
- O            optional;
- C            conditional;
- X            prohibited;
- I            out of scope;
- N/A         not applicable.

### Support Column Symbols

The support column is to be used by the implementer to state whether a feature is supported by entering Y, N, or N/A, indicating:

- Y        Yes, supported by the implementation.
- N        No, not supported by the implementation.
- N/A     Not applicable.

The support column should also be used, when appropriate, to enter values supported for a given capability.

## **A1.3 INSTRUCTIONS FOR COMPLETING THE RL**

An implementer shows the extent of compliance to the Recommended Standard by completing the RL; that is, the state of compliance with all mandatory requirements and the options supported are shown. The resulting completed RL is called an ICS. The implementer shall complete the RL by entering appropriate responses in the support or values supported column, using the notation described in A1.2. If a conditional requirement is inapplicable, N/A should be used. If a mandatory requirement is not satisfied, exception information must be supplied by entering a reference  $X_i$ , where  $i$  is a unique identifier, to an accompanying rationale for the noncompliance.

## A2 ICS PROFORMA FOR ORBIT DATA MESSAGES

### A2.1 IDENTIFICATION OF ICS

Date of Statement (DD/MM/YYYY)	
ICS serial number	
System Conformance statement cross-reference	

### A2.2 IDENTIFICATION OF IMPLEMENTATION UNDER TEST

Implementation name	
Implementation version	
Special Configuration	
Other Information	

### A2.3 IDENTIFICATION OF SUPPLIES

Supplier	
Contact Point for Queries	
Implementation Name(s) and Versions	
Other information necessary for full identification, for example, name(s) and version(s) for machines and/or operating systems; System Name(s)	

### A2.4 DOCUMENT VERSIONS

CCSDS 502.0 Document Version	3
Have any exceptions been required? NOTE – A YES answer means that the implementation does not conform to the Recommended Standard. Non-supported mandatory capabilities are to be identified in the ICS, with an explanation of why the implementation is non-conforming.	Yes ____ No ____

## A2.5 REQUIREMENTS LISTS

### A2.5.1 Orbit Parameter Message Requirements List

Item	Feature	Keyword	Reference	Status M/O/C	Support
1	OPM Header	N/A	Table 3-1	M	
2	OPM Version	CCSDS_OPM_VERS	Table 3-1	M	
3	Comment	COMMENT	Table 3-1	O	
4	Message creation date and time	CREATION_DATE	Table 3-1	M	
5	Message originator	ORIGINATOR	Table 3-1	M	
6	Unique message identifier	MESSAGE_ID	Table 3-1	O	
7	OPM Metadata	N/A	Table 3-2	M	
8	Comment	COMMENT	Table 3-2	O	
9	Name of space object	OBJECT_NAME	Table 3-2	M	
10	Identifier of space object	OBJECT_ID	Table 3-2	M	
11	Orbit center	CENTER_NAME	Table 3-2	M	
12	Reference frame	REF_FRAME	Table 3-2	M	
13	Epoch of reference frame	REF_FRAME_EPOCH	Table 3-2	C	
14	Time system applicable to data	TIME_SYSTEM	Table 3-2	M	
15	OPM Data	N/A	Table 3-3	M	
16	State Vector logical block	N/A	Table 3-3	M	
17	Comment	COMMENT	Table 3-3	O	
18	Epoch of the state vector	EPOCH	Table 3-3	M	
19	X component of position	X	Table 3-3	M	
20	Y component of position	Y	Table 3-3	M	
21	Z component of position	Z	Table 3-3	M	
22	X component of velocity	X_DOT	Table 3-3	M	
23	Y component of velocity	Y_DOT	Table 3-3	M	
24	Z component of velocity	Z_DOT	Table 3-3	M	
25	Keplerian Elements logical block	N/A	Table 3-3	O	

DRAFT CCSDS RECOMMENDED STANDARD FOR ORBIT DATA MESSAGES

Item	Feature	Keyword	Reference	Status M/O/C	Support
26	Comment	COMMENT	Table 3-3	O	
27	Semi-major axis of orbit	SEMI_MAJOR_AXIS	Table 3-3	C	
28	Eccentricity of orbit	ECCENTRICITY	Table 3-3	C	
29	Inclination of orbit	INCLINATION	Table 3-3	C	
30	Right ascension of ascending node of orbit	RA_OF_ASC_NODE	Table 3-3	C	
31	Argument of pericenter of orbit	ARG_OF_PERICENTER	Table 3-3	C	
32	True or mean anomaly of orbit	TRUE_ANOMALY or MEAN_ANOMALY	Table 3-3	C	
33	Gravitational coefficient of the central body	GM	Table 3-3	C	
34	Spacecraft Parameters logical block	N/A	Table 3-3	O	
35	Comment	COMMENT	Table 3-3	O	
36	Mass of the spacecraft	MASS	Table 3-3	C	
37	Solar radiation area of the spacecraft	SOLAR_RAD_AREA	Table 3-3	C	
38	Solar radiation coefficient of the spacecraft	SOLAR_RAD_COEFF	Table 3-3	C	
39	Drag area of the spacecraft	DRAG_AREA	Table 3-3	C	
40	Drag coefficient of the spacecraft	DRAG_COEFF	Table 3-3	C	
41	Position/velocity Covariance Matrix logical block	N/A	Table 3-3	O	
42	Comment	COMMENT	Table 3-3	O	
43	Cov reference frame	COV_REF_FRAME	Table 3-3	C	
44	Covariance matrix [1,1]	CX_X	Table 3-3	C	
45	Covariance matrix [2,1]	CY_X	Table 3-3	C	
46	Covariance matrix [2,2]	CY_Y	Table 3-3	C	
47	Covariance matrix [3,1]	CZ_X	Table 3-3	C	
48	Covariance matrix [3,2]	CZ_Y	Table 3-3	C	
49	Covariance matrix [3,3]	CZ_Z	Table 3-3	C	
50	Covariance matrix [4,1]	CX_DOT_X	Table 3-3	C	
51	Covariance matrix [4,2]	CX_DOT_Y	Table 3-3	C	
52	Covariance matrix [4,3]	CX_DOT_Z	Table 3-3	C	

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Item	Feature	Keyword	Reference	Status M/O/C	Support
53	Covariance matrix [4,4]	CX_DOT_X_DOT	Table 3-3	C	
54	Covariance matrix [5,1]	CY_DOT_X	Table 3-3	C	
55	Covariance matrix [5,2]	CY_DOT_Y	Table 3-3	C	
56	Covariance matrix [5,3]	CY_DOT_Z	Table 3-3	C	
57	Covariance matrix [5,4]	CY_DOT_X_DOT	Table 3-3	C	
58	Covariance matrix [5,5]	CY_DOT_Y_DOT	Table 3-3	C	
59	Covariance matrix [6,1]	CZ_DOT_X	Table 3-3	C	
60	Covariance matrix [6,2]	CZ_DOT_Y	Table 3-3	C	
61	Covariance matrix [6,3]	CZ_DOT_Z	Table 3-3	C	
62	Covariance matrix [6,4]	CZ_DOT_X_DOT	Table 3-3	C	
63	Covariance matrix [6,5]	CZ_DOT_Y_DOT	Table 3-3	C	
64	Covariance matrix [6,6]	CZ_DOT_Z_DOT	Table 3-3	C	
65	Maneuver Parameters logical block	N/A	Table 3-3	O	
66	Comment	COMMENT	Table 3-3	O	
67	Time of maneuver start	MAN_EPOCH_IGNITION	Table 3-3	O	
68	Duration of maneuver	MAN_DURATION	Table 3-3	O	
69	Change of mass due to maneuver	MAN_DELTA_MASS	Table 3-3	O	
70	Relevant reference frame for maneuver	MAN_REF_FRAME	Table 3-3	O	
71	1st component of velocity change	MAN_DV_1	Table 3-3	O	
72	2nd component of velocity change	MAN_DV_2	Table 3-3	O	
73	3rd component of velocity change	MAN_DV_3	Table 3-3	O	
74	User-Defined Parameters logical block	N/A	Table 3-3	O	
75	As defined by user, 'essential information that cannot be conveyed in COMMENT statements'	USER_DEFINED_x	Table 3-3	O	

**A2.5.2 Orbit Mean Elements Message Requirements List**

Item	Feature	Keyword	Reference	Status M/O/C	Support
1	OMM Header	N/A	Table 4-1	M	
2	OMM Version	CCSDS_OMM_VERS	Table 4-1	M	
3	Comment	COMMENT	Table 4-1	O	
4	Message creation date and time	CREATION_DATE	Table 4-1	M	
5	Message originator	ORIGINATOR	Table 4-1	M	
6	Unique message identifier	MESSAGE_ID	Table 4-1	O	
7	OMM Metadata	N/A	Table 4-2	M	
8	Comment	COMMENT	Table 4-2	O	
9	Name of space object	OBJECT_NAME	Table 4-2	M	
10	Identifier of space object	OBJECT_ID	Table 4-2	M	
11	Orbit center	CENTER_NAME	Table 4-2	M	
12	Reference frame	REF_FRAME	Table 4-2	M	
13	Epoch of reference frame	REF_FRAME_EPOCH	Table 4-2	C	
14	Time system applicable to data	TIME_SYSTEM	Table 4-2	M	
15	Mean element set theory of data	MEAN_ELEMENT_THEORY	Table 4-2	M	
16	OMM Data	N/A	Table 4-3	M	
17	Mean Keplerian elements logical block	N/A	Table 4-3	M	
18	Comment	COMMENT	Table 4-3	O	
19	Epoch of the orbital elements	EPOCH	Table 4-3	M	
20	Semi-major axis or mean motion	SEMI_MAJOR_AXIS or MEAN_MOTION	Table 4-3	M	
21	Eccentricity	ECCENTRICITY	Table 4-3	M	
22	Inclination	INCLINATION	Table 4-3	M	
23	Right ascension of ascending node	RA_OF_ASC_NODE	Table 4-3	M	
24	Argument of pericenter	ARG_OF_PERICENTER	Table 4-3	M	
25	Mean anomaly	MEAN_ANOMALY	Table 4-3	M	
26	Gravitational Coefficient	GM	Table 4-3	O	
27	Spacecraft Parameters logical block	N/A	Table 4-3	O	



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Item	Feature	Keyword	Reference	Status M/O/C	Support
28	Comment	COMMENT	Table 4-3	O	
29	Spacecraft Mass	MASS	Table 4-3	O	
30	Solar Radiation Pressure Area	SOLAR_RAD_AREA	Table 4-3	O	
31	Solar Radiation Pressure Coefficient	SOLAR_RAD_COEFF	Table 4-3	O	
32	Drag Area	DRAG_AREA	Table 4-3	O	
33	Drag Coefficient	DRAG_COEFF	Table 4-3	O	
34	TLE logical block	N/A	Table 4-3	O	
35	Comment	COMMENT	Table 4-3	O	
36	Ephemeris Type	EPHEMERIS_TYPE	Table 4-3	O	
37	Classification Type	CLASSIFICATION_TYPE	Table 4-3	O	
38	NORAD Catalog Number	NORAD_CAT_ID	Table 4-3	O	
39	Element set number	ELEMENT_SET_NO	Table 4-3	O	
40	Revolution Number	REV_AT_EPOCH	Table 4-3	O	
41	SGP/SGP4 drag-like coefficient	BSTAR	Table 4-3	O	
42	First Time Derivative of the Mean Motion	MEAN_MOTION_DOT	Table 4-3	O	
43	Second Time Derivative of Mean Motion	MEAN_MOTION_DDOT	Table 4-3	O	
44	Pos/Vel/Cov logical block		Table 4-3	O	
45	Comment	COMMENT	Table 4-3	O	
46	Cov reference frame	COV_REF_FRAME	Table 4-3	C	
47	Covariance [1,1]	CX_X	Table 4-3	C	
48	Covariance [2,1]	CY_X	Table 4-3	C	
49	Covariance [2,2]	CY_Y	Table 4-3	C	
50	Covariance [3,1]	CZ_X	Table 4-3	C	
51	Covariance [3,2]	CZ_Y	Table 4-3	C	
52	Covariance [3,3]	CZ_Z	Table 4-3	C	
53	Covariance [4,1]	CX_DOT_X	Table 4-3	C	
54	Covariance [4,2]	CX_DOT_Y	Table 4-3	C	
55	Covariance [4,3]	CX_DOT_Z	Table 4-3	C	
56	Covariance [4,4]	CX_DOT_X_DOT	Table 4-3	C	
57	Covariance [5,1]	CY_DOT_X	Table 4-3	C	

Item	Feature	Keyword	Reference	Status M/O/C	Support
58	Covariance [5,2]	CY_DOT_Y	Table 4-3	C	
59	Covariance [5,3]	CY_DOT_Z	Table 4-3	C	
60	Covariance [5,4]	CY_DOT_X_DOT	Table 4-3	C	
61	Covariance [5,5]	CY_DOT_Y_DOT	Table 4-3	C	
62	Covariance [6,1]	CZ_DOT_X	Table 4-3	C	
63	Covariance [6,2]	CZ_DOT_Y	Table 4-3	C	
64	Covariance [6,3]	CZ_DOT_Z	Table 4-3	C	
65	Covariance [6,4]	CZ_DOT_X_DOT	Table 4-3	C	
66	Covariance [6,5]	CZ_DOT_Y_DOT	Table 4-3	C	
67	Covariance [6,6]	CZ_DOT_Z_DOT	Table 4-3	C	
68	User-Defined Parameters logical block	N/A	Table 4-3	O	
69	As defined by user, 'essential information that cannot be conveyed in COMMENT statements'	USER_DEFINED_x	Table 4-3	O	

### A2.5.3 Orbit Ephemeris Message Requirements List

Item	Feature	Keyword	Reference	Status M/O/C	Support
1	OEM Header	N/A	Table 5-2	M	
2	OEM Version	CCSDS_OEM_VERS	Table 5-2	M	
3	Comment	COMMENT	Table 5-2	O	
4	Message creation date and time	CREATION_DATE	Table 5-2	M	
5	Message originator	ORIGINATOR	Table 5-2	M	
6	Unique message identifier	MESSAGE_ID	Table 5-2	O	
7	Metadata logical block	N/A	Table 5-3	M	
8	Start of OEM Metadata	META_START	Table 5-3	M	
9	Comment	COMMENT	Table 5-3	O	
10	Name of space object	OBJECT_NAME	Table 5-3	M	
11	Identifier of space object	OBJECT_ID	Table 5-3	M	
12	Orbit center	CENTER_NAME	Table 5-3	M	
13	Reference frame	REF_FRAME	Table 5-3	M	

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Item	Feature	Keyword	Reference	Status M/O/C	Support
14	Epoch of reference frame	REF_FRAME_EPOCH	Table 5-3	C	
15	Time system applicable to data	TIME_SYSTEM	Table 5-3	M	
16	Start of TOTAL time span covered by data	START_TIME	Table 5-3	M	
17	Start of useable orbit data	USEABLE_START_TIME	Table 5-3	O	
18	End of useable orbit data	USEABLE_STOP_TIME	Table 5-3	O	
19	End of TOTAL time span covered by data	STOP_TIME	Table 5-3	M	
20	Recommended interpolation method	INTERPOLATION	Table 5-3	O	
21	Recommended interpolation degree	INTERPOLATION_DEGREE	Table 5-3	C	
22	End of OEM Metadata	META_STOP	Table 5-3	M	
24	OEM Data logical block	N/A	Table 5-3	M	
23	Ephemeris lines	... <insert ephemeris data lines here.>	Table 5-3	M	
24	OEM Covariance logical block	N/A	Table 5-3	O	
25	Start of OEM Covariance logical block	COVARIANCE_START	Table 5-3	M	
26	Epoch of the navigation solution related to the covariance matrix	EPOCH	Table 5-3	C	
27	Reference frame of the covariance matrix, if different from that of the states in the ephemeris	COV_REF_FRAME	Table 5-3	C	
28	Covariance lines	... <insert covariance matrices here>	Table 5-3	O	
29	End of OEM Covariance logical block	COVARIANCE_STOP	Table 5-3	M	

## A2.5.4 Orbit Comprehensive Message Requirements List

### A2.5.4.1 OCM Header

Item	Feature	Keyword	Reference	Status M/O/C	Support
1	OCM Header	N/A	Table 6-2	M	
2	OCM Version	CCSDS_OCM_VERS	Table 6-2	M	
3	Comment	COMMENT	Table 6-2	O	
4	Message creation date and time	CREATION_DATE	Table 6-2	M	
5	Message originator	ORIGINATOR	Table 6-2	M	
6	Unique message identifier	MESSAGE_ID	Table 6-2	O	

### A2.5.4.2 OCM Metadata

Item	Feature	Keyword	Reference	Status M/O/C	Support
1	Metadata logical block	N/A	Table 6-3	M	
2	OCM Metadata Start	META_START	Table 6-3	M	
3	Comment	COMMENT	Table 6-3	O	
4	Message classification/caveats	CLASSIFICATION	Table 6-2	O	
5	Spacecraft name for the object	OBJECT_NAME	Table 6-3	O	
6	International designator for the object	INTERNATIONAL_DESIGNATOR	Table 6-3	O	
7	Satellite catalog source	CATALOG_NAME	Table 6-3	O	
8	Unique satellite identification designator	OBJECT_DESIGNATOR	Table 6-3	O	
9	Alternate name(s) of space object used by spacecraft operator, State Actors, commercial SSA providers and/or media.	ALTERNATE_NAMES	Table 6-3	O	
10	Message originator or programmatic Point-of-Contact	ORIGINATOR_POC	Table 6-3	O	
11	Contact position of the originator PoC	ORIGINATOR_POSITION	Table 6-3	O	
12	Originator PoC phone number	ORIGINATOR_PHONE	Table 6-3	O	

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Item	Feature	Keyword	Reference	Status M/O/C	Support
13	Originator PoC email address	ORIGINATOR_EMAIL	Table 6-3	O	
14	Originator PoC physical address	ORIGINATOR_ADDRESS	Table 6-3	O	
15	Creating agency or operator	TECH_ORG	Table 6-3	O	
16	Technical Point-of-Contact	TECH_POC	Table 6-3	O	
17	Contact position of the technical PoC	TECH_POSITION	Table 6-3	O	
18	Technical PoC phone number	TECH_PHONE	Table 6-3	O	
19	Technical PoC email address	TECH_EMAIL	Table 6-3	O	
20	Technical PoC physical address	TECH_ADDRESS	Table 6-3	O	
21	ID that uniquely identifies the previous message from this message originator for this particular space object	PREVIOUS_MESSAGE_ID	Table 6-3	O	
22	ID that uniquely identifies the next message from this message originator for this particular space object	NEXT_MESSAGE_ID	Table 6-3	O	
23	Unique identifier of linked Attitude Data Message(s)	ADM_MSG_LINK	Table 6-3	O	
24	Unique identifier of linked Conjunction Data Message(s)	CDM_MSG_LINK	Table 6-3	O	
25	Unique identifier of linked Pointing Request Message(s)	PRM_MSG_LINK	Table 6-3	O	
26	Unique identifier of linked Reentry Data Message(s)	RDM_MSG_LINK	Table 6-3	O	
27	Unique identifier of linked Tracking Data Message(s)	TDM_MSG_LINK	Table 6-3	O	
28	Operator of the space object	OPERATOR	Table 6-3	O	
29	Owner of the space object	OWNER	Table 6-3	O	

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Item	Feature	Keyword	Reference	Status M/O/C	Support
30	Country where the owner or responsible party is based	COUNTRY	Table 6-3	O	
31	Name of the constellation	CONSTELLATION	Table 6-3	O	
32	Type of object	OBJECT_TYPE	Table 6-3	O	
33	Time system for all absolute time stamps in this OCM including EPOCH_TZERO	TIME_SYSTEM	Table 6-3	M	
34	Default epoch to which all relative times are referenced	EPOCH_TZERO	Table 6-3	M	
35	Operational status of the space object	OPS_STATUS	Table 6-3	O	
36	Type of orbit	ORBIT_CATEGORY	Table 6-3	O	
37	Elements of information included in this message	OCM_DATA_ELEMENTS	Table 6-3	O	
38	Spacecraft clock epoch	SCLK_OFFSET_AT_EPOCH	Table 6-3	C	
39	Spacecraft clock rate	SCLK_SEC_PER_SI_SEC	Table 6-3	C	
40	Creation epoch of the previous message from this originator for this particular space object	PREVIOUS_MESSAGE_EPOCH	Table 6-3	O	
41	Anticipated (or actual) epoch of the next message from this originator for this particular space object	NEXT_MESSAGE_EPOCH	Table 6-3	O	
42	Time of the earliest data	START_TIME	Table 6-3	O	
43	Time of the latest data	STOP_TIME	Table 6-3	O	
44	Span of time that the OCM covers, measured in days	TIME_SPAN	Table 6-3	O	
45	Difference (TAI – UTC) in seconds	TAIMUTC_AT_TZERO	Table 6-3	O	
46	Epoch of next leap second(s)	NEXT_LEAP_EPOCH	Table 6-3	O	
47	Difference (TAI – UTC) after next leap second(s) are introduced	NEXT_LEAP_TAIMUTC	Table 6-3	O	
48	Difference (UT1 – UTC) in seconds	UT1MUTC_AT_TZERO	Table 6-3	O	

Item	Feature	Keyword	Reference	Status M/O/C	Support
49	Source and version of the message originator's EOP	EOP_SOURCE	Table 6-3	O	
50	Method used to select or interpolate sequential EOP data	INTERP_METHOD_EOP	Table 6-3	O	
51	Source and version of the message originator's celestial body (e.g., Sun/Earth/Planetary) ephemeris data	CELESTIAL_SOURCE	Table 6-3	O	
52	Metadata Stop	META_STOP	Table 6-3	M	

#### A2.5.4.3 OCM Data: Trajectory State Time History

Item	Feature	Keyword	Reference	Status M/O/C	Support
1	Trajectory state time history logical block	N/A	Table 6-4	O	
2	Trajectory state time history start	TRAJ_START	Table 6-4	M	
3	Comment	COMMENT	Table 6-4	O	
4	Identification number for this trajectory state time history block	TRAJ_ID	Table 6-4	O	
5	Identification number for the previous trajectory state time history	TRAJ_PREV_ID	Table 6-4	O	
6	Identification number for the next trajectory state time history	TRAJ_NEXT_ID	Table 6-4	O	
7	Basis of this Trajectory state time history data	TRAJ_BASIS	Table 6-4	O	
8	Identification number for the orbit determination, navigation solution, or simulation	TRAJ_BASIS_ID	Table 6-4	O	
9	Recommended interpolation method	INTERPOLATION	Table 6-4	O	
10	Recommended interpolation degree	INTERPOLATION_DEGREE	Table 6-4	C	
11	Orbit averaging technique used	ORB_AVERAGING	Table 6-4	M	

12	Orbit propagator used	PROPAGATOR	Table 6-4	O	
13	Origin of the orbit reference frame	CENTER_NAME	Table 6-4	M	
14	Reference frame of the trajectory state time history	TRAJ_REF_FRAME	Table 6-4	M	
15	Epoch of the orbit data reference frame	TRAJ_FRAME_EPOCH	Table 6-4	C	
16	Start of useable orbit data	USEABLE_START_TIME	Table 6-4	O	
17	End of useable orbit data	USEABLE_STOP_TIME	Table 6-4	O	
18	Orbit revolution number (an integer)	ORB_REVNUM	Table 6-4	O	
19	Orbit revolution basis number	ORB_REVNUM_BASIS	Table 6-4	C	
20	Orbit element set type	TRAJ_TYPE	Table 6-4	M	
21	Orbital element units for data elements that follow the time tag	TRAJ_UNITS	Table 6-4	O	
22	OCM trajectory state time history	... <insert trajectory state time history here>	Table 6-4	M	
23	Trajectory state time history end	TRAJ_STOP	Table 6-4	M	

#### A2.5.4.4 OCM Data: Space Object Physical Characteristics

Item	Feature	Keyword	Reference	Status M/O/C	Support
1	Space Object Physical Characteristics logical block	N/A	Table 6-5	O	
2	Start of a Space Object Physical Characteristics specification	PHYS_START	Table 6-5	M	
3	Comment	COMMENT	Table 6-5	O	
4	Satellite manufacturer name	MANUFACTURER	Table 6-5	O	
5	Satellite manufacturer's spacecraft bus model name	BUS_MODEL	Table 6-5	O	
6	Space objects that this object is docked to	DOCKED_WITH	Table 6-5	O	
7	Additional drag Area facing the relative wind vector beyond that represented by AREA_ALONG_OEB_<MAX, MED, MIN>	DRAG_CONST_AREA	Table 6-5	O	
8	Drag coefficient	DRAG_COEFF_NOM	Table 6-5	O	



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Item	Feature	Keyword	Reference	Status M/O/C	Support
9	Drag coeff. $1\sigma$ uncertainty	DRAG_UNCERTAINTY	Table 6-5	O	
10	Space object total mass at beginning of life	INITIAL_WET_MASS	Table 6-5	O	
11	Space object wet mass	WET_MASS	Table 6-5	O	
12	Space object dry mass	DRY_MASS	Table 6-5	O	
13	Parent reference frame which maps to the OEB frame	OEB_PARENT_FRAME	Table 6-5	C	
14	Epoch of the OEB parent reference frame	OEB_PARENT_FRAME_EPOCH	Table 6-5	C	
15	Quaternion $q_1$	OEB_Q1	Table 6-5	O	
16	Quaternion $q_2$	OEB_Q2	Table 6-5	O	
17	Quaternion $q_3$	OEB_Q3	Table 6-5	O	
18	Quaternion $q_c$	OEB_QC	Table 6-5	O	
19	Maximum physical OEB dimension	OEB_MAX	Table 6-5	O	
20	Medium physical OEB dimension	OEB_INT	Table 6-5	O	
21	Minimum physical dimension	OEB_MIN	Table 6-5	O	
22	Cross-sectional area viewed along max OEB dimension	AREA_ALONG_OEB_MAX	Table 6-5	O	
23	Cross-sectional area viewed along intermediate OEB dimension	AREA_ALONG_OEB_INT	Table 6-5	O	
24	Cross-sectional area viewed along min OEB dimension	AREA_ALONG_OEB_MIN	Table 6-5	O	
25	Minimum cross-sectional area for collision probability	AREA_MIN_FOR_PC	Table 6-5	O	
26	Maximum cross-sectional area for collision probability	AREA_MAX_FOR_PC	Table 6-5	O	
27	Typical cross-sectional area for collision probability	AREA_TYP_FOR_PC	Table 6-5	O	
28	Typical (50th percentile) effective Radar Cross Section	RCS	Table 6-5	O	
29	Minimum Radar Cross Section	RCS_MIN	Table 6-5	O	
30	Maximum Radar Cross Section	RCS_MAX	Table 6-5	O	
31	SRP additional area	SRP_CONST_AREA	Table 6-5	O	
32	Solar Radiation Pressure Coefficient	SOLAR_RAD_COEFF	Table 6-5	O	

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Item	Feature	Keyword	Reference	Status M/O/C	Support
33	Solar Radiation Pressure $1\sigma$ percent uncertainty	SOLAR_RAD_UNCERTAINTY	Table 6-5	O	
34	Typical (50th percentile) absolute Visual Magnitude	VM_ABSOLUTE	Table 6-5	O	
35	Minimum apparent Visual Magnitude	VM_APPARENT_MIN	Table 6-5	O	
36	Typical (50th percentile) apparent Visual Magnitude	VM_APPARENT	Table 6-5	O	
37	Maximum apparent Visual Magnitude	VM_APPARENT_MAX	Table 6-5	O	
38	Typical (50th percentile) object surface reflectance	REFLECTANCE	Table 6-5	O	
39	Primary mode of attitude control	ATT_CONTROL_MODE	Table 6-5	O	
40	Type of actuator for attitude control	ATT_ACTUATOR_TYPE	Table 6-5	O	
41	Accuracy of attitude knowledge	ATT_KNOWLEDGE	Table 6-5	O	
42	Ability (e.g., deadband) to control attitude	ATT_CONTROL	Table 6-5	O	
43	Combined ability to both knowledge and control attitude	ATT_POINTING	Table 6-5	O	
44	Average maneuver frequency	AVG_MANEUVER_FREQ	Table 6-5	O	
45	Maximum composite thrust	MAX_THRUST	Table 6-5	O	
46	Total $\Delta V$ capability of the spacecraft at beginning of life	DV_BOL	Table 6-5	O	
47	Total $\Delta V$ remaining	DV_REMAINING	Table 6-5	O	
48	Moment of Inertia about the X-axis	IXX	Table 6-5	O	
49	Moment of Inertia about the Y-axis	IYY	Table 6-5	O	
50	Moment of Inertia about the Z-axis	IZZ	Table 6-5	O	
51	Inertia Cross Product of the X & Y axes	IXY	Table 6-5	O	
52	Inertia Cross Product of the X & Z axes	IXZ	Table 6-5	O	
53	Inertia Cross Product of the Y & Z axes	IYZ	Table 6-5	O	
54	End of the Space Object Physical Characteristics specification	PHYS_STOP	Table 6-5	M	

**A2.5.4.5 OCM Data: Covariance Time History**

Item	Feature	Keyword	Reference	Status M/O/C	Support
1	Covariance time history logical block	N/A	Table 6-6	O	
2	OCM start of a covariance time history section	COV_START	Table 6-6	M	
3	Comment	COMMENT	Table 6-6	O	
4	Identification number for this covariance time history block	COV_ID	Table 6-6	O	
5	Identification number for the previous covariance time history	COV_PREV_ID	Table 6-6	O	
6	Identification number for the next covariance time history	COV_NEXT_ID	Table 6-6	O	
7	Basis of this covariance time history	COV_BASIS	Table 6-6	O	
8	Identification number for the orbit determination, navigation solution, or simulation	COV_BASIS_ID	Table 6-6	O	
9	Reference frame of the covariance time history	COV_REF_FRAME	Table 6-6	M	
10	Epoch of the covariance data reference frame	COV_FRAME_EPOCH	Table 6-6	C	
11	Minimum scale factor to apply to this covariance data	COV_SCALE_MIN	Table 6-6	O	
12	Maximum scale factor to apply to this covariance data	COV_SCALE_MAX	Table 6-6	O	
13	Confidence in the covariance errors matching reality	COV_CONFIDENCE	Table 6-6	O	
14	Covariance composition	COV_TYPE	Table 6-6	M	
15	Covariance element ordering	COV_ORDERING	Table 6-6	M	
16	Units of covariance data line standard deviations (i.e., the square root of the variances supplied in the covariance matrix diagonal elements) that follow the time tag	COV_UNITS	Table 6-6	O	

Item	Feature	Keyword	Reference	Status M/O/C	Support
17	Covariance data	... <Insert covariance data here>	Table 6-6	M	
18	End of a covariance time history section	COV_STOP	Table 6-6	M	

#### A2.5.4.6 OCM Data: Maneuver Specification

Item	Feature	Keyword	Reference	Status M/O/C	Support
1	Maneuver time history logical block	N/A	Table 6-7	O	
2	Start of a maneuver data block	MAN_START	Table 6-7	M	
3	Comment	COMMENT	Table 6-7	O	
4	Identification number for this maneuver	MAN_ID	Table 6-7	M	
5	Identification number for the previous maneuver	MAN_PREV_ID	Table 6-7	O	
6	Identification number for the next maneuver	MAN_NEXT_ID	Table 6-7	O	
7	Basis of this maneuver time history data	MAN_BASIS	Table 6-7	O	
8	Identification number of the orbit determination	MAN_BASIS_ID	Table 6-7	O	
9	Maneuver device identifier	MAN_DEVICE_ID	Table 6-7	M	
10	Completion time of the previous maneuver	MAN_PREV_EPOCH	Table 6-7	O	
11	Start time of the next maneuver	MAN_NEXT_EPOCH	Table 6-7	O	
12	Specifies the purpose of the maneuver	MAN_PURPOSE	Table 6-7	O	
13	Specifies the source of the orbit and/or attitude state(s)	MAN_PRED_SOURCE	Table 6-7	O	
14	Reference frame in which the maneuver vector direction data is provided	MAN_REF_FRAME	Table 6-7	M	
15	Epoch of the maneuver data reference frame	MAN_FRAME_EPOCH	Table 6-7	C	
16	Origin of maneuver gravitational assist body	GRAV_ASSIST_NAME	Table 6-7	O	
17	Duty cycle type to use for this maneuver time history	DC_TYPE	Table 6-7	M	

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Item	Feature	Keyword	Reference	Status M/O/C	Support
18	Start time of the duty cycle-based maneuver window	DC_WIN_OPEN	Table 6-7	C	
19	End time of the duty cycle-based maneuver window	DC_WIN_CLOSE	Table 6-7	C	
20	Minimum number of ‘ON’ duty cycles	DC_MIN_CYCLES	Table 6-7	O	
21	Maximum number of ‘ON’ duty cycles	DC_MAX_CYCLES	Table 6-7	O	
22	Start time of the initial duty cycle-based maneuver sequence	DC_EXEC_START	Table 6-7	C	
23	End time of the final duty cycle-based maneuver sequence	DC_EXEC_STOP	Table 6-7	C	
24	THRUST duty cycle reference time tag	DC_REF_TIME	Table 6-7	C	
25	Thruster pulse ‘ON’ duration	DC_TIME_PULSE_DURATION	Table 6-7	C	
26	Elapsed time between the start of one pulse and the start of the next	DC_TIME_PULSE_PERIOD	Table 6-7	C	
27	Specifies the ‘ON’ reference unit vector direction	DC_REF_DIR	Table 6-7	C	
28	Body reference frame that DC_BODY_TRIGGER direction is expressed in	DC_BODY_FRAME	Table 6-7	C	
29	Body frame reference unit vector ‘trigger’ direction	DC_BODY_TRIGGER	Table 6-7	C	
30	Phase angle offset of thruster pulse start	DC_PA_START_ANGLE	Table 6-7	C	
31	Phase angle of thruster pulse stop	DC_PA_STOP_ANGLE	Table 6-7	C	
32	Set of maneuver elements of information to follow the maneuver time tag	MAN_COMPOSITION	Table 6-7	M	
33	Units of maneuver data line elements that follow the time tag(s)	MAN_UNITS	Table 6-7	O	
34	Maneuver time history data	... < Insert maneuver data here >	Table 6-7	M	
35	End maneuver data block	MAN_STOP	Table 6-7	M	

**A2.5.4.7 OCM Data: Perturbations Specification**

Item	Feature	Keyword	Reference	Status M/O/C	Support
1	Orbit perturbations parameters	N/A	Table 6-10	O	
2	Start of the Perturbations Specification	PERT_START	Table 6-10	M	
3	Comment	COMMENT	Table 6-10	O	
4	Atmosphere model used in the simulation	ATMOSPHERIC_MODEL	Table 6-10	O	
5	Gravity model used in the simulation	GRAVITY_MODEL	Table 6-10	O	
6	Oblate spheroid equatorial radius	EQUATORIAL_RADIUS	Table 6-10	O	
7	Gravitational coefficient of attracting body	GM	Table 6-10	O	
8	'N-body' gravitational perturbations bodies used	N_BODY_PERTURBATIONS	Table 6-10	O	
9	Central body angular rotation rate	CENTRAL_BODY_ROTATION	Table 6-10	O	
10	Inverse of the central body's oblate spheroid oblateness	OBLATE_FLATTENING	Table 6-10	O	
11	Ocean tides model	OCEAN_TIDES_MODEL	Table 6-10	O	
12	Solid tides model	SOLID_TIDES_MODEL	Table 6-10	O	
13	Reduction theory used for precession and nutation modeling	REDUCTION_THEORY	Table 6-10	O	
14	Albedo model	ALBEDO_MODEL	Table 6-10	O	
15	Number of grid points used in the albedo model	ALBEDO_GRID_SIZE	Table 6-10	O	
16	Shadow model used for Solar Radiation Pressure	SHADOW_MODEL	Table 6-10	O	
17	List of planetary bodies for which SRP shadowing is modeled	SHADOW_BODIES	Table 6-10	O	
18	SRP model used	SRP_MODEL	Table 6-10	O	
19	Source and version of the Space Weather data used in the creation of this message	SW_DATA_SOURCE	Table 6-10	O	
20	Epoch of the Space Weather data	SW_DATA_EPOCH	Table 6-10	O	

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Item	Feature	Keyword	Reference	Status M/O/C	Support
21	Method used to select or interpolate any and all sequential space weather data	SW_INTERP_METHOD	Table 6-10	O	
22	A fixed (time invariant) value of the planetary 3-hour-range geomagnetic index Kp	FIXED_GEOMAG_KP	Table 6-10	O	
23	A fixed (time invariant) value of the 3-hourly (equivalent range) geomagnetic index Ap	FIXED_GEOMAG_AP	Table 6-10	O	
24	A fixed (time invariant) value of the planetary 1-hour-range geomagnetic index Dst	FIXED_GEOMAG_DST	Table 6-10	O	
25	A fixed (time invariant) value of the solar flux daily proxy F10.7	FIXED_F10P7	Table 6-10	O	
26	A fixed (time invariant) value of the solar flux 81-day running center-averaged proxy F10.7	FIXED_F10P7_MEAN	Table 6-10	O	
27	A fixed (time invariant) value of the solar flux daily proxy M10.7	FIXED_M10P7	Table 6-10	O	
28	A fixed (time invariant) value of the solar flux 81-day running center-averaged proxy M10.7	FIXED_M10P7_MEAN	Table 6-10	O	
29	A fixed (time invariant) value of the solar flux daily proxy S10.7	FIXED_S10P7	Table 6-10	O	
30	A fixed (time invariant) value of the solar flux 81-day running center-averaged proxy S10.7	FIXED_S10P7_MEAN	Table 6-10	O	
31	A fixed (time invariant) value of the solar flux daily proxy Y10.7	FIXED_Y10P7	Table 6-10	O	
32	A fixed (time invariant) value of the solar flux 81-day running center-averaged proxy Y10.7	FIXED_Y10P7_MEAN	Table 6-10	O	
33	Perturbations Data Block Stop	PERT_STOP	Table 6-10	M	

**A2.5.4.8 OCM Data: Orbit Determination Data**

Item	Feature	Keyword	Reference	Status M/O/C	Support
1	Orbit determination parameters logical block	N/A	Table 6-11	O	
2	Start of an orbit determination data section	OD_START	Table 6-11	M	
3	Comment	COMMENT	Table 6-11	O	
4	Identification number for this orbit determination	OD_ID	Table 6-11	M	
5	Identification number for the previous orbit determination	OD_PREV_ID	Table 6-11	O	
6	Type of orbit determination method used	OD_METHOD	Table 6-11	M	
7	UTC epoch of the orbit determination solved-for state	OD_EPOCH	Table 6-11	M	
8	Days elapsed between first accepted observation and OD_EPOCH	DAYS_SINCE_FIRST_OBS	Table 6-11	O	
9	Days elapsed between last accepted observation and OD_EPOCH	DAYS_SINCE_LAST_OBS	Table 6-11	O	
10	Number of days of observations recommended for the OD	RECOMMENDED_OD_SPAN	Table 6-11	O	
11	Actual time span in days used for the OD	ACTUAL_OD_SPAN	Table 6-11	O	
12	Number of observations available within the actual OD time span	OBS_AVAILABLE	Table 6-11	O	
13	Number of observations accepted within the actual OD time span	OBS_USED	Table 6-11	O	
14	Number of sensor tracks available for the OD within the actual time span	TRACKS_AVAILABLE	Table 6-11	O	
15	Number of sensor tracks accepted for the OD within the actual time span	TRACKS_USED	Table 6-11	O	



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Item	Feature	Keyword	Reference	Status M/O/C	Support
16	Maximum time between observations in the OD	MAXIMUM_OBS_GAP	Table 6-11	O	
17	Positional error ellipsoid $1\sigma$ major eigenvalue at OD	OD_EPOCH_EIGMAJ	Table 6-11	O	
18	Positional error ellipsoid $1\sigma$ intermediate eigenvalue at OD	OD_EPOCH_EIGMED	Table 6-11	O	
19	Positional error ellipsoid $1\sigma$ minor eigenvalue at OD	OD_EPOCH_EIGMIN	Table 6-11	O	
20	Max positional error ellipsoid $1\sigma$ major eigenvalue	OD_MAX_PRED_EIGMAJ	Table 6-11	O	
21	Min positional error ellipsoid $1\sigma$ minor eigenvalue	OD_MIN_PRED_EIGMIN	Table 6-11	O	
22	OD confidence metric, which by definition spans 0 to 100%	OD_CONFIDENCE	Table 6-11	O	
23	Generalized Dilution of Precision for this orbit determination	GDOP	Table 6-11	O	
24	Number of solve-for states in the orbit determination	SOLVE_N	Table 6-11	O	
25	State elements solved for in the orbit determination	SOLVE_STATES	Table 6-11	O	
26	Number of consider parameters used in the orbit determination	CONSIDER_N	Table 6-11	O	
27	Consider parameters used in the orbit determination	CONSIDER_PARAMS	Table 6-11	O	
28	Specific Energy Dissipation Rate	SEDR	Table 6-11	O	
29	Number of sensors used in the orbit determination	SENSORS_N	Table 6-11	O	
30	Sensors used in the orbit determination	SENSORS	Table 6-11	O	
31	Weighted RMS residual ratio	WEIGHTED_RMS	Table 6-11	O	

Item	Feature	Keyword	Reference	Status M/O/C	Support
32	File name(s) and/or associated identification number(s) of TDM (reference [9]) observations	TDM_MSG_LINK	Table 6-11	O	
33	Observation data types utilized in this orbit determination	DATA_TYPES	Table 6-11	O	
34	End of an orbit determination data section	OD_STOP	Table 6-11	M	

#### A2.5.4.9 OCM Data: User-Defined Parameters

Item	Feature	Keyword	Reference	Status M/O/C	Support
1	User-Defined Parameters logical block	N/A	Table 6-12	O	
2	OCM User-Defined Parameters start	USER_START	Table 6-12	M	
3	Comment	COMMENT	Table 6-12	O	
4	As defined by user, 'essential information that cannot be conveyed in COMMENT statements'	(User-defined keywords)	Table 6-12	M	
5	OCM User-Defined Parameters end	USER_STOP	Table 6-12	M	

## ANNEX B

### VALUES FOR SELECTED KEYWORDS

#### (NORMATIVE)

The values in this annex represent the recommended values for selected keywords present in OPM, OMM, OEM, or OCM message. For details and descriptions of the keyword interpretations, the reader is directed to references [H1] and [H7]. The message creator should seek to confirm with the recipient(s) that their software can support the selected keyword value, particularly for more complex content such as reference frames, orbital elements, and covariance definitions.

These recommended values are stored on the SANA Registry, globally accessible on the CCSDS SANA registry website located at:

[https://sanaregistry.org/r/navigation\\_standard\\_normative\\_annexes](https://sanaregistry.org/r/navigation_standard_normative_annexes)

It should be noted that the message creator or recipient may wish to automate processing of SANA registry normative content, which can be done by ingesting and processing of such content in electronic format. These formats can be accessed via the ‘Actions’ link on each registry, for example, for the Orbital Elements registry, a Comma Separated Value (CSV) format can be exported at: [https://www.sanaregistry.org/r/orbital\\_elements?\\_export=csv](https://www.sanaregistry.org/r/orbital_elements?_export=csv) and a JSON format at: [https://www.sanaregistry.org/r/orbital\\_elements?\\_export=json](https://www.sanaregistry.org/r/orbital_elements?_export=json). It should be noted that both the registry and these electronic data formats specify the number of vector elements corresponding to each keyword value.

Exchange partners may submit additional (new) keyword values for consideration for future inclusion into the SANA registry by submitting a detailed email request (<mailto:info@sanaregistry.org>) per annex C, subsection C4. The CCSDS Area or Working Group responsible for the maintenance of the ODM at the time of the request is the approval authority. Until a suggested value is included in the SANA registry, exchange partners may define and use values that are not listed in the SANA registry if mutually agreed between message exchange partners.

#### **B1 MESSAGE ORIGINATORS**

The set of recommended values for the **ORIGINATOR** keyword is enumerated in the *SANA Registry of Organizations*, located at:

<https://sanaregistry.org/r/organizations>

## B2 REFERENCE FRAME CENTERS AND THIRD-BODY PERTURBATIONS

A set of allowed values for the reference frame center keywords (**CENTER\_NAME** for OPM, OEM, OMM, and OCM, as well as **N\_BODY\_PERTURBATIONS** in the OCM) is enumerated in the *SANA Registry of Orbit Centers*, located at:

[https://sanaregistry.org/r/orbit\\_centers](https://sanaregistry.org/r/orbit_centers)

It should be noted that these values may also be useful to specify another platform (satellite, airframe, ground vehicle, etc.) as the reference frame origin to permit the specification of relative positional state time history data. In this case, message authors shall clearly communicate to recipients that the orbit center is not a gravitational center, that propagation of ephemeris vectors or extrapolation of ephemeris start/stop states is not advisable, and that interpolation of state time histories should not be accomplished using classical orbit propagation forces (e.g., gravitational constants, drag).

## B3 TIME SYSTEMS

A set of allowed values for the **TIME\_SYSTEM** keyword is enumerated in the *SANA Registry of Time Systems*, located at:

[https://sanaregistry.org/r/time\\_systems](https://sanaregistry.org/r/time_systems)

## B4 CELESTIAL BODY REFERENCE FRAMES

A set of allowed celestial body reference frame values for **\*\_REF\_FRAME** keywords is enumerated in the *SANA Registry of Celestial Body Reference Frames*, located at:

[https://sanaregistry.org/r/celestial\\_body\\_reference\\_frames](https://sanaregistry.org/r/celestial_body_reference_frames)

## B5 ORBIT-RELATIVE REFERENCE FRAMES

In addition to the above reference frames, maneuver and covariance data may be selected from the list of allowed orbit-relative reference frames using **\*\_REF\_FRAME** keyword values enumerated in the *SANA Registry of Orbit-Relative Reference Frames*, located at:

[https://sanaregistry.org/r/orbit\\_relative\\_reference\\_frames](https://sanaregistry.org/r/orbit_relative_reference_frames)

**It should be noted that two types of orbit-relative local reference frames exist: inertial and rotating. When transforming velocity terms between inertial and rotating frames, remember to properly incorporate the  $(\bar{\omega} \times \bar{r})$  contribution.**

## **B6 ADDITIONAL SPACECRAFT AND ATTITUDE REFERENCE FRAMES**

An additional allowed set of spacecraft and attitude control reference frame values for \***\_REF\_FRAME** keywords is enumerated in the *SANA Registry of Spacecraft and Attitude Control Reference Frames*, located at:

[https://sanaregistry.org/r/spacecraft\\_body\\_reference\\_frames](https://sanaregistry.org/r/spacecraft_body_reference_frames)

In numerous instances, these spacecraft body reference frames are specified by a keyword followed by an 'i' (e.g., ACTUATOR\_i), the 'i' should be replaced by an integer value (1, 2, ...) to denote the 'i<sup>th</sup>' reference frame within the set of those reference frames.

## **B7 ORBITAL ELEMENTS**

A set of allowed values for the **TRAJ\_TYPE** keyword is enumerated in the *SANA Registry of Orbital Elements*, located at:

[https://sanaregistry.org/r/orbital\\_elements](https://sanaregistry.org/r/orbital_elements)

Unique to the OCM, orbit element states and/or time histories may be specified in multiple element set types.

Orbit elements shall be interpreted as osculating elements unless either explicitly specified via the **ORB\_AVERAGING** keyword or as mutually agreed between message exchange partners to contain mean elements (e.g., singly or doubly averaged elements based upon Kozai, Brouwer, or other theories).

Inertial reference frames shall be specified when employing inertial element sets.

When employing non-inertial element sets, inertial reference frames shall not be specified.

## **B8 ADDITIONAL COVARIANCE REPRESENTATIONS**

Covariance matrices may either be specified as representing uncertainties expressed in the above 'Orbital Elements' types (e.g., **COV\_TYPE** may be set to an **TRAJ\_TYPE** such as **CARTPVA**) or may specify an event-based covariance type that includes event time uncertainties as enumerated in the *SANA Registry of Covariance Representations*, located at:

[https://sanaregistry.org/r/orbital\\_covariance\\_matrix\\_types](https://sanaregistry.org/r/orbital_covariance_matrix_types)

## **B9 ATMOSPHERE MODELS**

A set of allowed values for the **ATMOSPHERIC\_MODEL** keyword is enumerated in the *SANA Registry of Atmosphere Models*, located at:

[https://sanaregistry.org/r/atmosphere\\_models](https://sanaregistry.org/r/atmosphere_models)

## **B10 GRAVITY MODELS**

A set of allowed values for the **GRAVITY\_MODEL** keyword is enumerated in the *SANA Registry of Gravity Models*, located at:

[https://sanaregistry.org/r/gravity\\_models](https://sanaregistry.org/r/gravity_models)

## **B11 OBJECT TYPES**

A set of allowed values for the **OBJECT\_TYPE** keyword is enumerated in the *SANA Registry of Object Types*, located at:

[https://sanaregistry.org/r/object\\_types](https://sanaregistry.org/r/object_types)

## **B12 OPERATIONAL STATUS**

A set of allowed values for the **OPS\_STATUS** keyword is enumerated in the *SANA Registry of Operational Status of Space Object*, located at:

[https://sanaregistry.org/r/operational\\_status](https://sanaregistry.org/r/operational_status)

## **B13 ORBIT AVERAGING TECHNIQUES**

A set of allowed values for the **ORB\_AVERAGING** keyword is enumerated in the *SANA Registry of Orbit Averaging Techniques*, located at:

[https://sanaregistry.org/r/orbit\\_averaging](https://sanaregistry.org/r/orbit_averaging)

## **B14 ORBIT CATEGORIES**

A set of allowed values for the **ORBIT\_CATEGORY** keyword is enumerated in the *SANA Registry of Orbit Types*, located at:

[https://sanaregistry.org/r/orbit\\_categories](https://sanaregistry.org/r/orbit_categories)

## **ANNEX C**

### **SECURITY, SANA, AND PATENT CONSIDERATIONS**

#### **(INFORMATIVE)**

##### **C1 SECURITY CONSIDERATIONS**

##### **C2 ANALYSIS OF SECURITY CONSIDERATIONS**

This annex presents the results of an analysis of security considerations applied to the technologies specified in this Recommended Standard.

##### **C3 CONSEQUENCES OF NOT APPLYING SECURITY TO THE TECHNOLOGY**

The consequences of not applying security to the systems and networks on which this Recommended Standard is implemented could include potential loss, corruption, and theft of data. Because these messages are used in preparing pointing and frequency predicts used during spacecraft commanding, and may also be used in collision avoidance analyses, the consequences of not applying security to the systems and networks on which this Recommended Standard is implemented could include compromise or loss of the mission if malicious tampering of a particularly severe nature occurs.

###### **C3.1 POTENTIAL THREATS AND ATTACK SCENARIOS**

Potential threats or attack scenarios include, but are not limited to, (a) unauthorized access to the programs/processes that generate and interpret the messages, (b) unauthorized access to the messages during transmission between exchange partners, and (c) modification of the messages between partners. Protection from unauthorized access during transmission is especially important if the mission utilizes open ground networks, such as the Internet, to provide ground-station connectivity for the exchange of data formatted in compliance with this Recommended Standard. It is strongly recommended that potential threats or attack scenarios applicable to the systems and networks on which this Recommended Standard is implemented be addressed by the management of those systems and networks.

###### **C3.2 DATA PRIVACY**

Privacy of data formatted in compliance with the specifications of this Recommended Standard should be assured by the systems and networks on which this Recommended Standard is implemented.

### **C3.3 DATA INTEGRITY**

Integrity of data formatted in compliance with the specifications of this Recommended Standard should be assured by the systems and networks on which this Recommended Standard is implemented.

### **C3.4 AUTHENTICATION OF COMMUNICATING ENTITIES**

Authentication of communicating entities involved in the transport of data that complies with the specifications of this Recommended Standard should be provided by the systems and networks on which this Recommended Standard is implemented.

### **C3.5 DATA TRANSFER BETWEEN COMMUNICATING ENTITIES**

The transfer of data formatted in compliance with this Recommended Standard between communicating entities should be accomplished via secure mechanisms approved by the Information Technology Security functionaries of exchange participants.

### **C3.6 CONTROL OF ACCESS TO RESOURCES**

Control of access to resources should be managed by the systems upon which originator formatting and recipient processing are performed.

### **C3.7 AUDITING OF RESOURCE USAGE**

Auditing of resource usage should be handled by the management of systems and networks on which this Recommended Standard is implemented.

### **C3.8 UNAUTHORIZED ACCESS**

Unauthorized access to the programs/processes that generate and interpret the messages should be prohibited in order to minimize potential threats and attack scenarios.

### **C3.9 DATA SECURITY IMPLEMENTATION SPECIFICS**

Specific information-security interoperability provisions that may apply between agencies and other independent users involved in an exchange of data formatted in compliance with this Recommended Standard could be specified in an ICD.

## **C4 SANA CONSIDERATIONS**

The following ODM-related items have been registered with the SANA Operator.



- The ODM/XML schema (see section 8).

The following normative ODM elements should be selected from the SANA registry (see annex B):

- ODM originators;
- spacecraft identifiers;
- Reference Frame Center and Third-Body Perturbations;
- time systems;
- Reference Frames (inertial, quasi-inertial, orbit-relative, spacecraft and attitude frames);
- orbital element set and covariance matrix composition definitions;
- atmosphere models;
- gravity models;
- object types;
- operational status;
- orbit averaging techniques;
- orbit types.

The registration rule for new entries in the SANA registry is the approval of new requests by the CCSDS Area or Working Group responsible for the maintenance of the ODM at the time of the request. New requests for this registry should be sent to SANA (<mailto:info@sanaregistry.org>).

## **C5 PATENT CONSIDERATIONS**

The recommendations of this document have no patent issues.

## ANNEX D

## ABBREVIATIONS AND ACRONYMS

## (INFORMATIVE)

<u>Term</u>	<u>Meaning</u>
ADM	Attitude Data Message
ASCII	American Standard Code for Information Interchange
CATS	Critical Angle to the Sun
CCSDS	Consultative Committee for Space Data Systems
CDM	Conjunction Data Message
COSPAR	United Nations Committee on Space Research
DSST	Draper Semi-Analytic Satellite Theory
EOP	Earth Orientation Parameters
GPS	Global Positioning System
HEO	high Earth orbit
IAU	International Astronomical Union
ICD	Interface Control Document
ICRF	International Celestial Reference Frame
IEC	International Electro-technical Commission
IIRV	Improved Inter-Range Vector
IOD	Initial Orbit Determination
ISO	International Organization for Standardization
ITRF	International Terrestrial Reference Frame
GDoP	generalized dilution of precision
GRC	Greenwich Rotating Coordinate Frame
J2000	Earth Mean Equator and Equinox of J2000 (Julian Date 2000)
KVN	keyword = value notation
LEO	low Earth orbit
LS	Least Squares
LTM	Lower Triangular Matrix
MARSGRAM	Mars Global Reference Atmosphere Model
MSIS	Mass-Spectrometer-Incoherent-Scatter

NORAD	North American Aerospace Defense Command
OD	Orbit Determination
ODM	Orbit Data Message
OEB	Optimally Encompassing Box
OEM	Orbit Ephemeris Message
OCM	Orbit Comprehensive Message
OMM	Orbit Mean-Elements Message
OPM	Orbit Parameter Message
PoC	Point-of-Contact
PRM	Pointing Request Message
RDM	Reentry Data Message
RMS	Root Mean Square
RTN	Radial, Transverse (along-track), and Normal
RSO	Resident Space Object
SEDR	Specific Energy Dissipation Rate
SGP	Simplified General Perturbations
SGP4	US Air Force Simplified General Perturbations No. 4
SP	Sequential Processing
SPK	Satellite, Planetary Kernel
SRP	Solar Radiation Pressure
SSA	Space Situational Awareness
TAI	International Atomic Time
TCB	Barycentric Coordinate Time
TDB	Barycentric Dynamical Time
TDM	Tracking Data Message
TDR	True of Date Rotating
TEME	True Equator Mean Equinox
TLE	Two Line Element
TOD	True Equator and Equinox of Date
TT	Terrestrial Dynamical Time (see also 'TDT')
USM	Universal Semi-analytical Method
UTC	Coordinated Universal Time

UTM Upper Triangular Matrix

VENUSGRAM Venus Global Reference Atmosphere Model

W3C World Wide Web Consortium

WGS World Geodetic System

XML Extensible Markup Language

## ANNEX E

### RATIONALE FOR THIS STANDARD

#### (INFORMATIVE)

##### E1 OVERVIEW

This annex presents the rationale behind the design of each message. It may help the application engineer to select a suitable message.

A specification of requirements agreed to by all parties is essential to focus design and to ensure the product meets the needs of the Member Agencies and satellite operators. There are many ways of organizing requirements, but the categorization of requirements is not as important as the agreement to a sufficiently comprehensive set. In this annex, the requirements are organized into three categories:

- a) **Primary Requirements:** These are the most elementary and necessary requirements. They would exist no matter the context in which the CCSDS is operating, that is, regardless of pre-existing conditions within the CCSDS, its Member Agencies, or other independent users.
- b) **Heritage Requirements:** These are additional requirements that derive from pre-existing Member Agency or other independent user requirements, conditions, or needs. Ultimately, these carry the same weight as the Primary Requirements. This Recommended Standard reflects heritage requirements pertaining to some of the CCSDS Areas' home institutions collected during the preparation of the document; it does not speculate on heritage requirements that could arise from other sources. Corrections and/or additions to these requirements are expected during future updates.
- c) **Desirable Characteristics:** These are not requirements, but they are felt to be important or useful features of the Recommended Standard.

**E2 REQUIREMENTS ACCEPTED BY THE ORBIT DATA MESSAGES**

**E2.1 PRIMARY REQUIREMENTS**

#	Requirement	OPM?	OMM?	OEM?	OCM?
P1	Data must be provided in digital form (computer file).	Y	Y	Y	Y
P2	The file specification must not require of the receiving exchange partner the separate application of, or modeling of, spacecraft dynamics or gravitational force models, or integration or propagation.	N	N	Y	Y
P3	The interface must facilitate the receiver of the message to generate a six-component Cartesian state vector (position and velocity) at any required epoch.	Y	Y	Y	Y
P4	State vector information must be provided in a reference frame that is clearly identified and unambiguous.	Y	Y	Y	Y
P5	Identification of the object and the center(s) of motion must be clearly identified and unambiguous.	Y	Y	Y	Y
P6	Time measurements (time stamps, or epochs) must be provided in a commonly used, clearly specified system.	Y	Y	Y	Y
P7	The time bounds of the ephemeris must be unambiguously specified.	N/A	N/A	Y	Y
P8	The Recommended Standard must provide for clear specification of units of measure.	Y	Y	Y	Y
P9	Files must be readily ported between, and useable within, 'all' computational environments in use by Member Agencies.	Y	Y	Y	Y
P10	Files must have means of being uniquely identified and clearly annotated. The file name alone is considered insufficient for this purpose.	Y	Y	Y	Y
P11	File name syntax and length must not violate computer constraints for those computing environments in use by Member Agencies.	Y	Y	Y	Y
P12	A means to convey information about the uncertainty of the state shall be provided.	Y	Y	Y	Y

**E2.2 HERITAGE REQUIREMENTS**

#	Requirement	OPM?	OMM?	OEM?	OCM?
H1	Ephemeris data is reliably convertible into the SPICE SPK (NASA) format (reference [H4]) and IIRV (NASA) format (reference [H5]) using a standard, multi-mission, unsupervised pipeline process. A complete ephemeris, not subject to integration or propagation by the customer, must be provided.	N	N	Y	Y
H2	Ephemeris data provided for scheduling or operations (metric predicts) is to be certified by the providing Agency as correct and complete for the intended purpose. The receiving Agency cannot provide evaluation, trajectory propagation, or other usability services.	N	N	Y	Y
H3	The ODM shall provide a mechanism by which messages may be uniquely identified and clearly annotated. It facilitates discussion between the recipient and the message originator, should that be necessary.	Y	Y	Y	Y
H4	The ODM shall provide a mechanism by which maneuvers may be uniquely identified and clearly annotated. It facilitates discussion between the recipient and the message originator, should that be necessary.	N	N	N	Y
H5	The Recommended Standard is, or includes, an ASCII format.	Y	Y	Y	Y
H6	The Recommended Standard does not require software supplied by other Agencies.	Y	N	Y	Y

**E2.3 DESIRABLE CHARACTERISTICS**

#	Requirement	OPM?	OMM?	OEM?	OCM?
DC1	The Recommended Standard applies to non-traditional objects, such as landers, rovers, balloons, and natural bodies (e.g., asteroids, comets).	Y	N	Y	Y
DC2	The Recommended Standard allows state vectors to be provided in other than the traditional J2000 inertial reference frame; one example is the International Astronomical Union (IAU) Mars body-fixed frame. (In such a case, provision or ready availability of supplemental information needed to transform data into a standard frame must be arranged.)	Y	Y	Y	Y
DC3	The Recommended Standard is extensible with no disruption to existing users/uses.	Y	Y	Y	Y

#	Requirement	OPM?	OMM?	OEM?	OCM?
DC4	The Recommended Standard is consistent with, and ideally a part of, ephemeris products and processes used for other space science purposes.	Y	Y	Y	Y
DC5	The Recommended Standard is as consistent as reasonable with any related CCSDS ephemeris Recommended Standards used for Earth-to-spacecraft or spacecraft-to-spacecraft applications.	Y	Y	Y	Y

#### E2.4 APPLICABILITY OF CRITERIA TO MESSAGE OPTIONS

The selection of one message will depend on the optimization criteria in the given application. The following table compares the four recommended messages in terms of the relevant selection criteria identified by the CCSDS.

#### E2.5 APPLICABILITY OF THE CRITERIA TO ORBIT DATA MESSAGES

Criteria	Definition	Applicable to OPM?	Applicable to OMM?	Applicable to OEM?	Applicable to OCM?
Modeling Fidelity	Permits modeling of any dynamic perturbation to the trajectory.	N	N	Y	Y
Human Readability	Provides an easily readable message corresponding to a widely used orbit representation.	Y	Y	Y	Y
Remote Body Extensibility	Permits use for assets on remote solar system bodies.	Y	N	Y	Y
Lander/Rover Compatibility	Permits exchange of non-orbit trajectories.	N	N	Y	Y

### E3 INCREASING ORBIT PROPAGATION FIDELITY OF AN OPM OR OMM

Some OPM or OMM users may desire/require a higher fidelity propagation of the state vector or Keplerian elements. A higher fidelity technique may be desired/required to minimize inconsistencies in predictions generated by diverse, often operator-unique propagation schemes. Nominally, the OPM and OMM are engineered only for low- to medium-fidelity orbit propagation. However, with the inclusion of additional context information, it is possible for users to provide data that could be used to provide a relatively higher fidelity orbit propagation. For this relatively higher fidelity orbit propagation, a much greater amount of ancillary information regarding spacecraft properties and dynamical models should be provided. Higher fidelity orbit propagations may be useful in special studies such as orbit conjunction studies.



Spacecraft orbit determination is a stochastic estimation problem; observations are inherently uncertain, and not all of the phenomena that influence satellite motion are clearly discernible. State vectors and Keplerian elements with their respective covariances are best propagated with models that include the same forces and phenomena that were used for determining the orbit. Including this information in an OPM or OMM allows exchange partners to compare the results of their respective orbit propagations.

With additional context information, the OPM and OMM may be used for assessing mutual physical or electromagnetic interference among Earth-orbiting spacecraft, developing collaborative maneuvers, and propagating the orbits of active satellites, inactive man-made objects, and near-Earth debris fragments. The additional information facilitates dynamic modeling of any user’s approach to conservative and non-conservative phenomena.

The primary vehicle for the provision of additional optional ancillary information to be used when propagating an OPM or OMM is the COMMENT mechanism. Alternatively, the ‘USER\_DEFINED\_’ keyword prefix may be used, though this usage is not encouraged.

**E4 SERVICES RELATED TO THE DIFFERENT ORBIT DATA MESSAGE FORMATS**

The different orbit data messages have been distinguished by the self-interpretability of the messages. The different services that can be achieved without special arrangements between users of the CCSDS orbit data messages are listed in table E-1

**Table E-1: Services Available with Orbit Data Messages**

Service	Definition	Applicable to OPM?	Applicable to OMM?	Applicable to OEM?	Applicable to OCM?
Absolute Orbit Interpretation	State availability at specific times for use in additional computations (geometry, event detection, etc.).	Y	Y	Y	Y
Relative Orbit Interpretation	Trajectory comparison and differencing for events based on the same time source.	Only at time specified at Epoch	Only at time specified at Epoch	Y	Y

## ANNEX F

## TECHNICAL MATERIAL AND CONVENTIONS FOR ODM DATA

## (INFORMATIVE)

## F1 SATELLITE PHYSICAL CHARACTERISTICS: OPTIMALLY ENCOMPASSING BOX

This section of the informative technical annex defines satellite dimensional and orientational parameters of the OCM's satellite physical characteristics specification.

To facilitate improved modeling of the physical space occupied by a space object, the space object's attitude/orientation, the probability of a hard body collision occurring, and drag and SRP acceleration forces, the OCM allows the specification of an OEB. It should be noted that the OEB describes the physical space occupied by the space object, which may or may not align with the inertia tensor for that object.

For a box-shaped satellite (e.g., a CubeSat) without appendages, the satellite's volume in three-dimensional space and a corresponding OEB would have a one-to-one mapping.

For a satellite having solar arrays that extend from the spacecraft body structure, the OEB would extend from the main satellite body to encompass the deployed solar arrays as well.

The OEB shape is shown in figure F-1 below. As illustrated, the OEB reference frame axes (depicted as **RED** dotted lines) are defined by convention as follows:

The OEB x-axis is along the **longest** dimension of the box ( $\hat{X}_{OEB\_MAX}$ ). This is sometimes referred to the 'span' of the space object.

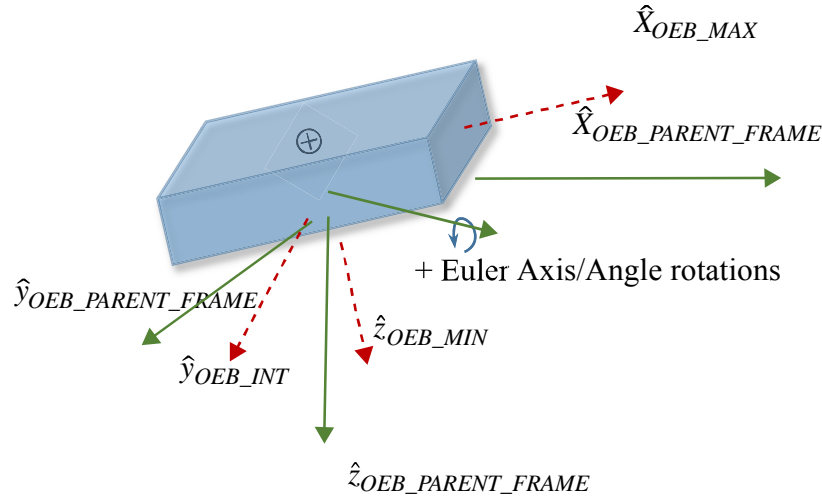
The OEB y-axis is along the **intermediate** orthonormal dimension ( $\hat{Y}_{OEB\_INT}$ ).

The OEB z-axis is along the **shortest** orthonormal dimension ( $\hat{Z}_{OEB\_MIN}$ ).

The BOX shape can easily represent a cube by setting all orthonormal dimensions equal. If the longest two or three orthonormal dimensions are equivalent, then  $\hat{X}_{OEB\_MAX}$  is defined as the direction along one of those longest dimensions, and the next as  $\hat{Y}_{OEB\_INT}$ .

If the longest two or three principal axis dimensions of the box are equivalent,  $\hat{X}_{OEB\_MAX}$  is defined as the direction along one of those longest principal dimensions, and the next as  $\hat{Y}_{OEB\_MED}$ .

The OEB z-axis is always defined as:  $\hat{Z}_{OEB\_MIN} = \hat{X}_{OEB\_MAX} \times \hat{Y}_{OEB\_INT}$ .



**Figure F-1: Depiction of Optimally Enclosing Box and Definitions of MAX, INT, and MIN Orientation Vectors Relative to OEB Parent Frame**

NOTE – Parent and body axis are shown in proximity to each other for display purposes only but could generally be in any orientation as specified by the quaternion.

A fixed orientation of the Optimally Encompassing Box with respect to the user-specified ‘OEB\_PARENT\_FRAME’ is defined using a quaternion that maps from the user-specified OEB\_PARENT\_FRAME to the Optimally Encompassing Box vector directions. The above figure shows the proper definitions and sign conventions. The resulting transformation sequence is:

$$\begin{bmatrix} x \\ y \\ z \end{bmatrix}_{OEB} = [M] \begin{bmatrix} x \\ y \\ z \end{bmatrix}_{OEB\_PARENT\_FRAME}$$

Where the frame transformation matrix [M] is a function of the quaternion components:

$$[M] = \begin{bmatrix} Q_1^2 - Q_2^2 - Q_3^2 + Q_c^2 & 2(Q_1Q_2 + Q_3Q_c) & 2(Q_1Q_3 - Q_2Q_c) \\ 2(Q_1Q_2 - Q_3Q_c) & -Q_1^2 + Q_2^2 - Q_3^2 + Q_c^2 & 2(Q_2Q_3 + Q_1Q_c) \\ 2(Q_1Q_3 + Q_2Q_c) & 2(Q_2Q_3 - Q_1Q_c) & -Q_1^2 - Q_2^2 + Q_3^2 + Q_c^2 \end{bmatrix}$$

The physical dimensions of the OEB (long, intermediate, and short dimensions) are specified via OEB\_MAX, OEB\_INT, and OEB\_MIN, respectively.

The cross-sectional area is modeled in the OCM as a combination of two parameter types:

- a) an attitude-independent, constant cross-sectional area (e.g., DRAG\_CONST\_AREA or SRP\_CONST\_AREA);
- b) attitude-dependent cross-sectional area as viewed along the OEB x, y, and z axes (long, intermediate, and short dimension directions) via

AREA\_ALONG\_OEB\_MAX, AREA\_ALONG\_OEB\_INT, and AREA\_ALONG\_OEB\_MIN, respectively.

The analyst may use one or a combination of both parameter types to best represent the total cross-sectional area profile of the space object to be used in drag, lift, and SRP force estimates. The total cross-sectional area observed when viewed from an arbitrary unit-vector direction  $[x \ y \ z]$  could be:

$$\text{TOTAL\_AREA} + \begin{bmatrix} \text{AREA\_ALONG\_OEB\_MAX} \\ \text{AREA\_ALONG\_OEB\_INT} \\ \text{AREA\_ALONG\_OEB\_MIN} \end{bmatrix} \cdot [M] \begin{bmatrix} \widehat{x} \\ y \\ z \end{bmatrix}_{\text{OEB\_PARENT\_FRAME}} = \text{DRAG\_CONST\_AREA}$$

For example, to model drag forces, a two-meter diameter spherical space object would be best modeled as a constant area (DRAG\_CONST\_AREA=3.1415m<sup>2</sup>) with all three AREA\_ALONG\_OEB parameters defaulting to zero. Conversely, a ten-meter long, 10 cm box-cross section gravity gradient boom would best be modeled by leaving DRAG\_CONST\_AREA defaulting to zero and setting AREA\_ALONG\_OEB\_MAX = AREA\_ALONG\_OEB\_INT = 1 m<sup>2</sup> and AREA\_ALONG\_OEB\_MIN = 0.01 m<sup>2</sup>. Finally, one can model a sphere encircling the gravity gradient boom's centroid using a combination of these approaches by setting DRAG\_CONST\_AREA = 3.1415m<sup>2</sup>, AREA\_ALONG\_OEB\_MAX = AREA\_ALONG\_OEB\_INT = 1 m<sup>2</sup> and AREA\_ALONG\_OEB\_MIN = 0.01 m<sup>2</sup>.

As a second example, when modeling SRP forces, it should be noted that many GEO spacecraft have very large solar arrays that dominate the total cross-sectional surface of the spacecraft. Importantly, these solar arrays are designed to remain as normal to the Sun as possible irrespective of current spacecraft bus orientation. As such, the cross-sectional area of these arrays is best portrayed in the OCM by setting SRP\_CONST\_AREA to the combined solar array surface area as observed along the Sun viewing vector, and the remaining bus cross-sectional areas can be set using the three attitude-dependent AREA\_ALONG\_OEB parameters.

## F2 APPARENT-TO-ABSOLUTE VISUAL MAGNITUDE RELATIONSHIP

This section of the informative technical annex presents the relationships to be used to map apparent to absolute visual magnitude for inclusion in an OCM. These equations, based on reference [H8], examine signal magnitude for reflected illumination by a Resident Space Object (RSO) that is exoatmospheric, meaning that its illumination by the Sun is not reduced or impeded by atmospheric transmission losses. The equations do not account for spatial distribution across multiple detectors, which involves characterizing the Point Spread Function of the system.

Definitions:

$A_{\text{Target}}$  Effective area of the target [m<sup>2</sup>].  
 $E_{\text{EntranceAperture}}$  The point source irradiance reaching the sensor aperture [W/m<sup>2</sup>].

$d_{SunToTarget}$	Distance from the sun to the target [m] (e.g., 1 AU = $1.4959787066 \times 10^{11}$ m).
$d_{TargetToSensor}$	Distance from target to sensor [m].
$dia_{Target}$	Effective diameter of the target, [m].
$E_{Sun}$	Exoatmospheric solar irradiance, nominally 1380 [ $W/m^2$ ] at 1 AU.
$E_{Target}$	Target Irradiance at Sensor without atmospheric loss [ $W/m^2$ ].
$E_0$	Ref. Visual Magnitude (Vega) Irradiance [ $2.77894 \times 10^{-8} W/m^2$ ].
$F$	General shadowing term accounting for the penumbra region's influence [unitless, $0 \leq F \leq 1$ , 0 = umbra, and 1 = full Sun illumination].
$I_{Sun}$	Solar Intensity $\approx 3.088374161 \times 10^{25}$ [ $W/sr$ ].
$I_{Target}$	Intensity of reflected energy from target treated as a point source [ $W/sr$ ].
$Phase(\varphi)$	Geometric reflectance phase function [unitless, $0 < Phase(\varphi) \leq 1$ ].
$\varphi$	Critical Angle to the Sun (CATS) from sun to the sensor, as shown in figure F-2 and referenced to the observed target [rad].
$\pi$	Pi constant.
$\rho$	Reflectance of the target [between 0 (none) and 1 (perfect reflectance)].
$\tau_{Atmosphere}$	Effective transmission of the atmosphere [unitless, $0 < \tau \leq 1$ ].

Given an optical sensor's measured target entrance aperture radiance:

$$E_{target} = \frac{E_{EntranceAperture}}{\tau_{Atmosphere}} [W/m^2]$$

$$VM_{apparent} = -2.5 \log_{10} \frac{E_{target}}{E_0}, \text{ measured on the visual magnitude scale}$$

$$\text{or if } VM_{apparent} \text{ known: } E_{target} = E_0 10^{\left[-\frac{VM_{apparent}}{2.5}\right]}$$

$$I_{target} = E_{target} d_{TargetToSensor}^2 [W]$$

$$E_{Sun} = \frac{I_{Sun}}{d_{SunToTarget}^2} [W/m^2]$$

$$Phase(\varphi) = \frac{\sin \varphi + (\pi - \varphi) \cos \varphi}{\pi}$$

$$A_{Target} = \frac{\pi I_{Target}}{\rho F E_{Sun} Phase(\varphi)} [m^2]$$

## NOTES

- 1  $A_{Target}$  is undefined in umbra ( $F=0$ =darkness), or no reflection ( $\rho = 0$ ).
- 2 If reflectance is unknown, one may assume a standard reference reflectance of fifteen percent]

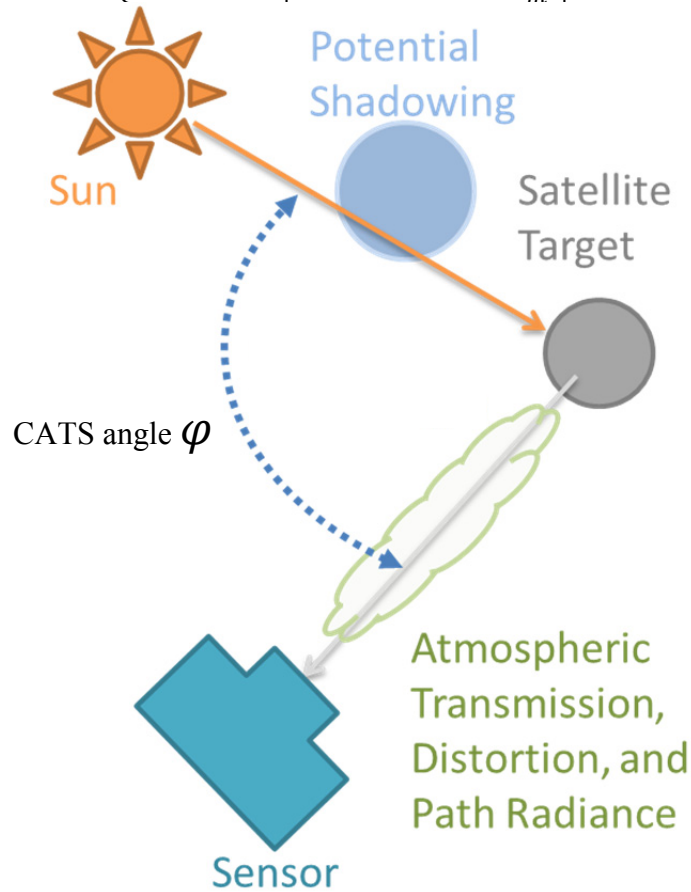
From which an effective diameter of the physical object can be roughly approximated as:

$$dia_{Target} \approx \sqrt{\frac{4 A_{Target}}{\pi}}$$

From the above equations,  $VM_{absolute}$  ‘normalized’ to a 1 AU Sun-to-target distance, a phase angle of  $0^\circ$  and an example reference 40,000 km target-to-sensor distance (equivalent to a GEO satellite tracked at  $15.6^\circ$  elevation above the optical site’s local horizon), is obtained as:

$$VM_{absolute} = -2.5 \log_{10} \left\{ \frac{E_{target}}{E_0} \right\}, \text{ from which:}$$

$$VM_{absolute} = -2.5 \log_{10} \left\{ \frac{[E_{Sun1AU}=1380 W/m^2] [Phase(0 rad)=1.0] [\rho A_{Target} \text{ from above, in } m^2]}{\pi [E_0=2.77894 \times 10^{-8} W/m^2] [(40,000,000^2) m^2]} \right\}$$



**Figure F-2: Depiction of Optical Viewing CATS Angle Geometry**

### F3 MANEUVER AND DUTY CYCLE DIAGRAMS

This section of the informative technical annex defines time-based and phase-angle-based duty cycle parameters.

A ‘duty cycle’ is a cycle of thruster operation that operates intermittently rather than continuously, having an ‘on’ interval followed by an ‘off’ interval.

#### Time-based duty cycle parameters

Time-based duty cycle parameters define a window of duty cycle operations, the actual execution interval and ‘ON’ and ‘OFF’ intervals, as shown in figure F-3.

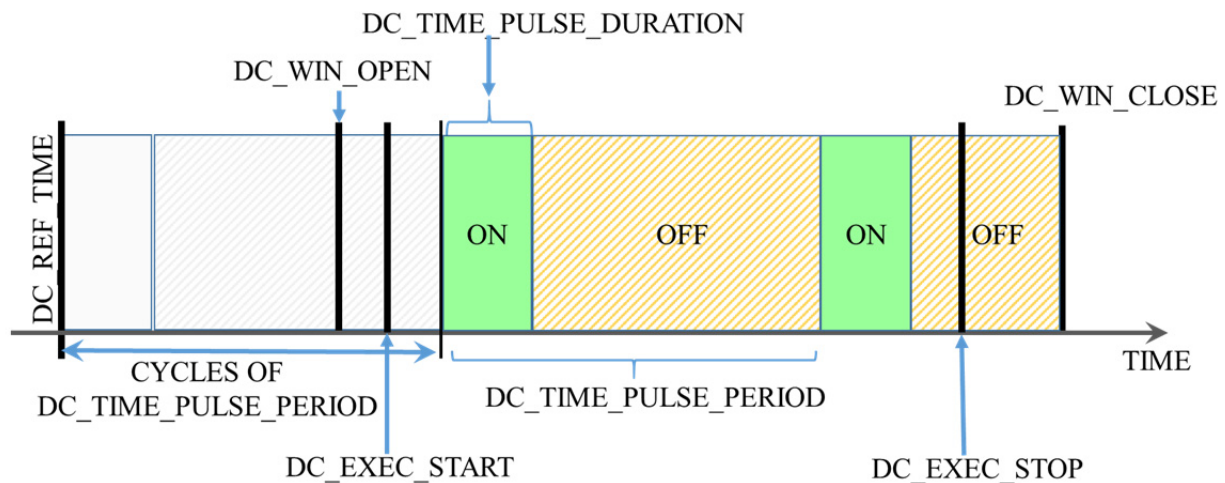
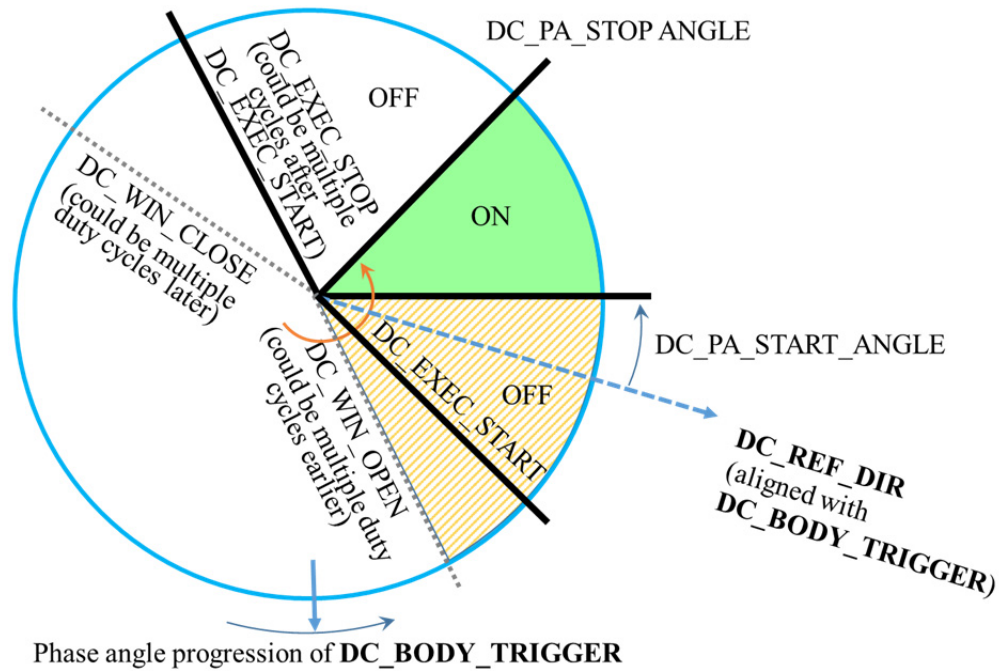


Figure F-3: Diagram of Time-based Duty Cycle (DC\_TYPE = ‘TIME’)

#### Angle-based duty cycle parameters

Angle-based duty cycle parameters also define a window of duty cycle operations and actual execution interval and ‘ON’ and ‘OFF’ intervals, but in this case the ‘ON’ and ‘OFF’ intervals are triggered by angular limits as shown in figure F-4.



**Figure F-4: Diagram of a Rotating Spacecraft Body's Progression through an Inertial Clock Angle-based Duty Cycle (DC\_TYPE = 'TIME\_AND\_ANGLE')**

#### F4 ORBIT DETERMINATION GENERALIZED DILUTION OF PRECISION (GDOP) FORMULATION

This section of the informative technical annex defines the Generalized Dilution of Precision (GDOP) formulation used in the orbit determination section of the OCM.

As described in reference [H16], GDOP provides a method to assess the navigation performance over a time-integrated orbit solution. GDOP broadens the DoP concept from the more common instantaneous geometric or kinematic solution of multiple transmit sources at one time to a scenario associated with a receiver that can integrate metric range and/or Doppler (or range-rate) measurements over time, potentially from different transmit sources, to estimate the user's orbital position and velocity state. It is defined as a function of the sum of information matrices to obtain an observability grammian associated with a set of metric tracking measurements collected over time.

The following equation for GDOP represents the uncertainty of an orbit state estimate as observed over time.

$$GDOP = \sqrt{\max \left[ eig \left( \left( \sum_{t_0}^{t_n} \tilde{H}_0^T W \tilde{H}_0 \right)^{-1} \right) \right]}$$



Where  $H$  represents the measurement matrix modeled in the state estimate, and  $W$  is a diagonal matrix of relative weights that represents the accuracy of the measurements.  $\tilde{H}_0^T W \tilde{H}_0$  represents the information matrix, the inverse of which is the covariance matrix. By summing over time, one obtains an estimate of the state uncertainty from the time-derived measurement set.

## F5 EULER AXIS/ANGLE INTERPOLATION

The Euler Axis and Angle representation of Euler’s Theorem (see reference [H17] 10–14) is an effective way to interpolate a series of covariance matrices, reference frames or maneuvers, thrust, or acceleration vector directions.

### Interpolation of a series of three-dimensional vectors:

As presented in reference [H17], and consistent with the nomenclature of reference [H1] where  $e_1$ ,  $e_2$ , and  $e_3$  represent the three vector components of axis of rotation  $\hat{e}$  and  $\varphi$  represents the angle of rotation, a time-based interpolation of adjacent unit vectors  $\hat{v}_A$  and  $\hat{v}_B$  in a reference frame can be undertaken as:

a) The axis of rotation  $\hat{e}$  can be obtained as:  $\hat{e} = \frac{\hat{v}_B \times \hat{v}_A}{|\hat{v}_B \times \hat{v}_A|}$ ;

b) Assuming a constant rotational rate during this interval,  $\varphi(t) = \frac{(t-t_1) \cos^{-1}(\hat{v}_A \cdot \hat{v}_B)}{(t_2-t_1)}$ ;

c) The orthonormal rotation matrix  $[M(t)]$  is then:

$$= \begin{pmatrix} (1 - \cos \varphi) \hat{e}_x^2 + \cos \varphi & (1 - \cos \varphi) \hat{e}_x \hat{e}_y + \hat{e}_z \sin \varphi & (1 - \cos \varphi) \hat{e}_x \hat{e}_z - \hat{e}_y \sin \varphi \\ (1 - \cos \varphi) \hat{e}_y \hat{e}_x - \hat{e}_z \sin \varphi & (1 - \cos \varphi) \hat{e}_y^2 + \cos \varphi & (1 - \cos \varphi) \hat{e}_y \hat{e}_z + \hat{e}_x \sin \varphi \\ (1 - \cos \varphi) \hat{e}_z \hat{e}_x + \hat{e}_y \sin \varphi & (1 - \cos \varphi) \hat{e}_z \hat{e}_y - \hat{e}_x \sin \varphi & (1 - \cos \varphi) \hat{e}_z^2 + \cos \varphi \end{pmatrix};$$

d) From which the interpolated vector at time  $t$  is then  $\hat{v}(t) = [M(t)]\hat{v}_A$ ;

e) The accompanying vector magnitudes (e.g., eigenvalues or thrust or acceleration magnitudes) may be interpolated using Lagrange polynomials or linear expressions.

### Interpolation of a series of reference (or covariance eigenvector) frames:

The eigenvector matrix  $[E(t)]$  contains the row-wise storage of the major, intermediate, and minor eigenvectors at time  $t$ , taking care to ensure that this ordered ‘triad’ of vectors adheres to the righthand rule. When interpolating between two eigenvector matrices  $[E_1]$  and  $[E_2]$ , derived from two adjacent covariance matrices, respectively,  $[E(t)]$  can be evaluated as:

a) The rotation occurring between  $[E_1]$  and  $[E_2]$  is:  $[M_{BA}] = [E_2][E_1]^T$ ;

b) Compute  $\sigma = (M_{BA11} + M_{BA22} + M_{BA33})$ ;

- c) The angle of rotation from A to B is:  $\varphi_{BA} = \cos^{-1} \left[ \frac{1}{2}(\sigma - 1) \right]$ ;
- d) Exercising caution to accommodate nonunique cases (when  $\sin \varphi = 0$ ) as described in reference [H17], the axis of rotation
- $$\hat{e} = \left[ \frac{(M_{BA23} - M_{BA32})}{2 \sin \varphi} \quad \frac{(M_{BA31} - M_{BA13})}{2 \sin \varphi} \quad \frac{(M_{BA12} - M_{BA21})}{2 \sin \varphi} \right];$$
- e) The angle of rotation at time t is  $\varphi(t) = \frac{(t - t_1)\varphi_{BA}}{(t_2 - t_1)}$ ,
- f)  $[M(t)]$  can be computed using the above expression in step (3);
- g) And finally, the eigenvector matrix  $[E(t)] = [M(t)][E_1]$ ;
- h) When interpolating a series of covariance matrices, the accompanying eigenvalues may be interpolated using Lagrange polynomials or linear expressions.

## F6 REGULAR EXPRESSIONS FOR VALIDATION AND INGEST OF ODMs

To accomplish validation and ingest of KVN versions of the ODM, the use of Regular Expressions (referred to as ‘Regex’) is strongly encouraged whenever possible. Regex offers a detailed and rigorous way to ensure proper validation, interpretation, and conformance to Orbit Data Message content. Most programming languages support the Regex feature, including C#, C++, Delphi, HTML5, Java and Javascript, MySQL, Oracle, PCRE, Perl, PHP, PowerShell, Python, R, Ruby, Scala, TCL, VBscript, Visual Basic, and XML Schema.

While these RegEx sequences can provide a good level of validation of the entries, the reader is cautioned that using them on a long series of orbit or covariance data can be very inefficient and slow. RegEx sequences are best utilized on individual values such as Keyword = SingleValue.

### **Sample Regular Expression for ‘Keyword = CCSDS Date/Time Format’**

The CCSDS Timecode format specified in 7.5.10 provides a convenient illustration of the power of using regular expressions. The color-coded Regex string below may be used to readily match any general ODM KVN line that sets a KVN keyword to a Timecode value. The group naming capability (color-coded in green below) inherent in Regex is particularly useful, by which the keyword, year, month, day, hour, minute, and second can be readily extracted and processed. As shown in figure F-5, this Regex sequence enforces the requirement that KVN keywords must be uppercase and can only consist of letters A-Z, digits 0-9, or underscores. It should be noted that in this expression, the optional inclusion by the message creator of one or more white space characters with the ‘(s\*)?’ sequences allow for maximum flexibility while still retaining a rigorous validation.

NOTE – In some languages, ‘^’ and ‘\$’ matching at string line breaks must explicitly be enabled to process a string containing a series of lines of the message.

```
^(?:\s*)?(?<keyword>[0-9A-Z_]*)(?:\s*)?=(?:\s*)?(?<yr>(?:\d{4}))-(?<mo>(?:\d{1,2}))-(?<dy>(?:\d{1,2}))T(?<hr>(?:\d{1,2})):(?<mn>(?:\d{1,2})):(?<sc>(?:\d{0,2})(?:\.\d*)?)?(?:\s*)?$
```

Figure F-5: Regex Pattern for CCSDS Timecode

**Regex Pattern Matching Sequence:**

Applying the above CCSDS Timecode Regex to a file containing the string ‘CREATION\_DATE = 2020-09-13T00:09:47.059345’ as an example, figure F-6 illustrates how this Regex expression is used to rigorously validate and match the string and identify the specified group names.

```

Beginning match attempt at character 45
Match attempt failed after 1 steps
Beginning match attempt at character 46
ok
ok
ok
CREATION_DATE
CREATION_DATE
CREATION_DATE
CREATION_DATE
CREATION_DATE =
CREATION_DATE =
CREATION_DATE = 2020
CREATION_DATE = 2020
CREATION_DATE = 2020
CREATION_DATE = 2020-
CREATION_DATE = 2020-09
CREATION_DATE = 2020-09
CREATION_DATE = 2020-09
CREATION_DATE = 2020-09-
CREATION_DATE = 2020-09-13
CREATION_DATE = 2020-09-13
CREATION_DATE = 2020-09-13
CREATION_DATE = 2020-09-13T
CREATION_DATE = 2020-09-13T00
CREATION_DATE = 2020-09-13T00
CREATION_DATE = 2020-09-13T00
CREATION_DATE = 2020-09-13T00:
CREATION_DATE = 2020-09-13T00:09
CREATION_DATE = 2020-09-13T00:09
CREATION_DATE = 2020-09-13T00:09
CREATION_DATE = 2020-09-13T00:09:
CREATION_DATE = 2020-09-13T00:09:47
CREATION_DATE = 2020-09-13T00:09:47.
CREATION_DATE = 2020-09-13T00:09:47.059345
CREATION_DATE = 2020-09-13T00:09:47.059345
CREATION_DATE = 2020-09-13T00:09:47.059345
CREATION_DATE = 2020-09-13T00:09:47.059345
CREATION_DATE = 2020-09-13T00:09:47.059345
CREATION_DATE = 2020-09-13T00:09:47.059345
CREATION_DATE = 2020-09-13T00:09:47.059345
CREATION_DATE = 2020-09-13T00:09:47.059345backtrac
CREATION_DATE = 2020-09-13T00:09:47.059345ok
CREATION_DATE = 2020-09-13T00:09:47.059345ok
CREATION_DATE = 2020-09-13T00:09:47.059345ok
Match found in 42 steps
Beginning match attempt at character 89
Match attempt failed after 1 steps
Beginning match attempt at character 90

```

Figure F-6: Regex Pattern Matching Sequence for CCSDS Timecode

**Sample Regular Expression for ‘Keyword = NonDecimalString’**

Another common and useful Regex addresses the setting of keywords to a free-text string. Consistent with requirements for KVN values as specified in 7.5, the Regex shown in figure F-7 matches this sequence.

```
^(?:\s*)(?<keyword>[0-9A-Z_]*)(?:\s*)=(?:\s*)(?<value>(?:(?:[0-9A-Z_\- ]*)|(?:[0-9a-z_\- ]*)))(?:\s*)$
```

Figure F-7: Regex for a Non-Decimal String

**Sample Regular Expression for ‘Keyword = FreeTextString’**

Free-text strings that contain exclusively uppercase or lowercase alphabetical characters, numbers, and/or the decimal point, hyphen, space or underscore characters may be matched as follows. It should be noted that the inclusion of a decimal point in this Regex sequence means that numerical KVN values can be matched with potentially unintended consequences; numbers should be specifically matched as a number (as shown below). Consistent with requirements for KVN values as specified in 7.5, the Regex shown in figure F-8 matches this sequence.

```
^(?:\s*)(?<keyword>[0-9A-Z_]*)(?:\s*)=(?:\s*)(?<value>(?:(?:[0-9A-Z_\.\- ]*)|(?:[0-9a-z_\.\- ]*)))(?:\s*)$
```

Figure F-8: Regex for Free-Text String

**Sample Regular Expression for ‘Keyword = CCSDS Numerical Value with optional units’**

Consistent with the requirements set forth in 7.5.4 through 7.5.7, KVN values that are integers or non-integer fixed or floating-point numbers may be matched with the Regex sequence provided in figure F-9, with named groups ‘keyword’, ‘value’, and ‘units’ set accordingly:

```
^(?:\s*)(?<keyword>[0-9A-Za-z_]*)(?:\s*)=(?:\s*)(?<value>(?:[+-]?(?:[0-9]+)(?:\.\d*)?(?:[eE][+-]?)(?:\d+)))(?:\s*)(?<unit>[0-9A-Za-z/_]*\s*)?(?:\s*)?$
```

Figure F-9: Regex for String Containing Numerical Value with Optional Units

**Sample Regular Expression for ‘Keyword = Multipartite CCSDS Numerical Values’**

For orbit, covariance, and/or maneuver lines providing a multipartite sequence of numerical values, the Regex pattern provided in figure F-10 may be used (in this example, to capture and set value1, value2, and value3 for a 3-element set of numbers):

```
^(?:\s*)(?<value1>(?:[-+]?)(?:[0-9]*)(?:\.\d*)?(?:[eE][+-]
]?\d+)?)(?:\s+)(?<value2>(?:[-+]?)(?:[0-
9]*)(?:\.\d*)?(?:[eE][+-]?\d+)?)(?:\s+)(?<value3>(?:[-
+]?)?(?:[0-9]*)(?:\.\d*)?(?:[eE][+-]?\d+)?)(?:\s*)$
```

**Figure F-10: Regex for String Containing Numerical Value with Optional Units**

**F7 SPECIFIC ENERGY DISSIPATION RATE (SEDR) FORMULATION**

An orbit’s Specific Energy Dissipation Rate (SEDR) represents the amount of energy (W/kg) being removed from a satellite’s orbit by atmospheric drag. It is a very useful metric for characterizing satellites since it accounts for both the drag environment (atmospheric density) and the ‘area to mass ratio’ of the specific object. It does this by including *drag acceleration* in the computation. Drag acceleration is proportional to atmospheric density and to satellite area to mass.

SEDR is computed as follows:

Instantaneous SEDR at time t is given by

$$SEDR(t) = -\vec{A}_b \cdot \vec{V}$$

where,

$\vec{A}_b$  = drag acceleration vector (inertial)

$\vec{V}$  = velocity vector (inertial)

Average SEDR over the orbit determination interval is given by

$$\frac{1}{T} \int_0^T SEDR(t) dt$$

where, to correctly average over a complete orbital revolution, *T* is an integer multiple of the satellite period. This consideration is primarily for eccentric orbits. Aside from this consideration, *T* is the orbit determination interval.

## **F8 ORBIT DETERMINATION PARAMETERS**

### **F8.1 GENERAL**

Satellite Orbit Determination (OD) estimates the position and velocity of an orbiting object from discrete observations (for greater detail, see reference [H7]). The set of observations includes external measurements from terrestrial or space-based sensors and measurements from instruments on the satellite itself. Satellite orbit propagation estimates the future state of motion of a satellite whose orbit has been determined from past observations. Though a satellite's motion is described by a set of ideal equations of motion representing physical hypotheses, the observations used in OD are subject to systematic and random uncertainties. Therefore OD and propagation are probabilistic and can only approximately describe the satellite's motion. The degree of approximation that can be tolerated depends on the intended use of the orbital information.

Satellite owners/operators employ different techniques to determine orbits from active and passive observations, such that the same data inputs lead to different predictions when they are used in different models. Satellite owners/operators often accept orbit descriptions developed using physical models that others employ. Differences in orbit predictions caused using different physical models and numerical techniques can be significant.

A spacecraft is influenced by a variety of external forces, including terrestrial gravity, atmospheric drag, multibody gravitation, solar radiation pressure, tides, and spacecraft thrusters. Selection of forces for modelling depends on the accuracy and precision required by the OD process and the amount of available data. The complex modelling of these forces results in a highly nonlinear set of dynamical equations. Many physical and computational uncertainties limit the accuracy and precision of the spacecraft state that can be determined. Similarly, the observational data are inherently nonlinear with respect to the state of motion of the spacecraft, and some influences might not have been included in models of the observation of the state of motion.

Satellite OD and propagation are stochastic estimation problems because observations are inherently noisy and uncertain and because not all phenomena that influence satellite motion are clearly discernible. Estimation is the process of extracting a desired time-varying signal from statistically noisy observations accumulated over time. Estimation encompasses data smoothing, which is statistical inference from past observations; filtering, which infers the signal from past observations and current observations; and prediction or propagation, which employs past and current observations to infer the future of the signal.

### **F8.2 INITIAL OD**

Initial OD (IOD) methods input tracking measurements with tracking platform locations, and output spacecraft position and velocity estimates. No a priori orbit estimate is required. Associated solution error magnitudes can be very large. IOD methods are sometimes nonlinear methods and are often trivial to implement. Measurement editing is typically not performed during IOD calculations because there are insufficient observations.

Operationally, the OD process is frequently begun, or restarted, with IOD. IOD methods were derived by various authors: LaPlace, Poincaré, Gauss, Lagrange, Lambert, Gibbs, Herrick, Williams, Stumpp, Lancaster, Blanchard, Gooding, and Smith. Restarting techniques are most easily accomplished by using a solution from another technique.

### **F8.3 METHODS FOR SUBSEQUENT OD**

#### **F8.3.1 Least Squares Differential Correction**

Least Squares (LS) methods input tracking measurements with tracking platform locations and an a priori orbit estimate and output a refined orbit estimate. Associated solution error magnitudes are small when compared to IOD outputs. LS methods consist of an iterative sequence of corrections where sequence convergence is defined as a function of tracking measurement residual Root Mean Square (RMS). Each correction is characterized by a minimization of the sum of squares of tracking measurement residuals. The LS method was derived first by Gauss in 1795 and then independently by Legendre.

#### **F8.3.2 Sequential Processing**

Sequential Processing (SP) methods are distinguished from LS processing methods in that batches of data are considered sequentially, collecting a set of observations over a specified time interval and batch-processing one interval after the next. SP can be thought of as a moving time window whose contents are captured and processed at intervals, independent of previously processed batches of data. The analysis does not include process noise inputs and calculations. It is in no way equivalent to filter processing, in which each new observation is added to past observations, improving estimates in a rigorous, traceable manner.

#### **F8.3.3 Filter Processing**

Filter methods output refined state estimates sequentially at each observation time. Filter methods are forward-time recursive sequential methods consisting of a repeating pattern of time updates of the state of motion estimate and measurement updates of the state of motion estimate. The filter time update propagates the state estimate forward, and the filter measurement update incorporates the next measurement. The recursive pattern includes an important interval of filter initialization. Filter-smoother methods are backward-time recursive sequential methods consisting of a repeating pattern of state estimate refinement using filter outputs and backwards transition. Time transitions for both filter and smoother are dominated most significantly by numerical orbit propagators. The search for sequential processing was begun by Wiener, Kalman, Bucy, and others.



## F8.4 REQUIRED INFORMATION FOR ORBIT DETERMINATION

### F8.4.1 Observations

When observation data are communicated for collaborative or independent determination of satellite orbits, it is important to convey the observation types upon which that information is based. Ground-based, airborne, and space-based sensor observations are routinely used in orbit determination. These are conveyed in the CCSDS navigation family of messages using the TDM (reference [9]). Many of these parameters are discussed in greater detail in reference [H7].

The following table describes some of the various observation types and sources.

**Table F-1: Space Surveillance Observation Product Description**

Content	Source
two angles and slant range	Radars
two angles	Baker-Nunn cameras, telescopes, binoculars, visual sightings
Azimuth	Direction finders, radio antenna, Radio telescope
Time of closest approach	Radars, radio receivers [for transmitting (Doppler) satellites], relay satellite
Range, angles, and rates	Radars, radio antenna, radio telescope
Pseudorange and carrier phase, as well as single, double, and triple differences of these basic measurement types	GPS or onboard inertial sensors
Direction cosines	Interferometric radars

### F8.4.2 Observation Location Information

When data are communicated for collaborative or independent determination of satellite orbits, the following information about the observation location and measuring devices is important:

- facility location latitude, longitude, altitude, and the reference from which such are measured, (e.g., WGS-84);
- calibration/correction values associated with the ground path;
- tracking station identification (ID);
- elevation cutoff;
- measurement biases;
- clock and/or almanac information.

### F8.4.3 Satellite Information

When performing OD using active transponder ranging, the transponder delay must be provided.

### F8.4.4 Estimation Parameters and Control

When data are communicated for collaborative or independent determination of satellite orbits, the following information about estimation parameters and control are necessary, as described in 6.2.10:

- estimation parameters;
- global force model controls;
- integration controls;
- database controls;
- observation uncertainties.

### F8.4.5 Force Model Settings

#### F8.4.5.1 General

Spacecraft are affected by conservative and non-conservative forces. Non-conservative phenomena dissipate spacecraft energy, for example by doing work on and heating the atmosphere, as described in 6.2.9.

#### F8.4.5.2 Gravity

Central body gravitational fields are typically described using terms of a Jacobi polynomial expansion of finite order and degree. Jacobi polynomials are a complete, orthonormal set over the unit sphere. There are two angular degrees of freedom, equivalent to latitude and longitude. Any analytic function within that space can be represented by a weighted doubly infinite series of Jacobi polynomials.

**Two-body motion or Keplerian motion** considers only the point-mass gravity of the attracting body. Both the spacecraft and the central body are considered point masses, with all mass concentrated at their centres of mass. This is the lowest-order zonal harmonic approximation.

A **J2 zonal perturbation** (first order) accounts for secular (constant rate over time) variations in the orbit elements due to central body oblateness, mainly nodal precession and rotation of the semi-major axis of orbit elements that are otherwise those of unperturbed, Newtonian orbits. J2 is a zonal harmonic coefficient in an infinite Jacobi polynomial series

representation of the central body's gravity field. The even zonal harmonic coefficients of the gravity field are the only coefficients that result in secular changes in satellite orbital elements. The J2 propagator includes only the dominant first-order secular effects.

For **generalized spherical harmonics**, it is impractical to determine the weights (coefficients) for a mathematically complete Jacobi polynomial series representation; therefore the series is truncated at meaningful (in terms of precision of the representation of the gravity field) order (latitudinal) and degree (longitudinal). Whenever practical, it is recommended to use the full degree and order of the determined spherical harmonics field, as further truncation leads to introduction of non-conservative forces and deviation from the intended fidelity of the gravitational model.

If the order and degree are equal, the truncation is 'square'. Since gravitational and other perturbations are not necessarily symmetrical in latitude and longitude, the best approximation for a given application is not necessarily square. The GRAVITY\_MODEL keyword in 6.2.9 can specify (independently) the degree and order that are used.

Static elements of the gravity field are the gravitation of the fixed portions of the distribution of the Earth's mass. The static gravity field is not uniform. Dynamic elements of the gravity are caused by the fluid elements of the Earth's core and by variations in the distribution of water. There are solid and ocean tides. The OCEAN\_TIDES\_MODEL and SOLID\_TIDES\_MODEL of 6.2.9 can be used to specify these settings.

**Multibody gravitation:** Certain phenomena only exist with more than two gravitationally interacting bodies. It is therefore important to describe information about third-body or multiple-body gravitational interactions if such are considered. The N\_BODY\_PERTURBATIONS keyword in 6.2.9 is used to specify which bodies were modelled.

### **F8.4.5.3 Atmospheric Resistance ('Drag')**

#### **F8.4.5.3.1 General**

Gas-dynamic resistance can be a significant dissipative force in low altitude orbits about any body with a significant atmosphere, for example, Earth (LEOs), Venus, and Mars. It is usually sufficient to represent it as aerodynamic drag, the product of dynamic pressure, aggregated drag coefficient, and cross-sectional area.

#### **F8.4.5.3.2 Drag Coefficient**

Drag coefficient depends upon satellite geometry, orientation, and gas-dynamic regime described by Knudsen number (ratio of object characteristic dimension to gas mean free path) and Mach number (ratio of object speed to acoustic propagation speed). When describing how atmospheric resistance is represented, data providers provide the value of drag coefficient employed using the keyword in 6.2.9.

#### **F8.4.5.3 Atmospheric Density Model**

Density within the Earth's atmosphere varies temporally and spatially. Those variations are important in LEO. Some acceptable and most-often used atmospheric density models are as follows (although many may be utilized, as specified by the `ATMOSPHERIC_MODEL` keyword in 6.2.9):

- 1976 Standard Harris-Priester;
- Jacchia 1970 and 1971;
- Jaccia-Roberts;
- Mass-Spectrometer-Incoherent-Scatter (MSIS) model, in several versions and extensions;
- Mars Global Reference Atmosphere Model (MARSGRAM);
- Venus Global Reference Atmosphere Model (VENUSGRAM).

These models typically require measurable input parameters that are 'proxies' for the variation of atmospheric parameters. These include solar flux/geomagnetic particle flux, which can be inferred from the meteorological observables as set by the keyword `SW_DATA_SOURCE` or `FIXED_YYYYYY` of 6.2.9:

- daily F10.7;
- average F10.7; and
- geomagnetic index.

#### **F8.4.5.4 Radiation Pressure**

Momentum transfer from photons to satellites can be an important force for High Earth Orbits (HEOs), cislunar trajectories, and interplanetary trajectories. For solar sail missions, such radiation pressure is likely the primary source of propulsion. Radiation pressure depends on the area and surface characteristics of the satellite and the nature of the incident radiative fluxes. The Sun is the predominant direct source of electromagnetic radiation, but the Earth and the Moon also emit and reflect electromagnetic radiation. The keywords provided in 6.2.9 (`SHADOW_BODIES`, `SRP_MODEL`, `ALBEDO_MODEL`, `ALBEDO_GRID_SIZE`) allow the user to specify radiation pressure settings used in the OD and orbit propagation regarding:

- solar radiation pressure coefficient;
- area-to-mass ratio;
- satellite reflectivity;
- shadow and shape factor models;
- eclipse models (cylindrical, dual-cone);
- albedo and intensity at the satellite.

#### **F8.4.6 Orbit Propagation**

Orbit propagation or prediction has evolved synchronously with advances in computational capability. Initially, force models were greatly simplified, and most important non-gravitational forces were approximated analytically. These generally linearized approaches were valid only over short intervals or for small variations from two-body Keplerian motion.

Even when more precise numerical integration became feasible, execution times were often too long, and computation was too expensive to employ numerical integration on a regular basis. Semi-analytical techniques emerged that reduced numerical complexity and maintained run speed efficiency (with some compromise to precision) by providing formulae from which significant elements of the propagation workflow could be extracted.

Purely numerical integration of force models (i.e., not employing singly or doubly averaged physical approximations to describe important physical phenomena) are degraded primarily to the numerical phenomena degradation common to all discrete computations.

Analytical, numerical, and semi-analytical orbit propagation techniques are all known as ‘orbit propagators’.

#### **F8.4.7 Orbit Elements**

Orbit elements are the sets of parameters that emerge from the smoothing, filtering, or predictive estimation schemes. Six independent quantities and orbit elements describe the orbit of a satellite. A seventh variable designates the satellite position at a specific time of interest (epoch). There are many different sets of orbit elements (see orbit element set type; selected per annex B, subsection B7). Each is best suited for a particular application, such as aiming antennas, ease of manipulation in various coordinate systems, or estimating orbits from different types of measurements.

The traditionally used set of orbital elements is called the set of Keplerian elements. Keplerian elements parameters can be encoded as text in several formats. In semi-analytical propagation, mean orbit elements are often used; the most common of them is as conveyed in the NASA/NORAD TLE format, originally designed for use with 80-column punched cards (but still in use because it is the most common format).

## ANNEX G

### ODM EXAMPLES

#### (INFORMATIVE)

#### G1 OVERVIEW

The following are examples of OPMs.

#### G2 OPM EXAMPLES IN KVN

The following figures are examples of OPMs in KVN format. The first has only a state; the second has state, Keplerian elements, and maneuvers; and the third and fourth include the position/velocity covariance matrix.

```

CCSDS_OPM_VERS = 3.0
CREATION_DATE  = 2022-11-06T09:23:57
ORIGINATOR     = JAXA

COMMENT        GEOCENTRIC, CARTESIAN, EARTH FIXED
OBJECT_NAME    = OSPREY 5
OBJECT_ID      = 1998-999A
CENTER_NAME    = EARTH
REF_FRAME      = ITRF2000
TIME_SYSTEM    = UTC

EPOCH =          2022-12-18T14:28:15.1172
X =             6503.514000
Y =             1239.647000
Z =            -717.490000
X_DOT =         -0.873160
Y_DOT =          8.740420
Z_DOT =         -4.191076
MASS =          3000.000000
SOLAR_RAD_AREA = 18.770000
SOLAR_RAD_COEFF = 1.000000
DRAG_AREA =     18.770000
DRAG_COEFF =    2.500000

```

**Figure G-1: Simple OPM File Example**

```

CCSDS_OPM_VERS      = 3.0

COMMENT  Generated by GSOC, R. Kiehling
COMMENT  Current intermediate orbit IO2 and maneuver planning data

CREATION_DATE       = 2021-06-03T05:33:00.000
ORIGINATOR          = GSOC

OBJECT_NAME         = EUTELSAT W4
OBJECT_ID           = 2021-028A
CENTER_NAME         = EARTH
REF_FRAME           = TOD
TIME_SYSTEM         = UTC

COMMENT  State Vector
EPOCH               = 2021-06-03T00:00:00.000
X                   = 6655.9942           [km]
Y                   = -40218.5751        [km]
Z                   = -82.9177           [km]
X_DOT               = 3.11548208         [km/s]
Y_DOT               = 0.47042605         [km/s]
Z_DOT               = -0.00101495        [km/s]

COMMENT  Keplerian elements
SEMI_MAJOR_AXIS     = 41399.5123         [km]
ECCENTRICITY        = 0.020842611
INCLINATION         = 0.117746           [deg]
RA_OF_ASC_NODE      = 17.604721         [deg]
ARG_OF_PERICENTER   = 218.242943        [deg]
TRUE_ANOMALY        = 41.922339         [deg]
GM                  = 398600.4415        [km**3/s**2]

COMMENT  Spacecraft parameters
MASS                 = 1913.000           [kg]
SOLAR_RAD_AREA      = 10.000             [m**2]
SOLAR_RAD_COEFF     = 1.300
DRAG_AREA           = 10.000             [m**2]
DRAG_COEFF          = 2.300

COMMENT  2 planned maneuvers

COMMENT  First maneuver: AMF-3
COMMENT  Non-impulsive, thrust direction fixed in inertial frame
MAN_EPOCH_IGNITION = 2021-06-03T09:00:34.1
MAN_DURATION        = 132.60             [s]
MAN_DELTA_MASS      = -18.418            [kg]
MAN_REF_FRAME       = EME2000
MAN_DV_1            = -0.02325700        [km/s]
MAN_DV_2            = 0.01683160         [km/s]
MAN_DV_3            = -0.00893444        [km/s]

COMMENT  Second maneuver: first station acquisition maneuver
COMMENT  impulsive, thrust direction fixed in RTN frame
MAN_EPOCH_IGNITION = 2021-06-05T18:59:21.0
MAN_DURATION        = 0.00               [s]
MAN_DELTA_MASS      = -1.469             [kg]
MAN_REF_FRAME       = RTN
MAN_DV_1            = 0.00101500         [km/s]
MAN_DV_2            = -0.00187300        [km/s]
MAN_DV_3            = 0.00000000         [km/s]

```

**Figure G-2: OPM File Example with Optional Keplerian Elements and Two Maneuvers**

```

CCSDS_OPM_VERS = 3.0

CREATION_DATE = 2022-11-06T09:23:57
ORIGINATOR    = JAXA
MESSAGE_ID    = OPM 201113719185

COMMENT       GEOCENTRIC, CARTESIAN, EARTH FIXED
OBJECT_NAME   = OSPREY 5
OBJECT_ID     = 2022-999A
CENTER_NAME   = EARTH
REF_FRAME     = ITRF1997
TIME_SYSTEM   = UTC

EPOCH =      2022-12-18T14:28:15.1172
X =          6503.514000
Y =          1239.647000
Z =          -717.490000
X_DOT =      -0.873160
Y_DOT =       8.740420
Z_DOT =      -4.191076

MASS =        3000.000000
SOLAR_RAD_AREA = 18.770000
SOLAR_RAD_COEFF = 1.000000
DRAG_AREA =   18.770000
DRAG_COEFF =  2.500000

CX_X = 3.331349476038534e-04
CY_X = 4.618927349220216e-04
CY_Y = 6.782421679971363e-04
CZ_X = -3.070007847730449e-04
CZ_Y = -4.221234189514228e-04
CZ_Z = 3.231931992380369e-04
CX_DOT_X = -3.349365033922630e-07
CX_DOT_Y = -4.686084221046758e-07
CX_DOT_Z = 2.484949578400095e-07
CX_DOT_X_DOT = 4.296022805587290e-10
CY_DOT_X = -2.211832501084875e-07
CY_DOT_Y = -2.864186892102733e-07
CY_DOT_Z = 1.798098699846038e-07
CY_DOT_X_DOT = 2.608899201686016e-10
CY_DOT_Y_DOT = 1.767514756338532e-10
CZ_DOT_X = -3.041346050686871e-07
CZ_DOT_Y = -4.989496988610662e-07
CZ_DOT_Z = 3.540310904497689e-07
CZ_DOT_X_DOT = 1.869263192954590e-10
CZ_DOT_Y_DOT = 1.008862586240695e-10
CZ_DOT_Z_DOT = 6.224444338635500e-10
    
```

**Figure G-3: File Example with Covariance Matrix**



```

CCSDS_OPM_VERS = 3.0
COMMENT Generated by GSOC, R. Kiehling
COMMENT Current intermediate orbit IO2 and maneuver planning data
CREATION_DATE = 2021-06-03T05:33:00.000
ORIGINATOR = GSOC
OBJECT_NAME = EUTELSAT W4
OBJECT_ID = 2021-028A
CENTER_NAME = EARTH
REF_FRAME = TOD
TIME_SYSTEM = UTC
COMMENT State Vector
EPOCH = 2021-06-03T00:00:00.000
X = 6655.9942 [km]
Y = -40218.5751 [km]
Z = -82.9177 [km]
X_DOT = 3.11548208 [km/s]
Y_DOT = 0.47042605 [km/s]
Z_DOT = -0.00101495 [km/s]
COMMENT Keplerian elements
SEMI_MAJOR_AXIS = 41399.5123 [km]
ECCENTRICITY = 0.020842611
INCLINATION = 0.117746 [deg]
RA_OF_ASC_NODE = 17.604721 [deg]
ARG_OF_PERICENTER = 218.242943 [deg]
TRUE_ANOMALY = 41.922339 [deg]
GM = 398600.4415 [km**3/s**2]
COMMENT Spacecraft parameters
MASS = 1913.000 [kg]
SOLAR_RAD_AREA = 10.000 [m**2]
SOLAR_RAD_COEFF = 1.300
DRAG_AREA = 10.000 [m**2]
DRAG_COEFF = 2.300
COV_REF_FRAME = RTN
CX_X = 3.331349476038534e-04 [km**2]
CY_X = 4.618927349220216e-04 [km**2]
CY_Y = 6.782421679971363e-04 [km**2]
CZ_X = -3.070007847730449e-04 [km**2]
CZ_Y = -4.221234189514228e-04 [km**2]
CZ_Z = 3.231931992380369e-04 [km**2]
CX_DOT_X = -3.349365033922630e-07 [km**2/s]
CX_DOT_Y = -4.686084221046758e-07 [km**2/s]
CX_DOT_Z = 2.484949578400095e-07 [km**2/s]
CX_DOT_X_DOT = 4.296022805587290e-10 [km**2/s**2]
CY_DOT_X = -2.211832501084875e-07 [km**2/s]
CY_DOT_Y = -2.864186892102733e-07 [km**2/s]
CY_DOT_Z = 1.798098699846038e-07 [km**2/s]
CY_DOT_X_DOT = 2.608899201686016e-10 [km**2/s**2]
CY_DOT_Y_DOT = 1.767514756338532e-10 [km**2/s**2]
CZ_DOT_X = -3.041346050686871e-07 [km**2/s]
CZ_DOT_Y = -4.989496988610662e-07 [km**2/s]
CZ_DOT_Z = 3.540310904497689e-07 [km**2/s]
CZ_DOT_X_DOT = 1.869263192954590e-10 [km**2/s**2]
CZ_DOT_Y_DOT = 1.008862586240695e-10 [km**2/s**2]
CZ_DOT_Z_DOT = 6.224444338635500e-10 [km**2/s**2]
USER_DEFINED_EARTH_MODEL = WGS-84
    
```

**Figure G-4: OPM File Example with Optional Keplerian Elements, Covariance Matrix, and a User-defined Parameter**

### G3 OPM EXAMPLE IN XML

Figure G-5 contains an example of an OPM in XML format.

```

<?xml version="1.0" encoding="UTF-8"?>
<opm
  xmlns:xsi:noNamespaceSchemaLocation="https://sanaregistry.org/r/ndmxml_unqualified/ndmxml-
  3.0.0-master-3.0.xsd"
  id="CCSDS_OPM_VERS" version="3.0">

  <header>
    <COMMENT>THIS IS AN XML VERSION OF THE OPM</COMMENT>
    <CREATION_DATE>2022-11-06T09:23:57</CREATION_DATE>
    <ORIGINATOR>JAXA</ORIGINATOR>
    <MESSAGE_ID>OPM 201113719185</MESSAGE_ID>
  </header>
  <body>
    <segment>
      <metadata>
        <COMMENT>GEOCENTRIC, CARTESIAN, EARTH FIXED</COMMENT>
        <OBJECT_NAME>OSPREY 5</OBJECT_NAME>
        <OBJECT_ID>2022-999A</OBJECT_ID>
        <CENTER_NAME>EARTH</CENTER_NAME>
        <REF_FRAME>ITRF1997</REF_FRAME>
        <TIME_SYSTEM>UTC</TIME_SYSTEM>
      </metadata>
      <data>
        <stateVector>
          <EPOCH>2022-12-18T14:28:15.1172</EPOCH>
          <X>6503.514000</X>
          <Y>1239.647000</Y>
          <Z>-717.490000</Z>
          <X_DOT>-0.873160</X_DOT>
          <Y_DOT>8.740420</Y_DOT>
          <Z_DOT>-4.191076</Z_DOT>
        </stateVector>
        <spacecraftParameters>
          <MASS>3000.000000</MASS>
          <SOLAR_RAD_AREA>18.770000</SOLAR_RAD_AREA>
          <SOLAR_RAD_COEFF>1.000000</SOLAR_RAD_COEFF>
          <DRAG_AREA>18.770000</DRAG_AREA>
          <DRAG_COEFF>2.500000</DRAG_COEFF>
        </spacecraftParameters>
        <covarianceMatrix>
          <COV_REF_FRAME>ITRF1997</COV_REF_FRAME>
          <CX_X>0.316</CX_X>
          <CY_X>0.722</CY_X>
          <CY_Y>0.518</CY_Y>
          <CZ_X>0.202</CZ_X>
          <CZ_Y>0.715</CZ_Y>
          <CZ_Z>0.002</CZ_Z>
          <CX_DOT_X>0.912</CX_DOT_X>
          <CX_DOT_Y>0.306</CX_DOT_Y>
          <CX_DOT_Z>0.276</CX_DOT_Z>
          <CX_DOT_X_DOT>0.797</CX_DOT_X_DOT>
          <CY_DOT_X>0.562</CY_DOT_X>
          <CY_DOT_Y>0.899</CY_DOT_Y>
          <CY_DOT_Z>0.022</CY_DOT_Z>
          <CY_DOT_X_DOT>0.079</CY_DOT_X_DOT>
          <CY_DOT_Y_DOT>0.415</CY_DOT_Y_DOT>
          <CZ_DOT_X>0.245</CZ_DOT_X>
          <CZ_DOT_Y>0.965</CZ_DOT_Y>
          <CZ_DOT_Z>0.950</CZ_DOT_Z>
          <CZ_DOT_X_DOT>0.435</CZ_DOT_X_DOT>
          <CZ_DOT_Y_DOT>0.621</CZ_DOT_Y_DOT>
          <CZ_DOT_Z_DOT>0.991</CZ_DOT_Z_DOT>
        </covarianceMatrix>
      </data>
    </segment>
  </body>
</opm>

```

**Figure G-5: OPM File Example in XML Format**

NOTE – The following are examples of OMMs. All of these examples are based on the TLE shown in figure G-6.

```
GOES 9 [P]
1 23581U 95025A 07064.44075725 -.00000113 00000-0 10000-3 0 9250
2 23581 3.0539 81.7939 0005013 249.2363 150.1602 1.00273272 43169
```

**Figure G-6: Example Two Line Element Set**

#### G4 OMM EXAMPLES IN KVN

The following figures are examples of OMMs in KVN format.

```
CCSDS_OMM_VERS = 3.0
CREATION_DATE = 2020-065T16:00:00
ORIGINATOR = NOAA
MESSAGE_ID = OMM 202013719185

OBJECT_NAME = GOES 9
OBJECT_ID = 1995-025A
CENTER_NAME = EARTH
REF_FRAME = TEME
TIME_SYSTEM = UTC
MEAN_ELEMENT_THEORY = SGP/SGP4

EPOCH = 2020-064T10:34:41.4264
MEAN_MOTION = 1.00273272
ECCENTRICITY = 0.0005013
INCLINATION = 3.0539
RA_OF_ASC_NODE = 81.7939
ARG_OF_PERICENTER = 249.2363
MEAN_ANOMALY = 150.1602
GM = 398600.8
EPHEMERIS_TYPE = 0
CLASSIFICATION_TYPE = U
NORAD_CAT_ID = 23581
ELEMENT_SET_NO = 0925
REV_AT_EPOCH = 4316
BSTAR = 0.0001
MEAN_MOTION_DOT = -0.00000113
MEAN_MOTION_DDOT = 0.0
```

**Figure G-7: OMM File Example without Covariance Matrix**

```

CCSDS_OMM_VERS = 3.0
CREATION_DATE  = 2020-065T16:00:00
ORIGINATOR     = NOAA

OBJECT_NAME    = GOES 9
OBJECT_ID      = 1995-025A
CENTER_NAME    = EARTH
REF_FRAME      = TEME
TIME_SYSTEM    = UTC
MEAN_ELEMENT_THEORY = SGP/SGP4

EPOCH          = 2020-064T10:34:41.4264
MEAN_MOTION    = 1.00273272
ECCENTRICITY   = 0.0005013
INCLINATION    = 3.0539
RA_OF_ASC_NODE = 81.7939
ARG_OF_PERICENTER = 249.2363
MEAN_ANOMALY   = 150.1602
GM             = 398600.8

EPHEMERIS_TYPE = 0
CLASSIFICATION_TYPE = U
NORAD_CAT_ID   = 23581
ELEMENT_SET_NO = 0925
REV_AT_EPOCH   = 4316
BSTAR          = 0.0001
MEAN_MOTION_DOT = -0.00000113
MEAN_MOTION_DDOT = 0.0

COV_REF_FRAME = TEME
CX_X = 3.331349476038534e-04
CY_X = 4.618927349220216e-04
CY_Y = 6.782421679971363e-04
CZ_X = -3.070007847730449e-04
CZ_Y = -4.221234189514228e-04
CZ_Z = 3.231931992380369e-04
CX_DOT_X = -3.349365033922630e-07
CX_DOT_Y = -4.686084221046758e-07
CX_DOT_Z = 2.484949578400095e-07
CY_DOT_X_DOT = 4.296022805587290e-10
CY_DOT_X = -2.211832501084875e-07
CY_DOT_Y = -2.864186892102733e-07
CY_DOT_Z = 1.798098699846038e-07
CY_DOT_X_DOT_DOT = 2.608899201686016e-10
CY_DOT_Y_DOT = 1.767514756338532e-10
CZ_DOT_X = -3.041346050686871e-07
CZ_DOT_Y = -4.989496988610662e-07
CZ_DOT_Z = 3.540310904497689e-07
CZ_DOT_X_DOT = 1.869263192954590e-10
CZ_DOT_Y_DOT = 1.008862586240695e-10
CZ_DOT_Z_DOT = 6.224444338635500e-10

```

**Figure G-8: OMM File Example with Covariance Matrix**

```

CCSDS_OMM_VERS = 3.0
CREATION_DATE = 2020-065T16:00:00
ORIGINATOR = NOAA

OBJECT_NAME = GOES 9
OBJECT_ID = 1995-025A
CENTER_NAME = EARTH
REF_FRAME = TEME
TIME_SYSTEM = UTC
MEAN_ELEMENT_THEORY = SGP/SGP4

EPOCH = 2020-064T10:34:41.4264
MEAN_MOTION = 1.00273272 [rev/day]
ECCENTRICITY = 0.0005013
INCLINATION = 3.0539 [deg]
RA_OF_ASC_NODE = 81.7939 [deg]
ARG_OF_PERICENTER = 249.2363 [deg]
MEAN_ANOMALY = 150.1602 [deg]
GM = 398600.8 [km**3/s**2]
EPHEMERIS_TYPE = 0
CLASSIFICATION_TYPE = U
NORAD_CAT_ID = 23581
ELEMENT_SET_NO = 0925
REV_AT_EPOCH = 4316
BSTAR = 0.0001 [1/ER]
MEAN_MOTION_DOT = -0.00000113 [rev/day**2]
MEAN_MOTION_DDOT = 0.0 [rev/day**3]

USER_DEFINED_EARTH_MODEL = WGS-84

```

**Figure G-9: OMM with Units and a User-defined Parameter**

## G5 OMM EXAMPLE IN XML

Figure G-10 contains an example of an OMM in XML format.

```

<?xml version="1.0" encoding="UTF-8"?>
<omm xmlns:
xsi:noNamespaceSchemaLocation="https://sanaregistry.org/r/ndmxml_unqualified/ndmxml-3.0.0-
master-3.0.xsd"
  id="CCSDS_OMM_VERS" version="3.0">
  <header>
    <COMMENT> THIS IS AN XML VERSION OF THE OMM </COMMENT>
    <CREATION_DATE>2020-065T16:00:00</CREATION_DATE>
    <ORIGINATOR>NOAA</ORIGINATOR>
    <MESSAGE_ID> OMM 202013719185</MESSAGE_ID>
  </header>

  <body>
    <segment>
      <metadata>
        <OBJECT_NAME>GOES-9</OBJECT_NAME>
        <OBJECT_ID>1995-025A</OBJECT_ID>
        <CENTER_NAME>EARTH</CENTER_NAME>
        <REF_FRAME>TEME</REF_FRAME>
        <TIME_SYSTEM>UTC</TIME_SYSTEM>
        <MEAN_ELEMENT_THEORY>SGP4</MEAN_ELEMENT_THEORY>
      </metadata>

      <data>
        <meanElements>
          <EPOCH>2020-064T10:34:41.4264</EPOCH>
          <MEAN_MOTION>1.00273272</MEAN_MOTION>
          <ECCENTRICITY>0.0005013</ECCENTRICITY>
          <INCLINATION>3.0539</INCLINATION>
          <RA_OF_ASC_NODE>81.7939</RA_OF_ASC_NODE>
          <ARG_OF_PERICENTER>249.2363</ARG_OF_PERICENTER>
          <MEAN_ANOMALY>150.1602</MEAN_ANOMALY>
          <GM>398600.8</GM>
        </meanElements>
        <tleParameters>
          <NORAD_CAT_ID>23581</NORAD_CAT_ID>
          <ELEMENT_SET_NO>0925</ELEMENT_SET_NO>
          <REV_AT_EPOCH>4316</REV_AT_EPOCH>
          <BSTAR>0.0001</BSTAR>
          <MEAN_MOTION_DOT>-0.00000113</MEAN_MOTION_DOT>
          <MEAN_MOTION_DDOT>0.0</MEAN_MOTION_DDOT>
        </tleParameters>
        <covarianceMatrix>
          <COV_REF_FRAME>TEME</COV_REF_FRAME>
          <CX_X>3.331349476038534e-04</CX_X>
          <CY_X>4.618927349220216e-04</CY_X>
          <CY_Y>6.782421679971363e-04</CY_Y>
          <CZ_X>-3.070007847730449e-04</CZ_X>
          <CZ_Y>-4.221234189514228e-04</CZ_Y>
          <CZ_Z>3.231931992380369e-04</CZ_Z>
          <CX_DOT_X>-3.349365033922630e-07</CX_DOT_X>
          <CX_DOT_Y>-4.686084221046758e-07</CX_DOT_Y>
          <CX_DOT_Z>2.484949578400095e-07</CX_DOT_Z>
          <CX_DOT_X_DOT>4.296022805587290e-10</CX_DOT_X_DOT>
          <CY_DOT_X>-2.211832501084875e-07</CY_DOT_X>
          <CY_DOT_Y>-2.864186892102733e-07</CY_DOT_Y>
          <CY_DOT_Z>1.798098699846038e-07</CY_DOT_Z>
          <CY_DOT_X_DOT>2.608899201686016e-10</CY_DOT_X_DOT>
          <CY_DOT_Y_DOT>1.767514756338532e-10</CY_DOT_Y_DOT>
          <CZ_DOT_X>-3.041346050686871e-07</CZ_DOT_X>
          <CZ_DOT_Y>-4.989496988610662e-07</CZ_DOT_Y>
          <CZ_DOT_Z>3.540310904497689e-07</CZ_DOT_Z>
          <CZ_DOT_X_DOT>1.869263192954590e-10</CZ_DOT_X_DOT>
          <CZ_DOT_Y_DOT>1.008862586240695e-10</CZ_DOT_Y_DOT>
          <CZ_DOT_Z_DOT>6.224444338635500e-10</CZ_DOT_Z_DOT>
        </covarianceMatrix>
      </data>
    </segment>
  </body>
</omm>

```

**Figure G-10: OMM File Example in XML Format**

**G6 OEM EXAMPLES IN KVN**

The following figures are examples of OEMs in KVN format. Some ephemeris data lines have been omitted to save space.

```

CCSDS_OEM_VERS = 3.0
CREATION_DATE = 1996-11-04T17:22:31
ORIGINATOR = NASA/JPL

META_START
OBJECT_NAME      = MARS GLOBAL SURVEYOR
OBJECT_ID        = 1996-062A
CENTER_NAME      = MARS BARYCENTER
REF_FRAME        = EME2000
TIME_SYSTEM      = UTC
START_TIME       = 2019-12-18T12:00:00.331
USEABLE_START_TIME = 2019-12-18T12:10:00.331
USEABLE_STOP_TIME  = 2019-12-28T21:23:00.331
STOP_TIME        = 2019-12-28T21:28:00.331
INTERPOLATION    = HERMITE
INTERPOLATION_DEGREE = 7
META_STOP

COMMENT This file was produced by M.R. Pigs, OSAR NAV/JPL, 2019NOV 04. It is
COMMENT to be used for DSN scheduling purposes only.

2019-12-18T12:00:00.331 2789.619 -280.045 -1746.755 4.73372 -2.49586 -1.04195
2019-12-18T12:01:00.331 2783.419 -308.143 -1877.071 5.18604 -2.42124 -1.99608
2019-12-18T12:02:00.331 2776.033 -336.859 -2008.682 5.63678 -2.33951 -1.94687

    < intervening data records omitted here >

2019-12-28T21:28:00.331 -3881.024 563.959 -682.773 -3.28827 -3.66735 1.63861

META_START
OBJECT_NAME      = MARS GLOBAL SURVEYOR
OBJECT_ID        = 1996-062A
CENTER_NAME      = MARS BARYCENTER
REF_FRAME        = EME2000
TIME_SYSTEM      = UTC
START_TIME       = 2019-12-28T21:29:07.267
USEABLE_START_TIME = 2019-12-28T22:08:02.5
USEABLE_STOP_TIME  = 2019-12-30T01:18:02.5
STOP_TIME        = 2019-12-30T01:28:02.267
INTERPOLATION    = HERMITE
INTERPOLATION_DEGREE = 7
META_STOP

COMMENT This block begins after trajectory correction maneuver TCM-3.

2019-12-28T21:29:07.267 -2432.166 -063.042 1742.754 7.33702 -3.495867 -1.041945
2019-12-28T21:59:02.267 -2445.234 -878.141 1873.073 1.86043 -3.421256 -0.996366
2019-12-28T22:00:02.267 -2458.079 -683.858 2007.684 6.36786 -3.339563 -0.946654

    < intervening data records omitted here >

2019-12-30T01:28:02.267 2164.375 1115.811 -688.131 -3.53328 -2.88452 0.88535
    
```

**Figure G-11: OEM Example with No Acceleration, No Covariance**

```

CCSDS_OEM_VERS = 3.0

COMMENT  OEM WITH OPTIONAL ACCELERATIONS

CREATION_DATE = 2019-11-04T17:22:31
ORIGINATOR = NASA/JPL

META_START
OBJECT_NAME      = MARS GLOBAL SURVEYOR
OBJECT_ID        = 1996-028A
CENTER_NAME      = MARS BARYCENTER
REF_FRAME        = EME2000
TIME_SYSTEM      = UTC
START_TIME       = 2019-12-18T12:00:00.331
USEABLE_START_TIME = 2019-12-18T12:10:00.331
USEABLE_STOP_TIME  = 2019-12-28T21:23:00.331
STOP_TIME        = 2019-12-28T21:28:00.331
INTERPOLATION    = HERMITE
INTERPOLATION_DEGREE = 7
META_STOP

COMMENT  This file was produced by M.R. Somebody, MSOO NAV/JPL, 2021 NOV 04. It is
COMMENT  to be used for DSN scheduling purposes only.

2019-12-18T12:00:00.331  2789.6 -280.0 -1746.8  4.73 -2.50 -1.04  0.008 0.001 -0.159
2019-12-18T12:01:00.331  2783.4 -308.1 -1877.1  5.19 -2.42 -2.00  0.008 0.001  0.001
2019-12-18T12:02:00.331  2776.0 -336.9 -2008.7  5.64 -2.34 -1.95  0.008 0.001  0.159

    < intervening data records omitted here >

2019-12-28T21:28:00.331 -3881.0  564.0 -682.8 -3.29 -3.67  1.64 -0.003 0.000  0.000
    
```

**Figure G-12: OEM Example with Optional Accelerations**



```

CCSDS_OEM_VERS = 3.0
CREATION_DATE = 2019-11-04T17:22:31
ORIGINATOR = NASA/JPL
MESSAGE_ID = OEM 201113719185
META_START
OBJECT_NAME           = MARS GLOBAL SURVEYOR
OBJECT_ID             = 1996-062A
CENTER_NAME           = MARS BARYCENTER
REF_FRAME             = EME2000
TIME_SYSTEM           = UTC
START_TIME            = 2019-12-28T21:29:07.267
USEABLE_START_TIME    = 2019-12-28T22:08:02.5
USEABLE_STOP_TIME     = 2019-12-30T01:18:02.5
STOP_TIME             = 2019-12-30T01:28:02.267
INTERPOLATION         = HERMITE
INTERPOLATION_DEGREE = 7
META_STOP

COMMENT This block begins after trajectory correction maneuver TCM-3.

2019-12-28T21:29:07.267 -2432.166 -063.042 1742.754 7.33702 -3.495867 -1.041945
2019-12-28T21:59:02.267 -2445.234 -878.141 1873.073 1.86043 -3.421256 -0.996366
2019-12-28T22:00:02.267 -2458.079 -683.858 2007.684 6.36786 -3.339563 -0.946654

  < intervening data records omitted here >

2019-12-30T01:28:02.267 2164.375 1115.811 -688.131 -3.53328 -2.88452 0.88535

COVARIANCE_START
EPOCH = 2019-12-28T21:29:07.267
COV_REF_FRAME = EME2000
 3.3313494e-04
 4.6189273e-04 6.7824216e-04
-3.0700078e-04 -4.2212341e-04 3.2319319e-04
-3.3493650e-07 -4.6860842e-07 2.4849495e-07 4.2960228e-10
-2.2118325e-07 -2.8641868e-07 1.7980986e-07 2.6088992e-10 1.7675147e-10
-3.0413460e-07 -4.9894969e-07 3.5403109e-07 1.8692631e-10 1.0088625e-10 6.2244443e-10

EPOCH = 2019-12-29T21:00:00
COV_REF_FRAME = EME2000
 3.4424505e-04
 4.5078162e-04 6.8935327e-04
-3.0600067e-04 -4.1101230e-04 3.3420420e-04
-3.2382549e-07 -4.5750731e-07 2.3738384e-07 4.3071339e-10
-2.1007214e-07 -2.7530757e-07 1.6870875e-07 2.5077881e-10 1.8786258e-10
-3.0302350e-07 -4.8783858e-07 3.4302008e-07 1.7581520e-10 1.0077514e-10 6.2244443e-10
COVARIANCE_STOP
    
```

**Figure G-13: OEM Example with Optional Covariance Matrices**

**G7 OEM EXAMPLE IN XML**

Figure G-14 contains an example of an Orbit Ephemeris Message in XML format.

```

<?xml version="1.0" encoding="UTF-8"?>
<oem xmlns:
xsi:noNamespaceSchemaLocation="https://sanaregistry.org/r/ndmxml_unqualified/ndmxml-3.0.0-
master-3.0.xsd"
    id="CCSDS_OEM_VERS" version="3.0">

  <header>
    <COMMENT>OEM WITH OPTIONAL ACCELERATIONS</COMMENT>
    <CREATION_DATE>2019-11-04T17:22:31</CREATION_DATE>
    <ORIGINATOR>NASA/JPL</ORIGINATOR>
    <MESSAGE_ID>OEM 201113719185</MESSAGE_ID>
  </header>
  <body>
    <segment>
      <metadata>
        <OBJECT_NAME>MARS GLOBAL SURVEYOR</OBJECT_NAME>
        <OBJECT_ID>2021-028A</OBJECT_ID>
        <CENTER_NAME>MARS BARYCENTER</CENTER_NAME>
        <REF_FRAME>EME2000</REF_FRAME>
        <TIME_SYSTEM>UTC</TIME_SYSTEM>
        <START_TIME>2019-12-18T12:00:00.331</START_TIME>
        <USEABLE_START_TIME>2019-12-18T12:10:00.331</USEABLE_START_TIME>
        <USEABLE_STOP_TIME>2019-12-28T21:23:00.331</USEABLE_STOP_TIME>
        <STOP_TIME>2019-12-28T21:28:00.331</STOP_TIME>
        <INTERPOLATION>HERMITE</INTERPOLATION>
        <INTERPOLATION_DEGREE>7</INTERPOLATION_DEGREE>
      </metadata>
      <data>
        <COMMENT>Produced by M.R. Somebody, MSOO NAV/JPL, 2019 OCT 11. It is</COMMENT>
        <COMMENT>to be used for DSN scheduling purposes only.</COMMENT>
        <stateVector>
          <EPOCH>2019-12-18T12:00:00.331</EPOCH>
          <X>2789.6</X>
          <Y>-280.0</Y>
          <Z>-1746.8</Z>
          <X_DOT>4.73</X_DOT>
          <Y_DOT>-2.50</Y_DOT>
          <Z_DOT>-1.04</Z_DOT>
          <X_DDOT>0.008</X_DDOT>
          <Y_DDOT>0.001</Y_DDOT>
          <Z_DDOT>-0.159</Z_DDOT>
        </stateVector>
        <stateVector>
          <EPOCH>2019-12-18T12:01:00.331</EPOCH>
          <X>2783.4</X>
          <Y>-308.1</Y>
          <Z>-1877.1</Z>
          <X_DOT>5.19</X_DOT>
          <Y_DOT>-2.42</Y_DOT>
          <Z_DOT>-2.00</Z_DOT>
          <X_DDOT>0.008</X_DDOT>
          <Y_DDOT>0.001</Y_DDOT>
          <Z_DDOT>0.001</Z_DDOT>
        </stateVector>
        <stateVector>
          <EPOCH>2019-12-18T12:02:00.331</EPOCH>
          <X>2776.0</X>
          <Y>-336.9</Y>
          <Z>-2008.7</Z>
          <X_DOT>5.64</X_DOT>
          <Y_DOT>-2.34</Y_DOT>
          <Z_DOT>-1.95</Z_DOT>
          <X_DDOT>0.008</X_DDOT>
          <Y_DDOT>0.001</Y_DDOT>
          <Z_DDOT>0.159</Z_DDOT>
        </stateVector>
      </data>
    </segment>
  </body>
</oem>

```

**Figure G-14: OEM File Example in XML Format**

```

<stateVector>
  <EPOCH>2019-12-28T21:28:00.331</EPOCH>
  <X>-3881.0</X>
  <Y>564.0</Y>
  <Z>-682.8</Z>
  <X_DOT>-3.29</X_DOT>
  <Y_DOT>-3.67</Y_DOT>
  <Z_DOT>1.64</Z_DOT>
  <X_DDOT>-0.003</X_DDOT>
  <Y_DDOT>0.000</Y_DDOT>
  <Z_DDOT>0.000</Z_DDOT>
</stateVector>
<covarianceMatrix>
  <EPOCH>2019-12-28T22:28:00.331</EPOCH>
  <COV_REF_FRAME>ITRF1997</COV_REF_FRAME>
  <CX_X>0.316</CX_X>
  <CY_X>0.722</CY_X>
  <CY_Y>0.518</CY_Y>
  <CZ_X>0.202</CZ_X>
  <CZ_Y>0.715</CZ_Y>
  <CZ_Z>0.002</CZ_Z>
  <CX_DOT_X>0.912</CX_DOT_X>
  <CX_DOT_Y>0.306</CX_DOT_Y>
  <CX_DOT_Z>0.276</CX_DOT_Z>
  <CX_DOT_X_DOT>0.797</CX_DOT_X_DOT>
  <CY_DOT_X>0.562</CY_DOT_X>
  <CY_DOT_Y>0.899</CY_DOT_Y>
  <CY_DOT_Z>0.022</CY_DOT_Z>
  <CY_DOT_X_DOT>0.079</CY_DOT_X_DOT>
  <CY_DOT_Y_DOT>0.415</CY_DOT_Y_DOT>
  <CZ_DOT_X>0.245</CZ_DOT_X>
  <CZ_DOT_Y>0.965</CZ_DOT_Y>
  <CZ_DOT_Z>0.950</CZ_DOT_Z>
  <CZ_DOT_X_DOT>0.435</CZ_DOT_X_DOT>
  <CZ_DOT_Y_DOT>0.621</CZ_DOT_Y_DOT>
  <CZ_DOT_Z_DOT>0.991</CZ_DOT_Z_DOT>
</covarianceMatrix>
</data>
</segment>
</body>
</oem>

```

**Figure G-14: OEM File Example in XML Format (Continued)**

**G8 OCM EXAMPLES IN KVN**

The following figures are examples of OCMs in KVN format. The first has only a time history of orbital states and constitutes a minimal content OCM. The second includes space object characteristics and Perturbations Specifications; the third includes a time series of maneuvers, a time history of Cartesian position and velocity trajectory states, followed by a time history of Keplerian elements; and the fourth includes a time series of covariance matrices.

```

CCSDS_OCM_VERS = 3.0
CREATION_DATE = 2022-11-06T09:23:57
ORIGINATOR = JAPAN AEROSPACE EXPLORATION AGENCY
META_START
TIME_SYSTEM = UTC
EPOCH_TZERO = 2022-12-18T14:28:15.1172
META_STOP
TRAJ_START
TRAJ_BASIS=SIMULATED
TRAJ_REF_FRAME=ITRF2000
TRAJ_UNITS= [km, km, km, km/s, km/s, km/s]
    0.0  2854.5  -2916.2  -5360.7  5.90  4.86  0.52  0.0037  -0.0038  -0.0070
   120.0  5478.6   434.3  -3862.5  2.50  5.87  4.29  0.0072   0.0006  -0.0051
   240.0  4146.0 -1655.8  -5038.3  4.80  5.58  2.16  0.0054  -0.0022  -0.0066
< intervening data records omitted here >
86400.0 -1553.4      -4848.7 -4406.5  6.73  1.01  -3.53  -0.002  -0.0063  -0.0058
TRAJ_STOP

```

**Figure G-15: Simple/Succinct OCM File Example with Only Cartesian Ephemeris**

NOTE – In this example, CENTER\_NAME defaults to EARTH and orbit type (TRAJ\_TYPE) to CARTPV. In this example, at the expense of readability, KVN values are unaligned to minimize message storage and transmission size.

DRAFT CCSDS RECOMMENDED STANDARD FOR ORBIT DATA MESSAGES

```

CCSDS_OCM_VERS = 3.0
COMMENT This OCM reflects the latest conditions post-maneuver A67Z
COMMENT This example shows the specification of multiple comment lines
CREATION_DATE = 2022-11-06T09:23:57
ORIGINATOR = JAPAN AEROSPACE EXPLORATION AGENCY
MESSAGE_ID = OCM 201113719185

META_START
OBJECT_NAME = OSPREY 5
INTERNATIONAL_DESIGNATOR = 2022-999A

ORIGINATOR_POC = R. Rabbit
ORIGINATOR_POSITION = Flight Dynamics Mission Design Lead
ORIGINATOR_PHONE = (719)555-1234
ORIGINATOR_ADDRESS = 5040 Spaceflight Ave., Cocoa Beach FL USA 12345

TECH_POC = Mr. Rodgers
TECH_PHONE = (719)555-1234
TECH_EMAIL = email@email.XXX

TIME_SYSTEM = UT1
EPOCH_TZERO = 2022-12-18T00:00:00.0000

TAIMUTC_AT_TZERO = 36 [s]
UT1MUTC_AT_TZERO = .357 [s]
META_STOP

TRAJ_START
COMMENT GEOCENTRIC, CARTESIAN, EARTH FIXED
COMMENT THIS IS MY SECOND COMMENT LINE
TRAJ_BASIS = PREDICTED
TRAJ_REF_FRAME = EFG
TRAJ_TYPE = CARTPV
TRAJ_UNITS = [km, km, km, km/s, km/s, km/s]
2022-12-18T14:28:25.1172 2854.533 -2916.187 -5360.774 5.688 4.652 0.520
TRAJ_STOP

PHYS_START
COMMENT Spacecraft Physical Characteristics:
WET_MASS = 100.0 [kg]
OEB_Q1 = 0.03123
OEB_Q2 = 0.78543
OEB_Q3 = 0.39158
OEB_QC = 0.47832
OEB_MAX = 2.0 [m]
OEB_INT = 1.0 [m]
OEB_MIN = 0.5 [m]
AREA_ALONG_OEB_MAX = 0.5 [m**2]
AREA_ALONG_OEB_INT = 1.0 [m**2]
AREA_ALONG_OEB_MIN = 2.0 [m**2]
PHYS_STOP

PERT_START
COMMENT Perturbations Specification:
ATMOSPHERIC_MODEL = NRLMSIS00
GRAVITY_MODEL = EGM-96: 36D 360
GM = 398600.4415 [km**3/s**2]
N_BODY_PERTURBATIONS = MOON, SUN
FIXED_GEOMAG_KP = 12.0
FIXED_F10P7 = 105.0
FIXED_F10P7_MEAN = 120.0
PERT_STOP

USER_START
USER_DEFINED_CONSOLE_POC = MAXWELL RAFERTY
USER_DEFINED_EARTH_MODEL = WGS-84
USER_STOP

```

Figure G-16: OCM Example with Space Object Characteristics and Perturbations

DRAFT CCSDS RECOMMENDED STANDARD FOR ORBIT DATA MESSAGES

```

CCSDS_OCM_VERS = 3.0
CREATION_DATE = 2022-11-06T09:23:57
ORIGINATOR = JAPAN AEROSPACE EXPLORATION AGENCY
META_START
EPOCH_TZERO = 2022-12-18T14:28:15.1172
META_STOP

TRAJ_START
COMMENT ORBIT EPHEMERIS INCORPORATING DEPLOYMENTS AND MANEUVERS (BELOW)
TRAJ_BASIS = PREDICTED
TRAJ_REF_FRAME = TOD_EARTH
TRAJ_FRAME_EPOCH = 2022-12-18T14:28:15.1172
TRAJ_TYPE = CARTPVA
TRAJ_UNITS = [km, km, km, km/s, km/s, km/s, km/s**2, km/s**2, km/s**2]
2022-12-18T14:36:05.0 2854.5 -2916.2 -5360.7 5.90 4.86 0.52 0.0037 -0.0038 -0.0070
2022-12-18T14:38:05.0 5478.6 434.3 -3862.5 2.50 5.87 4.29 0.0072 0.0006 -0.0051
2022-12-18T14:40:05.0 4146.0 -1655.8 -5038.3 4.80 5.58 2.16 0.0054 -0.0022 -0.0066
< intervening data records omitted here >
2022-12-19T14:36:05.0 -1553.4 -4848.7 -4406.5 6.73 1.01 -3.53 -0.002 -0.0063 -0.0058
TRAJ_STOP

PHYS_START
COMMENT Spacecraft Physical Characteristics:
DRAG_CONST_AREA = 10.00 [m**2]
DRAG_COEFF_NOM = 2.300
DRAG_UNCERTAINTY = 5.0
WET_MASS = 200.0 [kg]
SRP_CONST_AREA = 4.00
SOLAR_RAD_COEFF = 1.300
PHYS_STOP

MAN_START
COMMENT = Ten 1kg objects deployed at 1 m/s from 190kg host over 90 s time
COMMENT = 20 deg off of back-track direction
MAN_ID = E W 20160305B
MAN_BASIS = CANDIDATE
MAN_DEVICE_ID = DEPLOY
MAN_PURPOSE = DEPLOY
MAN_REF_FRAME = RSW_ROTATING
MAN_COMPOSITION = TIME_RELATIVE, DEPLOY_ID, DEPLOY_DV_X, DEPLOY_DV_Y, <cont.>
DEPLOY_DV_Z, DEPLOY_MASS, DEPLOY_DV_SIGMA, DEPLOY_DV_RATIO, DEPLOY_DV_CDA
MAN_UNITS = [n/a, km/s, km/s, km/s, kg, %, n/a, m**2]
500.0 CUBESAT_10 2.8773E-4 -9.3969E-4 1.8491E-4 -1.0 5.0 -0.005025 0.033
510.0 CUBESAT_11 1.4208E-4 -9.3969E-4 3.1111E-4 -1.0 5.0 -0.005051 0.033
520.0 CUBESAT_12 -4.8670E-5 -9.3969E-4 3.3854E-4 -1.0 5.0 -0.005076 0.033
530.0 CUBESAT_13 -2.2398E-4 -9.3969E-4 2.5848E-4 -1.0 5.0 -0.005102 0.033
540.0 CUBESAT_14 -3.2817E-4 -9.3969E-4 9.6360E-5 -1.0 5.0 -0.005128 0.033
550.0 CUBESAT_15 -3.2817E-4 -9.3969E-4 -9.6360E-5 -1.0 5.0 -0.005154 0.033
560.0 CUBESAT_16 -2.2398E-4 -9.3969E-4 -2.5848E-4 -1.0 5.0 -0.005181 0.033
570.0 CUBESAT_17 -4.8670E-5 -9.3969E-4 -3.3854E-4 -1.0 5.0 -0.005208 0.033
580.0 CUBESAT_18 1.4208E-4 -9.3969E-4 -3.1111E-4 -1.0 5.0 -0.005236 0.033
590.0 CUBESAT_19 2.8773E-4 -9.3969E-4 -1.8491E-4 -1.0 5.0 -0.005263 0.033
MAN_STOP

MAN_START
COMMENT = 100 s of 0.5N +in-track thr w/eta=0.95, Isp=300s, 5% 1-sigma error
COMMENT = NOTE that this OCM specifies a future compound maneuver, with
COMMENT = deployment during low-level host platform thrusting.
MAN_ID = E W 20160305B
MAN_BASIS = CANDIDATE
MAN_DEVICE_ID = THR_01
MAN_PURPOSE = ORBIT
MAN_REF_FRAME = RSW_ROTATING
MAN_COMPOSITION = TIME_ABSOLUTE, MAN_DURA, THR_X, THR_Y, THR_Z, THR_MAG_SIGMA,
THR_INTERP, THR_ISP, THR_EFFIC
MAN_UNITS = [s, N, N, N, %, n/a, s, n/a]
2022-12-18T14:36:35.1172 500.0 100.0 0.0 0.5 0.0 5.0 ON 300.0 0.95
MAN_STOP

PERT_START
COMMENT Perturbations specification
GM = 398600.4415 [km**3/s**2]
PERT_STOP

OD_START
COMMENT Orbit Determination information
OD_ID = OD #10059
OD_PREV_ID = OD #10058
OD_METHOD = SF: ODTK
OD_EPOCH = 2022-12-18T11:17:33
OBS_USED = 273
TRACKS_USED = 91
OD_STOP
    
```

**Figure G-17: OCM Example with Deployed Objects and Low-level Thrusting Maneuver during Deployment to Make ‘String-of-Pearls’ Deployment**

NOTES

- 1 This example is aligned with a multi-deployment simulation for a future mission; thus the specific mission is not identified.
- 2 The demarcation '<cont.>' indicates that this single line is shown, for display purposes only, in multiple lines. This demarcation shall not be used in the actual OCM message, as all specified content shall be provided on the same line as its matching time tag.

DRAFT CCSDS RECOMMENDED STANDARD FOR ORBIT DATA MESSAGES

```

CCSDS_OCM_VERS = 3.0
CREATION_DATE = 2022-11-06T09:23:57
ORIGINATOR = JAPAN AEROSPACE EXPLORATION AGENCY

META_START
OBJECT_NAME = OSPREY 5
OBJECT_DESIGNATOR= 98765
INTERNATIONAL_DESIGNATOR = 2022-999A
CATALOG_NAME = CSPOC

TIME_SYSTEM = UTC
EPOCH_TZERO = 2022-12-18T14:28:15.1172
META_STOP

TRAJ_START
TRAJ_BASIS = PREDICTED
TRAJ_REF_FRAME = TOD EARTH
TRAJ_FRAME_EPOCH = 2022-12-18T14:28:15.1172
USEABLE_START_TIME=2022-12-18T14:32:15.1172
USEABLE_STOP_TIME= 2022-12-19T14:26:15.1172
TRAJ_TYPE = CARTPVA
TRAJ_UNITS = [km, km, km, km/s, km/s, km/s, km/s**2, km/s**2, km/s**2]
0.0 2854.5 -2916.2 -5360.7 5.90 4.86 0.52 0.0037 -0.0038 -0.0070
120.0 5478.6 434.3 -3862.5 2.50 5.87 4.29 0.0072 0.0006 -0.0051
240.0 4146.0 -1655.8 -5038.3 4.80 5.58 2.16 0.0054 -0.0022 -0.0066
< intervening data records omitted here >
86400.0 -1553.4 -4848.7 -4406.5 6.73 1.01 -3.53 -0.002 -0.0063 -0.0058
TRAJ_STOP

TRAJ_START
TRAJ_BASIS = DETERMINED
TRAJ_REF_FRAME = J2000
TRAJ_TYPE = KEPLERIAN
TRAJ_UNITS = [km, n/a, deg, deg, deg, deg]
0.000000 6600.0 .03 28.5 50.0 30.0 10.0
10.000000 6600.0 .03 28.5 50.0 30.0 10.1
20.000000 6600.0 .03 28.5 50.0 30.0 10.2
< intervening data records omitted here >
500.000000 6600.0 .03 28.5 50.0 30.0 35.0
TRAJ_STOP

PHYS_START
COMMENT Spacecraft Physical Characteristics:
DRAG_CONST_AREA = 10.00 [m**2]
DRAG_COEFF_NOM = 2.300
WET_MASS = 100.0 [kg]
SRP_CONST_AREA = 4.00
SOLAR_RAD_COEFF = 1.300
PHYS_STOP

MAN_START
COMMENT = 200 s of 10N thrust (in-track transitioning to radial)
COMMENT = w/effic η=0.95, Isp=300s, 5% 1-sigma error
MAN_ID = E W 20160305B
MAN_BASIS = CANDIDATE
MAN_DEVICE_ID = THR 01
MAN_PURPOSE = ORBIT
MAN_REF_FRAME = RSW ROTATING
MAN_COMPOSITION = TIME RELATIVE, MAN_DURA, THR_X, THR_Y, THR_Z, THR_MAG_SIGMA,
THR_INTERP, THR_ISP, THR EFFIC
MAN_UNITS = [s, N, N, %, n/a, s, n/a]
500.0 100.0 0.0 10.0 0.0 5.0 ON 300.0 0.95
600.0 100.0 10.0 0.0 0.0 5.0 OFF 300.0 0.95
MAN_STOP

PERT_START
COMMENT Perturbations specification
GM = 398600.4415 [km**3/s**2]
PERT_STOP

OD_START
COMMENT Orbit Determination information
OD_ID = OD #10059
OD_PREV_ID = OD #10058
OD_METHOD = BWLS: BAHN
OD_EPOCH = 2022-12-06T11:17:33
OBS_USED = 273
TRACKS_USED = 91
OD_STOP

```

**Figure G-18: OCM Example with Multiple Orbit Time Histories, a Maneuver, Od, Cartesian, and Keplerian Ephemeris**



```

CCSDS_OCM_VERS = 3.0

CREATION_DATE = 2022-11-06T09:23:57
ORIGINATOR = JAPAN AEROSPACE EXPLORATION AGENCY

META_START
OBJECT_NAME = OSPREY 5
INTERNATIONAL_DESIGNATOR = 2022-999A
TIME_SYSTEM = UTC
EPOCH_TZERO = 2022-12-18T14:28:15.1172
META_STOP

TRAJ_START
COMMENT = GEOCENTRIC, CARTESIAN, EARTH FIXED
TRAJ_BASIS = PREDICTED
CENTER_NAME = EARTH
TRAJ_REF_FRAME = TOD EARTH
TRAJ_FRAME_EPOCH = 2022-12-18T14:28:15.1172
TRAJ_TYPE = CARTPVA

0.0 2854.5 -2916.2 -5360.7 5.90 4.86 0.52 0.0037 -0.0038 -0.0070
120.0 5478.6 434.3 -3862.5 2.50 5.87 4.29 0.0072 0.0006 -0.0051
240.0 4146.0 -1655.8 -5038.3 4.80 5.58 2.16 0.0054 -0.0022 -0.0066
< intervening data records omitted here >
86400 -1553.4 -4848.7 -4406.5 6.73 1.01 -3.53 -0.002 -0.0063 -0.0058
TRAJ_STOP

PHYS_START
COMMENT Spacecraft Physical Characteristics:
DRAG_CONST_AREA = 10.000 [m**2]
DRAG_COEFF_NOM = 2.300
WET_MASS = 1913.000 [kg]
SRP_CONST_AREA = 10.000 [m**2]
SOLAR_RAD_COEFF = 1.300
PHYS_STOP

COV_START
COV_BASIS = PREDICTED
COV_REF_FRAME = J2000
COV_TYPE = ADBARV
COV_ORDERING = LTM
COV_UNITS = [deg, deg, deg, deg, km, km/s]
10.00 3.331349e-04 4.618927e-04 6.782421e-04 -3.070007e-04 -4.221234e-04 <cont.>
3.231931e-04 -3.349365e-07 -4.686084e-07 2.484949e-07 4.296022e-10 <cont.>
-2.211832e-07 -2.864186e-07 1.798098e-07 2.608899e-10 1.767514e-10 <cont.>
-3.041346e-07 -4.989496e-07 3.540310e-07 1.869263e-10 1.008862e-10 6.224444e-10
< ... intervening data records omitted here ... >
20.0 3.442450e-04 4.507816e-04 6.893532e-04 -3.060006e-04 -4.110123e-04 <cont.>
3.342042e-04 -3.238254e-07 -4.575073e-07 2.373838e-07 4.307133e-10 <cont.>
-2.100721e-07 -2.753075e-07 1.687087e-07 2.507788e-10 1.878625e-10 <cont.>
-3.030235e-07 -4.878385e-07 3.430200e-07 1.758152e-10 1.007751e-10 6.224444e-10
COV_STOP

COV_START
COV_BASIS = PREDICTED
COV_REF_FRAME = FIXED EARTH
COV_TYPE = CARTP
COV_UNITS = [km, km, km]
2022-12-18T14:31:35.1172 3.331349e-04 4.618927e-04 6.782421e-04 -3.070007e-04 <cont.>
-4.221234e-04 3.231931e-04
COV_STOP

PERT_START
COMMENT Perturbations specification
GM = 398600.4415 [km**3/s**2]
PERT_STOP
    
```

**Figure G-19: OCM Example with Covariance Matrix**

NOTE – The demarcation ‘<cont.>’ indicates that this single line is shown, for display purposes only, in multiple lines. This demarcation shall not be used in the actual OCM message, as all specified content shall be provided on the same line as its matching time tag.

## G9 OCM EXAMPLE IN XML

The following is an example of an Orbit Comprehensive Message in XML format.

```

<?xml version="1.0" encoding="UTF-8"?>
<ocm xmlns:
xsi:noNamespaceSchemaLocation="https://sanaregistry.org/r/ndmxml_unqualified/ndmxml-3.0.0-master-3.0.xsd"
id="CCSDS_OCM_VERS" version="3.0">
  <header>
    <COMMENT>ODM V.3 Example G-2</COMMENT>
    <COMMENT>OCM example with space object characteristics and
perturbations.</COMMENT>
    <COMMENT>This OCM reflects the latest conditions post-maneuver A67Z</COMMENT>
    <COMMENT>This example shows the specification of multiple comment
lines</COMMENT>
    <CREATION_DATE>2022-11-06T09:23:57</CREATION_DATE>
    <ORIGINATOR>JAPAN AEROSPACE EXPLORATION AGENCY</ORIGINATOR>
    <MESSAGE_ID> OCM 201113719185</MESSAGE_ID> </header>
  <body>
    <segment>
      <metadata>
        <OBJECT_NAME>OSPREY 5</OBJECT_NAME>
        <INTERNATIONAL DESIGNATOR>2022-999A</INTERNATIONAL DESIGNATOR>
        <ORIGINATOR_POC>R. Rabbit</ORIGINATOR_POC>
        <ORIGINATOR_POSITION>Flight Dynamics Mission Design
Lead</ORIGINATOR_POSITION>
        <ORIGINATOR_PHONE>(719) 555-1234</ORIGINATOR_PHONE>
        <ORIGINATOR_ADDRESS>5040 Spaceflight Ave., Cocoa Beach FL USA
12345</ORIGINATOR_ADDRESS>
        <TECH_POC>Mr. Rodgers</TECH_POC>
        <TECH_PHONE>(719) 555-1234</TECH_PHONE>
        <TECH_EMAIL>email@email.XXX</TECH_EMAIL>
        <TIME_SYSTEM>UT1</TIME_SYSTEM>
        <EPOCH_TZERO>2022-12-18T00:00:00.0000</EPOCH_TZERO>
        <TAIMUTC_AT_TZERO units="s">28</TAIMUTC_AT_TZERO>
        <UT1MUTC_AT_TZERO units="s">.357</UT1MUTC_AT_TZERO>
      </metadata>
      <data>
        <traj>
          <COMMENT>GEOCENTRIC, CARTESIAN, EARTH FIXED</COMMENT>
          <COMMENT>THIS IS MY SECOND COMMENT LINE</COMMENT>
          <TRAJ BASIS>PREDICTED</TRAJ BASIS >
          <TRAJ_REF_FRAME>EFG</TRAJ_REF_FRAME>
          <TRAJ_TYPE>CARTPVA</TRAJ_TYPE>
          <trajLine>2022-12-18T14:28:25.1172 2854.5 -2916.2 -5360.7 5.90 4.86
0.52 0.0037 -0.0038 -0.0070</trajLine>
        </traj>
        <phys>
          <COMMENT>Spacecraft Physical Characteristics</COMMENT>
          <WET_MASS units="kg">100.0</WET_MASS>
          <OEB_Q1>0.03123</OEB_Q1>
          <OEB_Q2>0.78543</OEB_Q2>
          <OEB_Q3>0.39158</OEB_Q3>
          <OEB_QC>0.47832</OEB_QC>
          <OEB_MAX units="m">2.0</OEB_MAX>
          <OEB_INT units="m">1.0</OEB_INT>
          <OEB_MIN units="m">0.5</OEB_MIN>
          <AREA_ALONG_OEB_MAX units="m**2">0.5</AREA_ALONG_OEB_MAX>
          <AREA_ALONG_OEB_INT units="m**2">1.0</AREA_ALONG_OEB_INT>
          <AREA_ALONG_OEB_MIN units="m**2">2.0</AREA_ALONG_OEB_MIN>
        </phys>
        <pert>
          <COMMENT>Perturbations Specification</COMMENT>
          <ATMOSPHERIC_MODEL>NRLMSIS00</ATMOSPHERIC_MODEL>
          <GRAVITY_MODEL>EGM-96: 36D 36O</GRAVITY_MODEL>
          <GM units="km**3/s**2">398600.4415</GM>
          <N_BODY PERTURBATIONS>MOON, SUN</N_BODY PERTURBATIONS>
          <FIXED_GEOMAG_KP>12.0</FIXED_GEOMAG_KP>
          <FIXED_F10P7>I05.0</FIXED_F10P7>
          <FIXED_F10P7_MEAN>120.0</FIXED_F10P7_MEAN>
        </pert>
        <user>
          <USER_DEFINED parameter="CONSOLE_POC">MAXWELL RAFERTY</USER_DEFINED>
          <USER_DEFINED parameter="EARTH_MODEL">WGS-84</USER_DEFINED>
        </user>
      </data>
    </segment>
  </body>
</ocm>

```

Figure G-20: OCM Example in XML Format

## G10 AGGREGATING MULTIPLE ODMs IN A SINGLE NDM XML FILE

The following examples illustrate how multiple Orbit Data Messages can be aggregated in a single XML file using the NDM 'combined instantiation' schema.

```

<?xml version="1.0" encoding="UTF-8"?>
<ndm xmlns:
xsi:noNamespaceSchemaLocation="https://sanaregistry.org/r/ndmxml_unqualified/ndmxml-3.0.0-master-3.0.xsd">
  <omm id="CCSDS_OMM_VERS" version="3.0">
    <header>
      <COMMENT>GENERATED VIA SPACE-TRACK.ORG API</COMMENT>
      <CREATION_DATE>2020-05-16T14:00:01</CREATION_DATE>
      <ORIGINATOR>18 SPCS</ORIGINATOR>
    </header>
    <body>
      <segment>
        <metadata>
          <OBJECT_NAME>STARLINK-1073</OBJECT_NAME>
          <OBJECT_ID>2020-001A</OBJECT_ID>
          <CENTER_NAME>EARTH</CENTER_NAME>
          <REF_FRAME>TEME</REF_FRAME>
          <TIME_SYSTEM>UTC</TIME_SYSTEM>
          <MEAN_ELEMENT_THEORY>SGP4</MEAN_ELEMENT_THEORY>
        </metadata>
        <data>
          <meanElements>
            <EPOCH>2020-05-16T14:00:01</EPOCH>
            <MEAN_MOTION>15.05566242</MEAN_MOTION>
            <ECCENTRICITY>0.0001225</ECCENTRICITY>
            <INCLINATION>52.9981</INCLINATION>
            <RA_OF_ASC_NODE>157.6133</RA_OF_ASC_NODE>
            <ARG_OF_PERICENTER>93.35</ARG_OF_PERICENTER>
            <MEAN_ANOMALY>295.8599</MEAN_ANOMALY>
          </meanElements>
          <tleParameters>
            <EPHEMERIS_TYPE>0</EPHEMERIS_TYPE>
            <CLASSIFICATION_TYPE>U</CLASSIFICATION_TYPE>
            <NORAD_CAT_ID>44914</NORAD_CAT_ID>
            <ELEMENT_SET_NO>999</ELEMENT_SET_NO>
            <REV_AT_EPOCH>176</REV_AT_EPOCH>
            <BSTAR>0.00057678</BSTAR>
            <MEAN_MOTION_DOT>0.00008131</MEAN_MOTION_DOT>
            <MEAN_MOTION_DDOT>0</MEAN_MOTION_DDOT>
          </tleParameters>
          <userDefinedParameters>
            <USER_DEFINED parameter="TLE_LINE0">0 STARLINK-1073</USER_DEFINED>
            <USER_DEFINED parameter="TLE_LINE1">
              1 44914U 20211A 20137.58334491 +.00008131 +00000-0 +57678-3 0 9994
            </USER_DEFINED>
            <USER_DEFINED parameter="TLE_LINE2">
              2 44914 052.9981 157.6133 0001225 093.3500 295.8599
              15.05566242001761
            </USER_DEFINED>
          </userDefinedParameters>
        </data>
      </segment>
    </body>
  </omm>
</ndm>

```

**Figure G-21: Aggregating Multiple ODMs into a Single NDM File**

```

<body>
  <segment>
    <metadata>
      <OBJECT_NAME>STARLINK-1084</OBJECT_NAME>
      <OBJECT_ID>2020-001B</OBJECT_ID>
      <CENTER_NAME>EARTH</CENTER_NAME>
      <REF_FRAME>TEME</REF_FRAME>
      <TIME_SYSTEM>UTC</TIME_SYSTEM>
      <MEAN_ELEMENT_THEORY>SGP4</MEAN_ELEMENT_THEORY>
    </metadata>
    <data>
      <meanElements>
        <EPOCH>2020-05-16T14:00:01</EPOCH>
        <MEAN_MOTION>15.05603711</MEAN_MOTION>
        <ECCENTRICITY>0.000126</ECCENTRICITY>
        <INCLINATION>52.9991</INCLINATION>
        <RA_OF_ASC_NODE>157.611</RA_OF_ASC_NODE>
        <ARG_OF_PERICENTER>75.9191</ARG_OF_PERICENTER>
        <MEAN_ANOMALY>151.2104</MEAN_ANOMALY>
      </meanElements>
      <tleParameters>
        <EPHEMERIS_TYPE>0</EPHEMERIS_TYPE>
        <CLASSIFICATION_TYPE>U</CLASSIFICATION_TYPE>
        <NORAD_CAT_ID>44915</NORAD_CAT_ID>
        <ELEMENT_SET_NO>999</ELEMENT_SET_NO>
        <REV_AT_EPOCH>1986</REV_AT_EPOCH>
        <BSTAR>0.0006442</BSTAR>
        <MEAN_MOTION_DOT>0.00009125</MEAN_MOTION_DOT>
        <MEAN_MOTION_DDOT>0</MEAN_MOTION_DDOT>
      </tleParameters>
      <userDefinedParameters>
        <USER_DEFINED parameter="TLE_LINE0">0 STARLINK-1084</USER_DEFINED>
        <USER_DEFINED parameter="TLE_LINE1">
          1 44915U 20211B 20137.58334491 +.00009125 +00000-0 +64420-3 0 9992
        </USER_DEFINED>
        <USER_DEFINED parameter="TLE_LINE2">
          2 44915 052.9991 157.6110 0001260 075.9191 151.2104
          15.05603711019869
        </USER_DEFINED>
      </userDefinedParameters>
    </data>
  </segment>
</body>
</omm>
<omm id="CCSDS_OMM_VERS" version="3.0">
  <header>
    <COMMENT>GENERATED VIA SPACE-TRACK.ORG API</COMMENT>
    <CREATION_DATE>2020-05-16T14:00:01</CREATION_DATE>
    <ORIGINATOR>18 SPCS</ORIGINATOR>
  </header>
  <body>
    <segment>
      <metadata>
        <OBJECT_NAME>STARLINK-1097</OBJECT_NAME>
        <OBJECT_ID>2020-001C</OBJECT_ID>
        <CENTER_NAME>EARTH</CENTER_NAME>
        <REF_FRAME>TEME</REF_FRAME>
        <TIME_SYSTEM>UTC</TIME_SYSTEM>
        <MEAN_ELEMENT_THEORY>SGP4</MEAN_ELEMENT_THEORY>
      </metadata>
      <data>
        <meanElements>
          <EPOCH>2020-05-16T14:00:01</EPOCH>
          <MEAN_MOTION>15.05559315</MEAN_MOTION>
          <ECCENTRICITY>0.0001361</ECCENTRICITY>
          <INCLINATION>52.999</INCLINATION>
          <RA_OF_ASC_NODE>157.6123</RA_OF_ASC_NODE>
          <ARG_OF_PERICENTER>94.2334</ARG_OF_PERICENTER>
          <MEAN_ANOMALY>78.9025</MEAN_ANOMALY>
        </meanElements>

```

**Figure G-21: Aggregating Multiple ODMs into a Single NDM File (Continued)**

```

<tleParameters>
  <EPHEMERIS_TYPE>0</EPHEMERIS_TYPE>
  <CLASSIFICATION_TYPE>U</CLASSIFICATION_TYPE>
  <NORAD_CAT_ID>44916</NORAD_CAT_ID>
  <ELEMENT_SET_NO>999</ELEMENT_SET_NO>
  <REV_AT_EPOCH>1986</REV_AT_EPOCH>
  <BSTAR>0.00072742</BSTAR>
  <MEAN_MOTION_DOT>0.00010329</MEAN_MOTION_DOT>
  <MEAN_MOTION_DDOT>0</MEAN_MOTION_DDOT>
</tleParameters>
<userDefinedParameters>
  <USER_DEFINED parameter="TLE_LINE0">0 STARLINK-1097</USER_DEFINED>
  <USER_DEFINED parameter="TLE_LINE1">
    1 44916U 20211C 20137.58334491 +.00010329 +00000-0 +72742-3 0 9997
  </USER_DEFINED>
  <USER_DEFINED parameter="TLE_LINE2">
    2 44916 052.9990 157.6123 0001361 094.2334 078.9025
15.05559315019865
  </USER_DEFINED>
</userDefinedParameters>
</data>
</segment>
</body>
</omml>
</ndm>

```

**Figure G-21: Aggregating Multiple ODMs into a Single NDM File (Continued)**

```

<?xml version="1.0" encoding="UTF-8"?>
<ndm xmlns:
xsi:noNamespaceSchemaLocation="https://sanaregistry.org/r/ndmxml_unqualified/ndmxml-3.0.0-master-3.0.xsd">

<COMMENT>This example combines an OPM, OMM, OEM, and OCM in a single Navigation
Data Message XML.</COMMENT>
<COMMENT>NOTE - In this case the messages are unrelated, but show how, in
principle, related </COMMENT>
<COMMENT>messages could be combined to satisfy a specific use case.</COMMENT>

<opm id="CCSDS_OPM_VERS" version="3.0">
  <header>
    <COMMENT>Example Orbit Parameter Message</COMMENT>
    <COMMENT>The OPM includes a set of orbital elements at time t, and an
impulsive maneuver design with MAN EPOCH IGNITION = t.</COMMENT>
    <CREATION_DATE>2009-05-18T13:06:00</CREATION_DATE>
    <ORIGINATOR>NASA</ORIGINATOR>
  </header>
  <body>
    <segment>
      <metadata>
        <OBJECT_NAME>SOHO</OBJECT_NAME>
        <OBJECT_ID>2009-000A</OBJECT_ID>
        <CENTER_NAME>EARTH</CENTER_NAME>
        <REF_FRAME>EME2000</REF_FRAME>
        <TIME_SYSTEM>UTC</TIME_SYSTEM>
      </metadata>
      <data>
        <stateVector>
          <EPOCH>2009-05-12T15:30:00</EPOCH>
          <X>687754.36358524448</X>
          <Y>941287.85553999904</Y>
          <Z>520080.81101286016</Z>
          <X_DOT>-0.41396551860286032</X_DOT>
          <Y_DOT>0.29174462217893128</Y_DOT>
          <Z_DOT>0.11756781919443198</Z_DOT>
        </stateVector>
        <maneuverParameters>
          <COMMENT>The below contains information for the impulsive
maneuver.</COMMENT>
          <MAN_EPOCH_IGNITION>2009-05-12T15:30:00</MAN_EPOCH_IGNITION>
          <MAN_DURATION>0</MAN_DURATION>
          <MAN_DELTA_MASS>-0.0252069575402913408</MAN_DELTA_MASS>
          <MAN_REF_FRAME>EME2000</MAN_REF_FRAME>
          <MAN_DV_1>0.000028562811624</MAN_DV_1>
          <MAN_DV_2>3.0883529021E-7</MAN_DV_2>
          <MAN_DV_3>1.4646782842E-8</MAN_DV_3>
        </maneuverParameters>
      </data>
    </segment>
  </body>
</opm>

<omm id="CCSDS_OMM_VERS" version="3.0">
  <header>
    <COMMENT> Example Orbit Mean-Elements Message</COMMENT>
    <COMMENT> ODM PROTOTYPING FOR CCSDS 502.0-P-1.1 - TEST NUMBER 4</COMMENT>
    <CREATION_DATE>2009-091T16:00:00</CREATION_DATE>
    <ORIGINATOR>CNES</ORIGINATOR>
  </header>

```

**Figure G-22: Aggregating OPM, OMM, OEM, and OCM in a Single Navigation Data Message XML**

```

<body>
  <segment>
    <metadata>
      <OBJECT_NAME>ISS-ZARYA</OBJECT_NAME>
      <OBJECT_ID>2022-067A</OBJECT_ID>
      <CENTER_NAME>EARTH</CENTER_NAME>
      <REF_FRAME>TEME</REF_FRAME>
      <TIME_SYSTEM>UTC</TIME_SYSTEM>
      <MEAN_ELEMENT_THEORY>SGP4</MEAN_ELEMENT_THEORY>
    </metadata>
    <data>
      <COMMENT>USAF SGP4 IS THE ONLY PROPAGATOR THAT SHOULD BE USED FOR
THIS DATA</COMMENT>
      <meanElements>
        <EPOCH>2009-087T11:58:23.211264</EPOCH>
        <MEAN MOTION>15.71784080</MEAN MOTION>
        <ECCENTRICITY>0.0009738</ECCENTRICITY>
        <INCLINATION>51.6429</INCLINATION>
        <RA_OF_ASC_NODE>0.4159</RA_OF_ASC_NODE>
        <ARG_OF_PERICENTER>166.5533</ARG_OF_PERICENTER>
        <MEAN_ANOMALY>354.9076</MEAN_ANOMALY>
      </meanElements>
      <tleParameters>
        <EPHEMERIS_TYPE>0</EPHEMERIS_TYPE>
        <CLASSIFICATION_TYPE>U</CLASSIFICATION_TYPE>
        <NORAD_CAT_ID>25544</NORAD_CAT_ID>
        <ELEMENT_SET_NO>557</ELEMENT_SET_NO>
        <REV_AT_EPOCH>59325</REV_AT_EPOCH>
        <BSTAR>-0.21414E-3</BSTAR>
        <MEAN MOTION_DOT>0.00027894</MEAN MOTION_DOT>
        <MEAN MOTION_DDOT>0.0</MEAN MOTION_DDOT>
      </tleParameters>
    </data>
  </segment>
</body>
</omm>

<oem id="CCSDS_OEM_VERS" version="3.0">
  <header>
    <COMMENT>Example Orbit Ephemeris Message</COMMENT>
    <CREATION_DATE>2009-02-05T10:09:12</CREATION_DATE>
    <ORIGINATOR>JAXA</ORIGINATOR>
  </header>
  <body>
    <segment>
      <metadata>
        <OBJECT_NAME>SELENE</OBJECT_NAME>
        <OBJECT_ID>131</OBJECT_ID>
        <CENTER_NAME>EARTH</CENTER_NAME>
        <REF_FRAME>EME2000</REF_FRAME>
        <TIME_SYSTEM>UTC</TIME_SYSTEM>
        <START_TIME>2007-09-14T10:43:00.000000</START_TIME>
        <STOP_TIME>2007-09-14T10:47:00.000000</STOP_TIME>
        <INTERPOLATION>Hermite</INTERPOLATION>
        <INTERPOLATION_DEGREE>7</INTERPOLATION_DEGREE>
      </metadata>
      <data>
        <stateVector>
          <EPOCH>2007-09-14T10:43:00.000000</EPOCH>
          <X>1.336886031675458E+03</X>
          <Y>9.480146566872711E+04</Y>
          <Z>3.266088528177066E+04</Z>
          <X_DOT>-6.305293974707967E-01</X_DOT>
          <Y_DOT>1.848864859994888E+00</Y_DOT>
          <Z_DOT>9.377640352829226E-01</Z_DOT>
        </stateVector>
      </data>
    </segment>
  </body>
</oem>

```

**Figure G-22: Aggregating OPM, OMM, OEM, and OCM in a Single Navigation Data Message XML (Continued)**

```

<stateVector>
  <EPOCH>2007-09-14T10:44:00.000000</EPOCH>
  <X>1.299053338219702E+03</X>
  <Y>9.491233015399727E+04</Y>
  <Z>3.271712789822781E+04</Z>
  <X_DOT>-6.305602161673365E-01</X_DOT>
  <Y_DOT>1.846618917569059E+00</Y_DOT>
  <Z_DOT>9.369900978163187E-01</Z_DOT>
</stateVector>
<stateVector>
  <EPOCH>2007-09-14T10:45:00.000000</EPOCH>
  <X>1.261218825877070E+03</X>
  <Y>9.502306005072314E+04</Y>
  <Z>3.277332412355368E+04</Z>
  <X_DOT>-6.305900282238581E-01</X_DOT>
  <Y_DOT>1.844378570838882E+00</Y_DOT>
  <Z_DOT>9.362176632629805E-01</Z_DOT>
</stateVector>
<stateVector>
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  <X>1.223382554832220E+03</X>
  <Y>9.513365569386665E+04</Y>
  <Z>3.282947404774766E+04</Z>
  <X_DOT>-6.306188407720047E-01</X_DOT>
  <Y_DOT>1.842143793832282E+00</Y_DOT>
  <Z_DOT>9.354467257772656E-01</Z_DOT>
</stateVector>
<stateVector>
  <EPOCH>2007-09-14T10:47:00.000000</EPOCH>
  <X>1.185544584842622E+03</X>
  <Y>9.524411741683590E+04</Y>
  <Z>3.288557776045905E+04</Z>
  <X_DOT>-6.306466608867719E-01</X_DOT>
  <Y_DOT>1.839914560746108E+00</Y_DOT>
  <Z_DOT>9.346772795446093E-01</Z_DOT>
</stateVector>
</data>
</segment>
</body>
</oem>

<ocm id="CCSDS_OCM_VERS" version="3.0">

  <header>
    <COMMENT>Example G-4 from OCM P2.40</COMMENT>
    <COMMENT>OCM example with multiple trajectory state time histories, a
maneuver, OD, Cartesian and Keplerian ephemeris</COMMENT>
    <CREATION_DATE>2022-11-06T09:23:57</CREATION_DATE>
    <ORIGINATOR>JAPAN AEROSPACE EXPLORATION AGENCY</ORIGINATOR>
  </header>

  <body>
    <segment>
      <metadata>
        <OBJECT_NAME>OSPREY 5</OBJECT_NAME>
        <OBJECT_DESIGNATOR>98765</OBJECT_DESIGNATOR>
        <INTERNATIONAL_DESIGNATOR>2022-999A</INTERNATIONAL_DESIGNATOR>
        <CATALOG_NAME>CSPOC</CATALOG_NAME>
        <TIME_SYSTEM>UTC</TIME_SYSTEM>
        <EPOCH_TZERO>2022-12-18T14:28:15.1172</EPOCH_TZERO>
      </metadata>

      <data>
        <traj>
          <TRAJ_BASIS>PREDICTED</TRAJ_BASIS>
          <TRAJ_REF_FRAME>TOD EARTH</TRAJ_REF_FRAME>
          <TRAJ_FRAME_EPOCH>2022-12-18T14:28:15.1172</TRAJ_FRAME_EPOCH>
          <USEABLE_START_TIME>2022-12-18T14:28:15.1172</USEABLE_START_TIME>
          <USEABLE_STOP_TIME>2022-12-18T14:28:45.1172</USEABLE_STOP_TIME>
        </traj>
      </data>
    </segment>
  </body>
</ocm>

```

**Figure G-22: Aggregating OPM, OMM, OEM, and OCM in a Single Navigation Data Message XML (Continued)**



```

        <TRAJ_TYPE>CARTPVA</TRAJ_TYPE>
        <TRAJ_UNITS>[km, km, km, km/s, km/s, km/s, km/s**2, km/s**2,
km/s**2]</TRAJ_UNITS>
        <trajLine> 0.0 2854.5 -2916.2 -5360.7 5.90 4.86 0.52 0.0037 -
0.0038 -0.0070</trajLine>
        <trajLine>120.0 5478.6 434.3 -3862.5 2.50 5.87 4.29 0.0072
0.0006 -0.0051</trajLine>
        <trajLine>240.0 4146.0 -1655.8 -5038.3 4.80 5.58 2.16 0.0054 -
0.0022 -0.0066</trajLine>
        <trajLine>500.0 -1553.4 -4848.7 -4406.5 6.73 1.01 -3.53 -0.002 -
0.0063 -0.0058</trajLine>
    </traj>

    <traj>
        <TRAJ_BASIS>DETERMINED</TRAJ_BASIS>
        <TRAJ_REF_FRAME>J2000</TRAJ_REF_FRAME>
        <TRAJ_TYPE>KEPLERIAN</TRAJ_TYPE>
        <TRAJ_UNITS>[km, n/a, deg, deg, deg, deg]</TRAJ_UNITS>
        <trajLine>0.000000 6600.0 .03 28.5 50.0 30.0 10.0</trajLine>
        <trajLine>120.000000 6600.0 .03 28.5 50.0 30.0 10.1</trajLine>
        <trajLine>240.000000 6600.0 .03 28.5 50.0 30.0 10.2</trajLine>
        <trajLine>500.000000 6600.0 .03 28.5 50.0 30.0 35.0</trajLine>
    </traj>

    <phys>
        <COMMENT>Spacecraft Physical Characteristics</COMMENT>
        <DRAG_CONST_AREA units="m**2">10.00</DRAG_CONST_AREA>
        <DRAG_COEFF_NOM>2.3</DRAG_COEFF_NOM>
        <WET_MASS units="kg">100.0</WET_MASS>
        <SRP_CONST_AREA units="m**2">4</SRP_CONST_AREA>
        <SOLAR_RAD_COEFF >1.3</SOLAR_RAD_COEFF >
    </phys>

    <man>
        <COMMENT>200 s of 10N thrust (in-track transitioning to
radial)</COMMENT>
        <COMMENT>w/effic η=0.95, Isp=300s, 5% 1-sigma error</COMMENT>
        <MAN_ID>E_W_20160305B</MAN_ID>
        <MAN_BASIS>CANDIDATE</MAN_BASIS>
        <MAN_DEVICE_ID>THR_01</MAN_DEVICE_ID>
        <MAN_PURPOSE>ORBIT</MAN_PURPOSE>
        <MAN_REF_FRAME>RSW_ROTATING</MAN_REF_FRAME>
        <MAN_COMPOSITION>TIME_RELATIVE MAN_DURA, THR_X, THR_Y, THR_Z,
THR_MAG_SIGMA, THR_INTERP, THR_ISP, THR EFFIC</MAN_COMPOSITION>
        <MAN_UNITS>[s, N, N, N, %, n/a, s, n/a]</MAN_UNITS>
        <manLine>500.0 100.0 0.0 10.0 0.0 5.0 ON 300.0 0.95</manLine>
        <manLine>600.0 100.0 10.0 0.0 0.0 5.0 OFF 300.0 0.95</manLine>
    </man>

    <pert>
        <COMMENT>Perturbations specification</COMMENT>
        <GM>398600.4415</GM>
    </pert>

    <od>
        <COMMENT> Orbit Determination information</COMMENT>
        <OD_ID>OD #10059</OD_ID>
        <OD_PREV_ID>OD #10058</OD_PREV_ID>
        <OD_METHOD>BWLS: BAHN</OD_METHOD>
        <OD_EPOCH>2001-11-06T11:17:33</OD_EPOCH>
        <OBS_USED>273</OBS_USED>
        <TRACKS_USED>91</TRACKS_USED>
    </od>

    </data>
</segment>
</body>
</ocm>
</ndm>

```

**Figure G-22: Aggregating OPM, OMM, OEM, and OCM in a Single Navigation Data Message XML (Continued)**

## ANNEX H

## INFORMATIVE REFERENCES

## (INFORMATIVE)

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- [H16] Obed S. Sands, et al. “Dilution of Precision-Based Lunar Navigation Assessment for Dynamic Position Fixing.” In *Proceedings of the 2006 National Technical Meeting of The Institute of Navigation (January 18–20, 2006, Monterey, California)*, 260–268. Manassas, Virginia: ION, 2006.
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**ANNEX I****ITEMS FOR AN INTERFACE CONTROL DOCUMENT****(INFORMATIVE)****II STANDARD ICD ITEMS**

In several places in this document, there are references to items that should be specified in an Interface Control Document (ICD) between participants that supplements an exchange of ephemeris data. The ICD should be jointly produced by both participants in a cross-support involving the transfer of ephemeris data. This annex compiles those recommendations into a single section. Although the Orbit Data Messages described in this document may at times be used in situations in which participants have not negotiated ICDs, they should be developed and negotiated whenever specified in this Recommended Standard.

Item	Section
1) Detailed description of any user defined OPM, OMM, OEM, and OCM parameters used.	3.2, 4.2, 5.2, 6.2
2) Detailed description of any exceptions for keyword values not drawn from the SANA registry (sanaregistry.org).	annex B, subsection B6
3) Specific information security interoperability provisions that apply between agencies.	annex C

## ANNEX J

### CHANGES VERSUS PREVIOUS VERSION

#### (INFORMATIVE)

This annex lists the differences between ODM 2.0 and ODM 3.0. The differences are divided into those which affect the content of one or more of the orbit data messages, and those which only affect the document.

#### J1 CHANGES IN THE MESSAGES

1. The OCM was added to provide better support for ISO Technical Committee 20, Subcommittee 14 objectives (see section 4).
2. MESSAGE\_ID was added to the OPM, OMM, and OEM to provide better satisfaction of requirement P10 (identification and annotation of messages).

#### J2 CHANGES IN THE DOCUMENT

1. A new CCSDS repository for normative keyword values for navigation messages has been created at the SANA Registry, accessible on the Internet at: [https://sanaregistry.org/r/navigation\\_standard\\_normative\\_annexes/](https://sanaregistry.org/r/navigation_standard_normative_annexes/). (See annex B for details on the affected keywords and links to the content.)
2. Several annexes were added. Some are required by CCSDS rule changes, and some are for the provision of supplementary material.
3. Examples for OPM, OMM, and OEM that formerly appeared in sections 3, 4, and 5, respectively, have been moved to an informative annex.
4. The ‘Checklist ICD’ that was added in ODM Version 2 has been discontinued. This Checklist ICD, intended to convey information that the OPM, OEM, and OMM did not address, such as third-body perturbations, solar pressure model, solid tides, ocean tides, Earth albedo, and polar motion, has now been replaced by the material that can be specified in the Orbit Comprehensive Message.